NACA

RESEARCH MEMORANDUM

SOME EFFECTS OF WING HEIGHT ON THE VERTICAL-TAIL

PRESSURE DISTRIBUTIONS OF A COMPLETE MODEL

IN SIDESLIP AT HIGH SUBSONIC SPEEDS

By Albert G. Few, Jr., and William J. Alford, Jr.

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NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS
WASHINGTON

ERRATA

NACA Research Memorandum L57D22

By Albert G. Few, Jr., and William J. Alford, Jr.
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Page 3, last line, change formula in parenthesis to read:

$$\left(0.25 - \frac{c_{m,v}}{c_{n,v}}\right)$$

Page 4, line 3:

The formula for $\frac{\bar{x}}{\bar{c}_v}$ in the third line should be corrected to read:

$$0.25 - \left| \frac{\displaystyle\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta \beta c_{v,av}} \left[\left(\frac{z'}{b_v} - \frac{z}{b_v}\right)\frac{b_v}{\bar{c}_v}\tan\Lambda + \left(0.25 - \frac{\bar{x}}{c_v}\right)\frac{c_v}{\bar{c}_v}\right] d\left(\frac{z}{b_v}\right)}{\displaystyle\int_0^{1.0} \left(\frac{\Delta c_n c_v}{\Delta \beta c_{v,av}}\right)d\left(\frac{z}{b_v}\right)} \right|$$

where

z'   spanwise distance from vertical-tail root chord to vertical-
     tail mean aerodynamic chord, ft

Λ sweep angle of quarter-chord line, deg

Page 97:

Replace figure 11 with corrected figure 11 attached, where the plots
of $\bar{x}/\bar{c}_v$ have been revised on the basis of the corrected formulas given
in this errata.

Reference 8 (Tail force balance)

o   Present test (Pressure distributions)

$\frac{\bar{z}}{b_v}$

.60

.40

.20

0

$\frac{\bar{z}}{b_v}$ for $\frac{\bar{c}_v}{4}$

M = 0.92

.60

.40

.20

0

$\frac{\bar{x}}{\bar{c}_v}$

M = 0.92

$\frac{\bar{c}_v}{4}$

$\frac{\bar{z}}{b_v}$

.60

.40

.20

0

$\frac{\bar{z}}{b_v}$ for $\frac{\bar{c}_v}{4}$

M = 0.80

$\frac{\bar{x}}{\bar{c}_v}$

.60

.40

.20

0

M = 0.80

$\frac{\bar{c}_v}{4}$

0   5   10   15   20

Angle of attack, a, deg

0   5   10   15   20
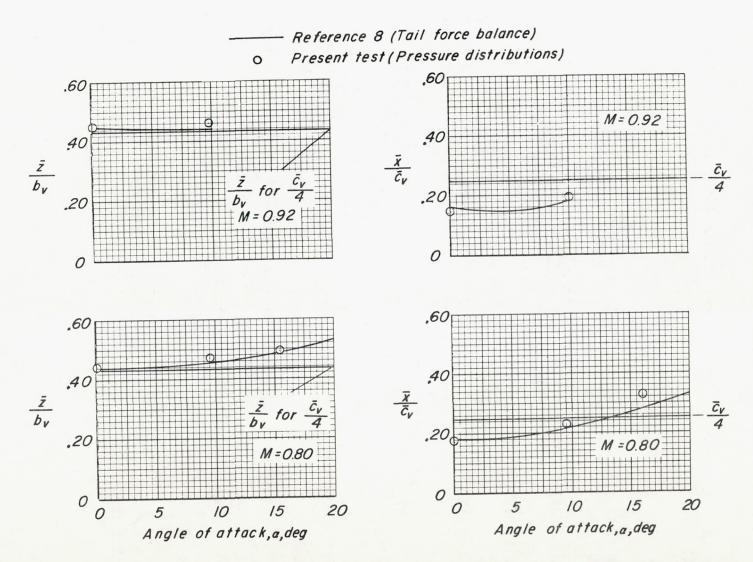
Angle of attack, a, deg

Figure 11.- Comparison of the location of the vertical-tail centers of pressure as obtained from force measurements and pressure measurements. $\Delta\beta \approx 8°$; $i_t = 0°$.

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SUMMARY

An investigation has been made in the Langley high-speed 7- by 10-foot tunnel of some effects of wing height and angle of attack on the vertical-tail pressure distribution of a complete model in sideslip at high subsonic speeds. The 45° swept wing had an aspect ratio of 4, a taper ratio of 0.30, and NACA 65A006 airfoil sections parallel to the model plane of symmetry. The wing was located in high, middle, and low positions on a circular fuselage. Tests were made with the horizontal tail off and with the horizontal tail on the fuselage center line. Test Mach numbers ranged from 0.60 to 0.92, which corresponds to a Reynolds number range from approximately 2.7×10^6 to 3.4×10^6 based on the wing mean aerodynamic chord. The sideslip angles varied from -3.9° to 12.7° at several selected angles of attack.

The results indicated that increased wing height progressively decreased the vertical-tail load for constant angles of attack and sideslip, the largest load reductions occurring in the region near the tail root chord. Increases in angle of attack for constant sideslip angle produced increases in the vertical-tail loads for the low- and mid-wing configurations over most of the tail span except in the region of the root chord where load reductions occurred. For the high-wing configuration, increases in angle of attack produced reductions in the tail loads over the complete span, the largest reduction occurring in the region near the tail root chord. Substantial movements in the spanwise centers of pressure toward the tip of the vertical tail were caused by increases in angle of attack for all wing-height configurations. Increases in angle of attack also caused rearward movements in the chordwise locations of the centers of pressure particularly in the region of the root chord. The effects of wing height on the center-of-pressure location were considerabl; smaller than the effects of angle of attack. The vertical-tail center-of-pressure locations as determined from previous tail-force tests and as determined from the present pressure measurements were generally in good agreement.

INTRODUCTION

The need for a better understanding of the nature of the air flow at the tail and its effects on the resulting vertical-tail loads has become increasingly important at the higher speeds of modern airplanes, inasmuch as several of these airplanes have, in maneuvering flight, been subjected to extreme vertical-tail loads while experiencing losses in directional stability. Of the many variables affecting the aerodynamic characteristics of a particular vertical-tail configuration, past research investigations (for example, refs. 1 to 3) have shown that wing height and fuselage cross-sectional shape are among the most important. Existing data on vertical-tail loads utilized configurations that are now outdated (ref. 4) or did not represent complete airplanes (ref. 5).

The purpose of this paper is to present the results of an experimental investigation made to determine the effects of changes in wing height on the chordwise pressure distributions and integrated loads on the vertical tail of a complete model in sideslip at high subsonic speeds. The complete model consisted of a 45° sweptback wing of aspect ratio 4.0 located in the low, middle, and high positions on a circular cross-section body in combination with a triangular horizontal tail of aspect ratio 4.0 and a vertical tail of aspect ratio 1.02 having a swept leading edge and an unswept trailing edge. The sideslip-angle range extended from -3.9° to 12.7° at angles of attack of approximately 0°, 9°, and 15°. The test Mach numbers varied from 0.60 to 0.92 with corresponding Reynolds number varying from 2.7×10^6 to 3.4×10^6.

COEFFICIENTS AND SYMBOLS

C_L lift coefficient, $\dfrac{\text{Lift}}{qS}$

S wing area, sq ft

C_p pressure coefficient, $\dfrac{p_l - p_o}{q}$

p_l local static pressure, lb/sq ft

p_o free-stream static pressure, lb/sq ft

q dynamic pressure, $\frac{1}{2}\rho V_o^2$, lb/sq ft

ρ air density, slugs/cu ft

V_O free-stream velocity, ft/sec

M Mach number

c_v local vertical-tail chord, ft

$\bar{c}_v$ vertical-tail mean aerodynamic chord, ft

$c_{v,av}$ average vertical-tail chord, ft

$\bar{c}$ wing mean aerodynamic chord, ft

b_v exposed vertical-tail span, ft

x chordwise distance from leading edge of vertical-tail local chord, ft

z spanwise distance from vertical-tail root chord (vertical-tail root chord 0.154 ft above fuselage center line), ft

$\Delta\left(\dfrac{x}{c_v}\right)$ increment of vertical-tail local chord over which the pressure at a particular orifice is assumed to act (distance between points midway between adjacent orifices)

$c_{n,v}$ vertical-tail section normal-force coefficient,

$$\sum_{\frac{x}{c_v}=0}^{1} \left(C_{p,L} - C_{p,R}\right)\Delta\left(\frac{x}{c_v}\right)$$

$c_{m,v}$ vertical-tail section moment coefficient about 0.25 local vertical-tail chord, $\displaystyle\sum_{\frac{x}{c_v}=0}^{1} \left(C_{p,L} - C_{p,R}\right)\left(\frac{\bar{x}}{c_v} - 0.25\right)\Delta\left(\frac{x}{c_v}\right)$

$\dfrac{\bar{x}}{c_v}$ local chordwise location of vertical-tail center of pressure, $\left(\dfrac{c_{m,v}}{c_{n,v}} - 0.25\right)$

$\dfrac{\bar{x}}{\bar{c}_v}$ chordwise center-of-pressure location of total vertical-tail
       load

$$\dfrac{\displaystyle\int_0^{1.0} \dfrac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\left(\dfrac{\bar{x}}{c_v}\dfrac{c_v}{\bar{c}_v}-0.25\right)d\!\left(\dfrac{z}{b_v}\right)}{\displaystyle\int_0^{1.0} \dfrac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\,d\!\left(\dfrac{z}{b_v}\right)}$$

$\dfrac{\bar{z}}{b_v}$ spanwise center-of-pressure location of total vertical-tail
       load

$$\dfrac{\displaystyle\int_0^{1.0} \dfrac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\dfrac{z}{b_v}\,d\!\left(\dfrac{z}{b_v}\right)}{\displaystyle\int_0^{1.0} \dfrac{\Delta c_n c_v}{\Delta\beta c_{v,av}}\,d\!\left(\dfrac{z}{b_v}\right)}$$

α angle of attack, deg

β angle of sideslip, deg

i_t horizontal-tail incidence angle, deg

Subscripts:

R   right side of vertical tail looking upstream

L   left side of vertical tail looking upstream

MODEL AND APPARATUS

Details of the complete model as tested are given in figure 1 and
a photograph of the model mounted on the sting-type support system is
given as figure 2. The fuselage was of fineness ratio 10.94 and was
constructed of aluminum. The physical characteristics of the fuselage
including afterbody ordinates are given in figure 1(c). The 45° swept
wing was made of aluminum and had an aspect ratio of 4, a taper ratio
of 0.30, and NACA 65A006 airfoil sections parallel to the model plane
of symmetry. The triangular horizontal tail ($i_t = 0°$) located on the
fuselage center line, was made of steel and covered with plastic and
fiberglass; it had an aspect ratio of 4 and NACA 65A006 airfoil sections.
The vertical tail was made of steel and covered with plastic and fiber-
glass; it had an aspect ratio of 1.02 (based on an exposed area and span),
a taper ratio of 0.46, quarter-chord sweep of 28.35°, and NACA 65A006 air-
foil sections. Twenty-three chordwise pressure orifices (in rows parallel

to fuselage center line) were located at each of the following spanwise locations: $0.11b_v$, $0.38b_v$, $0.66b_v$, and $0.93 b_v$. Details of the orifice locations are given in figure 1(b) and other details of model geometry are given in table I.

Tests were made on the sting-support system shown in figure 2. With this system the model can be remotely operated through an angle-of-attack range of approximately 26° in the plane of the vertical strut. By utilization of couplings in the sting behind the model, the model can be rolled 90° so that either angle of attack or angle of sideslip can be the remotely controlled variable. With the wings vertical, the couplings can be used to support the model at fixed angles of attack while the model is tested through the angle-of-sideslip range.

With the model at a given angle of attack and angle of sideslip, a record was taken of the pressures existing at the orifices by photographing a manometer board to which the orifices were connected. These pressures were reduced to coefficient form and are presented in tables II to VI.

TESTS AND CORRECTIONS

The tests were made in the Langley high-speed 7- by 10-foot tunnel through a Mach number range of 0.60 to 0.92, which corresponds to a Reynolds number range from approximately 2.7×10^6 to 3.4×10^6 based on the wing mean aerodynamic chord. Pressure measurements on the vertical tail were made through a sideslip-angle range from approximately -4° to 13° at angles of attack of 0°, 9°, and 15° with the horizontal tail ($i_t = 0°$) on the fuselage center line and the wing in low, middle, and high positions on a circular cross-section fuselage. Pressure measurements on the vertical tail were obtained for the mid-wing configuration with the horizontal tail removed and also with the fuselage with both horizontal and vertical tails.

Blockage corrections, calculated by the method of reference 6, were applied to Mach number and dynamic pressure. Jet-boundary corrections calculated by the method of reference 7 have been applied to the angle of attack. Corrections due to sting-support deflection have been applied to the angles of attack and sideslip.

PRESENTATION OF RESULTS

All the individual pressure coefficients are given in tables II to VI and some selected results from the investigation are presented in the

following figures:

DISCUSSION

Vertical-Tail Loads

For a given angle of attack, increases in the section load accompany
increases in the angle of sideslip for all wing positions, except at the
highest sideslip angle (12.7°) where some local load reductions are in
evidence. (See fig. 7.) The effect of changing the wing vertical loca-
tion on the fuselage from low to mid to high is, in general, to cause a
progressive reduction in the vertical-tail section loads for any of the
angles of attack and sideslip investigated. (See figs. 7 and 8.)

For the low- and mid-wing locations, increasing the angle of attack,
for a constant sideslip angle, generally causes increases in the vertical-
tail section loads over most of the span, although there are reductions in
the vicinity of the root chord. The largest increase in loading with
angle of attack occurs for the low-wing position. (See figs. 7 and 8.)
For the high-wing position increasing the angle of attack causes reductions
in the vertical-tail loads over the complete span, the largest reduction
again occurring in the region near the root chord. (See figs. 7 and 8.)

Increases in Mach number from 0.60 to 0.92, for small angles of sideslip, generally caused some increase in the vertical-tail section loads for given angles of attack, these increases being particularly noticeable near the tip chord of the tail for the higher Mach numbers. (See fig. 8.) For the conditions investigated, however, effects of Mach number appear to be of minor importance compared with the more pronounced effects of wing height and angle of attack.

From the foregoing discussion it is evident that the losses in vertical-tail loads due to raising the wing and to increasing the angle of attack would cause undesirable losses in directional stability, inasmuch as the largest load losses occur in the regions of the root chord which, because of taper, are the largest contributors to the total tail load.

Vertical-Tail Center of Pressure

The chordwise locations of local center of pressure as affected by wing height are presented in figure 9 for various model angles of attack and angles of sideslip. These data indicate no large effects of wing height. The individual configurations, however, are affected by angle of attack, inasmuch as the local chordwise center of pressure appears to move rearward for $z/b_V = 0.11$, especially for sideslip angles of 3.9°. Examination of the chordwise loadings presented in figure 6 indicates that the section normal force for $z/b_V = 0.11$ is small compared with that farther outboard. This small load is assumed to be the result of adverse sidewash velocities near the vertical-tail fuselage juncture. (See ref. 2.)

Increasing the angle of attack produced the largest effect on the spanwise location of the center of pressure and caused a movement toward the tip for each wing-height configuration as shown in figure 10. Changes in wing height produced only small movements of the spanwise centers of pressure as compared with the effects of angle of attack. The effect of the presence of the wing on the spanwise location of the vertical-tail center of pressure was negligible except for the highest angle of attack investigated (15.9°). (See figs. 10 and 11.)

The comparison between the spanwise and chordwise locations of the vertical-tail center of pressure as obtained from the present tests and those obtained by force measurements (ref. 8), presented in figure 11 for the wing-off configuration, indicates good agreement except in the case of the chordwise centers of pressure for a Mach number of 0.92. The cause for this latter disagreement is not fully understood.

CONCLUSIONS

The results of a wind-tunnel investigation at high subsonic speeds of the effects of wing height and angle of attack on the vertical-tail pressure distributions of a complete model having a 45° swept wing indicate the following conclusions:

1. Raising the wing progressively decreased the vertical-tail load for constant angles of attack and sideslip, the largest load reductions occurring in the region near the tail root chord.

2. Increases in angle of attack, for a constant sideslip angle, produced increases in the vertical-tail loads for the low- and mid-wing configurations over most of the tail span except in the region of the root chord where load reductions occurred. For the high-wing configuration increases in angle of attack produced reductions in the tail loads over the complete span, the largest reduction occurring in the region near the tail root chord.

3. Substantial movements in the spanwise centers of pressure toward the tip of the vertical tail were caused by increases in angle of attack for all wing-height configurations. Increases in angle of attack also caused rearward movements in the chordwise locations of the centers of pressure particularly in the region of the root chord. The effects of wing height on the center-of-pressure location were considerably smaller than the effects of angle of attack.

4. The vertical-tail center-of-pressure locations as determined from previous tail-force tests and as determined from the present pressure measurements were generally in good agreement.

Langley Aeronautical Laboratory,
    National Advisory Committee for Aeronautics,
        Langley Field, Va., April 3, 1957.

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TABLE I

PHYSICAL CHARACTERISTICS OF THE MODEL

Wing:
  Span, ft 3.00
  Root chord, ft 1.154
  Tip chord, ft 0.346
  Mean aerodynamic chord, ft 0.822
  Area, sq ft 2.25
  Aspect ratio 4.00
  Taper ratio 0.30
  Quarter-chord sweep, deg 45
  Airfoil section . . . . NACA 65A006

Horizontal tail:
  Span, ft 1.162
  Root chord, ft 0.581
  Tip chord, ft 0
  Mean aerodynamic chord, ft 0.388
  Area, sq ft 0.337
  Aspect ratio 4.00
  Taper ratio 0
  Quarter-chord sweep, deg 36.85
  Airfoil section . . . . NACA 65A006

Vertical tail:
  Span (measured from root chord), ft 0.683
  Root chord (located 0.154 ft above fuselage center line), ft . 0.912
  Tip chord, ft 0.420
  Mean aerodynamic chord, ft 0.696
  Area, sq ft 0.454
  Aspect ratio 1.02
  Taper ratio 0.46
  Quarter-chord sweep, deg 28.35
  Airfoil section . . . . NACA 65A006

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL

(a) $\alpha = 0°$; M = 0.60.

$\dfrac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.608	.649	.381	-.409	-1.289	.279	.728	.329	-.397	-1.179
.025		-.096	-.535	-1.146	-2.274	.328	-.087	-.612	-1.051	-1.163
.075		-.107	-.359	-.673	-1.524	.141	-.130	-.425	-.934	-1.105
.150	.033	-.134	-.293	-.477	-.724	.051	-.134	-.309	-.701	-1.055
.250	.012	-.098	-.194	-.298	-.420	-.013	-.144	-.259	-.425	-1.017
.350	-.026	-.107	-.162	-.232	-.341	-.067	-.162	-.244	-.325	-.864
.450	-.060	-.116	-.144	-.189	-.305	-.105	-.171	-.219	-.264	-.666
.550	-.071	-.103	-.112	-.149	-.307	-.094	-.139	-.160	-.198	-.497
.650	-.035	-.060	-.063	-.094	-.211	-.069	-.098	-.103	-.130	-.366
.750	-.010	-.019	-.015	-.049	-.132	-.026	-.040	-.038	-.065	-.247
.850	.028	.030	.028	.005	-.053	.026	.021	.028	.012	-.134
.900	.053	.057	.069	.035	-.010	.055	.060	.060	.053	-.073
Right side										
.025	-.453	-.033	.317	.607	.830	-.609	-.069	.372	.596	.738
.075	-.376	-.112	.127	.345	.546	-.417	-.123	.146	.360	.524
.150	-.290	-.125	.051	.213	.373	-.313	-.130	.051	.218	.364
.250	-.207	-.103	.019	.143	.262	-.254	-.141	-.013	.118	.233
.350	-.173	-.112	-.029	.064	.166	-.227	-.153	-.060	.046	.136
.450	-.159	-.125	-.065	.008	.089	-.211	-.157	-.090	-.011	.064
.550	-.110	-.103	-.065	-.011	.048	-.162	-.137	-.090	-.033	.021
.650	-.060	-.058	-.035	.003	.044	-.105	-.094	-.065	-.020	.012
.750	-.013	-.015	-.011	.019	.046	-.035	-.028	-.017	.010	.021
.850	.048	.046	.044	.062	.073	.046	.048	.062	.057	.046
.900	.069	.066	.069	.075	.073	.062	.060	.057	.062	.030
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.229	.719	.284	-.373	-.659	.398	.671	.236	-.275	-.492
.025	.335	-.119	-.773	-1.042	-.742	.193	-.198	-.678	-.825	-.528
.075	.148	-.144	-.474	-.945	-.724	.021	-.166	-.388	-.805	-.535
.150	.035	-.157	-.361	-.773	-.708	-.055	-.144	-.273	-.701	-.571
.250	-.028	-.159	-.284	-.526	-.672	-.031	-.089	-.194	-.540	-.560
.350	-.085	-.175	-.244	-.361	-.645					
.450	-.114	-.175	-.214	-.275	-.614					
.550	-.114	-.157	-.176	-.212	-.575					
.650	-.083	-.105	-.112	-.149	-.521	-.074	-.071	-.099	-.339	-.397
.750	-.044	-.046	-.054	-.081	-.458	-.055	-.040	-.072	-.316	-.377
.850	.019	.005	.008	-.024	-.384	-.031	-.006	-.054	-.280	-.350
.900	.030	.028	.030	.005	-.357	-.013	.017	-.047	-.248	-.337
Right side										
.025	-.710	-.078	.383	.587	.688	-.561	-.144	.234	.394	.461
.075	-.466	-.139	.143	.356	.485	-.378	-.180	.010	.134	.213
.150	-.324	-.144	.053	.211	.332	-.265	-.148	-.060	.003	.062
.250	-.277	-.162	-.029	.103	.199					
.350	-.252	-.173	-.081	.021	.091					
.450	-.209	-.173	-.108	-.026	.033					
.550	-.168	-.144	-.101	-.045	-.006					
.650	-.105	-.105	-.072	-.033	-.019	-.085	-.071	-.072	-.060	-.089
.750	-.046	-.033	-.029	-.008	-.039	-.074	-.046	-.060	-.067	-.125
.850	.028	.035	.041	.039	-.033	-.013	.012	-.006	-.022	-.114
.900	.060	.060	.062	.053	-.062	-.010	.035	.012	-.020	-.134

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(b) α = 0°; M = 0.80.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	z/bv = 0.11					z/bv = 0.38				
Left side										
.000	.715	.677	.553	.205	-.280	.472	.771	.535	.043	-.542
.025	.278	-.064	-.535	-1.094	-1.417	.330	-.078	-1.038	-1.716	-1.014
.075	.143	-.092	-.369	-.892	-1.207	.148	-.131	-.443	-1.513	-1.006
.150	.045	-.135	-.325	-.539	-1.002	.054	-.143	-.336	-.506	-.937
.250	.020	-.102	-.213	-.302	-.694	-.019	-.153	-.278	-.361	-.854
.350	-.025	-.110	-.175	-.234	-.456	-.078	-.177	-.260	-.325	-.743
.450	-.069	-.129	-.163	-.197	-.352	-.130	-.189	-.231	-.270	-.629
.550	-.086	-.117	-.127	-.141	-.268	-.107	-.153	-.165	-.179	-.512
.650	-.050	-.058	-.056	-.089	-.174	-.078	-.102	-.100	-.103	-.419
.750	-.018	-.016	-.003	-.036	-.123	-.030	-.030	-.026	-.024	-.316
.850	.034	.044	.056	.028	-.058	.043	.047	.051	.053	-.218
.900	.067	.079	.086	.066	-.012	.075	.083	.090	.090	-.156
Right side										
.025	-.414	.003	.331	.608	.831	-.965	-.048	.371	.607	.746
.075	-.392	-.096	.140	.360	.559	-.426	-.119	.157	.368	.535
.150	-.307	-.120	.056	.225	.390	-.332	-.131	.059	.228	.377
.250	-.215	-.102	.016	.158	.279	-.268	-.150	-.012	.126	.257
.350	-.186	-.119	-.032	.073	.178	-.242	-.167	-.068	.047	.157
.450	-.162	-.134	-.079	.008	.087	-.216	-.173	-.106	-.023	.060
.550	-.115	-.107	-.079	-.023	.044	-.166	-.147	-.110	-.045	.012
.650	-.048	-.055	-.047	-.004	.035	-.091	-.093	-.077	-.029	.001
.750	.001	-.013	-.014	.020	.030	-.012	-.013	-.011	.014	.006
.850	.070	.061	.051	.066	.060	.070	.071	.065	.070	.025
.900	.090	.086	.071	.082	.063	.093	.086	.075	.081	.006

$\frac{x}{c_v}$	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
	z/bv = 0.66					z/bv = 0.93				
Left side										
.000	.435	.759	.488	.008	-.382	.492	.704	.382	.000	-.322
.025	.340	-.117	-1.127	-1.437	-.710	.190	-.235	-.956	-.732	-.470
.075	.152	-.150	-.544	-1.249	-.707	-.001	-.201	-.511	-.734	-.482
.150	.040	-.171	-.399	-.960	-.710	-.075	-.156	-.293	-.716	-.512
.250	-.042	-.176	-.305	-.716	-.674	-.036	-.083	-.210	-.602	-.502
.350	-.095	-.189	-.257	-.480	-.621					
.450	-.136	-.195	-.230	-.305	-.567					
.550	-.125	-.168	-.178	-.196	-.518					
.650	-.091	-.105	-.109	-.106	-.464	-.072	-.054	-.097	-.349	-.382
.750	-.039	-.030	-.035	-.033	-.410	-.037	-.015	-.056	-.304	-.357
.850	.028	.038	.035	.028	-.355	-.009	.029	-.033	-.250	-.323
.900	.049	.065	.063	.053	-.328	.011	.052	-.020	-.219	-.314
Right side										
.025	-1.185	-.058	.383	.599	.705	-1.043	-.150	.220	.397	.500
.075	-.500	-.138	.158	.360	.512	-.461	-.205	-.017	.123	.248
.150	-.347	-.143	.057	.224	.358	-.267	-.159	-.077	-.013	.078
.250	-.295	-.174	-.036	.102	.228					
.350	-.262	-.188	-.097	.011	.106					
.450	-.223	-.183	-.130	-.042	.027					
.550	-.169	-.159	-.123	-.059	-.020					
.650	-.100	-.101	-.082	-.041	-.038	-.068	-.052	-.064	-.068	-.086
.750	-.024	-.018	-.029	-.004	-.046	-.053	-.019	-.042	-.056	-.124
.850	.061	.068	.056	.058	-.033	.013	.052	.022	-.001	-.104
.900	.096	.095	.080	.076	-.065	.016	.068	.035	-.013	-.126

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE, AND VERTICAL TAIL - Continued

(c) $\alpha = 0°$; $M = 0.85$.

$\frac{x}{c_v}$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.746	.687	.604	.323	-.110	.528	.781	.595	.189	-.359
.025	.282	-.054	-.518	-1.055	-1.175	.336	-.071	-1.068	-1.531	-.841
.075	.152	-.088	-.376	-.867	-1.034	.152	-.129	-.415	-1.370	-.856
.150	.050	-.138	-.348	-.686	-.912	.059	-.146	-.352	-1.202	-.825
.250	.022	-.107	-.229	-.357	-.749	-.022	-.163	-.303	-.330	-.796
.350	-.028	-.120	-.192	-.249	-.585	-.088	-.189	-.284	-.276	-.733
.450	-.080	-.142	-.175	-.197	-.471	-.146	-.211	-.249	-.266	-.644
.550	-.101	-.129	-.134	-.149	-.366	-.129	-.170	-.179	-.190	-.551
.650	-.062	-.073	-.065	-.096	-.249	-.095	-.111	-.107	-.107	-.461
.750	-.025	-.019	-.002	-.032	-.177	-.040	-.040	-.031	-.026	-.365
.850	.036	.049	.059	.035	-.087	.040	.047	.056	.059	-.267
.900	.069	.084	.090	.072	-.034	.083	.088	.100	.098	-.211
Right side										
.025	-.381	.006	.335	.609	.842	-.981	-.059	.366	.595	.753
.075	-.400	-.101	.141	.360	.566	-.448	-.127	.152	.358	.542
.150	-.323	-.128	.054	.220	.396	-.344	-.144	.053	.220	.381
.250	-.224	-.111	.016	.146	.285	-.284	-.158	-.022	.118	.257
.350	-.199	-.131	-.042	.063	.175	-.260	-.180	-.086	.032	.152
.450	-.170	-.145	-.092	-.006	.080	-.231	-.193	-.130	-.043	.049
.550	-.117	-.121	-.099	-.036	.030	-.170	-.161	-.134	-.073	-.008
.650	-.052	-.064	-.058	-.022	.019	-.093	-.103	-.093	-.049	-.024
.750	.002	-.019	-.022	.002	.008	-.009	-.016	-.024	-.004	-.017
.850	.071	.060	.049	.059	.039	.076	.070	.059	.062	.000
.900	.100	.090	.066	.073	.042	.101	.088	.076	.078	-.021

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.484	.766	.547	.140	-.266	.516	.709	.435	.045	-.270
.025	.339	-.110	-1.167	-1.622	-.747	.179	-.251	-1.090	-.840	-.474
.075	.155	-.153	-.594	-1.441	-.759	-.022	-.231	-.596	-.821	-.497
.150	.036	-.183	-.415	-1.082	-.766	-.098	-.176	-.287	-.800	-.539
.250	-.053	-.192	-.313	-.680	-.701	-.045	-.098	-.208	-.673	-.521
.350	-.118	-.214	-.270	-.481	-.628					
.450	-.158	-.216	-.245	-.300	-.593					
.550	-.151	-.187	-.189	-.189	-.545					
.650	-.103	-.117	-.110	-.099	-.488	-.074	-.064	-.100	-.380	-.407
.750	-.042	-.035	-.032	-.025	-.430	-.039	-.020	-.062	-.327	-.382
.850	.029	.032	.039	.032	-.366	-.001	.029	-.036	-.270	-.352
.900	.053	.062	.070	.059	-.339	.019	.050	-.018	-.234	-.339
Right side										
.025	-1.153	-.076	.373	.589	.714	-1.151	-.172	.207	.388	.503
.075	-.409	-.152	.149	.348	.513	-.586	-.237	-.048	.099	.248
.150	-.366	-.155	.046	.212	.359	-.231	-.177	-.109	-.056	.061
.250	-.309	-.193	-.049	.085	.221					
.350	-.274	-.216	-.120	-.011	.095					
.450	-.230	-.207	-.157	-.075	.008					
.550	-.175	-.170	-.148	-.089	-.046					
.650	-.098	-.103	-.099	-.062	-.060	-.073	-.060	-.069	-.079	-.111
.750	-.018	-.029	-.032	-.022	-.070	-.056	-.026	-.049	-.069	-.143
.850	.070	.064	.056	.049	-.055	.011	.056	.025	-.012	-.119
.900	.104	.095	.080	.075	-.085	.023	.070	.034	-.016	-.144

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(d) $\alpha = 0°$; M = 0.90.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.776	.695	.654	.430	.028	.588	.794	.655	.313	-.198
.025	.289	-.046	-.469	-.969	-1.040	.330	-.070	-.973	-1.333	-.725
.075	.156	-.083	-.390	-.791	-.923	.150	-.142	-.668	-1.206	-.730
.150	.046	-.142	-.370	-.684	-.814	.048	-.161	-.374	-1.100	-.737
.250	.010	-.115	-.239	-.525	-.691	-.031	-.182	-.311	-.660	-.723
.350	-.039	-.134	-.200	-.336	-.586	-.109	-.217	-.319	-.449	-.677
.450	-.105	-.159	-.190	-.258	-.517	-.181	-.251	-.252	-.245	-.616
.550	-.141	-.153	-.133	-.146	-.436	-.173	-.198	-.167	-.112	-.545
.650	-.083	-.087	-.059	-.070	-.330	-.129	-.125	-.092	-.056	-.473
.750	-.041	-.025	.011	-.009	-.233	-.052	-.035	-.012	.006	-.392
.850	.026	.051	.075	.051	-.137	.038	.051	.074	.083	-.316
.900	.062	.087	.101	.086	-.081	.084	.098	.115	.119	-.261
Right side										
.025	-.354	.021	.353	.617	.847	-.927	-.055	.373	.593	.759
.075	-.417	-.094	.159	.364	.576	-.654	-.135	.167	.357	.544
.150	-.365	-.131	.064	.226	.410	-.392	-.174	.048	.198	.253
.250	-.259	-.121	.026	.149	.295	-.313	-.174	-.014	.111	.265
.350	-.233	-.143	-.041	.057	.186	-.314	-.209	-.087	.023	.157
.450	-.203	-.165	-.096	-.017	.081	-.256	-.227	-.148	-.062	.044
.550	-.131	-.142	-.112	-.058	.023	-.179	-.187	-.153	-.101	-.017
.650	-.051	-.073	-.063	-.038	.004	-.093	-.118	-.103	-.080	-.040
.750	.004	-.022	-.025	-.021	-.012	-.006	-.014	-.022	-.014	-.034
.850	.074	.062	.050	.054	.016	.084	.079	.070	.058	-.017
.900	.100	.091	.075	.069	.011	.110	.094	.088	.078	-.052
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.535	.775	.609	.268	-.138	.551	.710	.492	.141	-.193
.025	.331	-.113	-1.087	-1.413	-.734	.152	-.269	-1.048	-1.373	-.460
.075	.146	-.166	-.903	-1.274	-.750	-.063	-.323	-.989	-1.317	-.479
.150	.021	-.203	-.390	-1.190	-.753	-.195	-.206	-.370	-1.110	-.523
.250	-.076	-.227	-.390	-1.118	-.698	-.080	-.113	-.143	-.743	-.509
.350	-.161	-.254	-.223	-.639	-.596					
.450	-.211	-.253	-.230	-.206	-.574					
.550	-.193	-.213	-.182	-.064	-.545					
.650	-.128	-.119	-.097	-.032	-.508	-.081	-.061	-.089	-.312	-.409
.750	-.031	-.031	-.014	.011	-.459	-.041	-.015	-.049	-.288	-.387
.850	.026	.043	.059	.059	-.398	.000	.037	-.025	-.249	-.359
.900	.054	.074	.088	.085	-.378	.024	.055	-.009	-.216	-.352
Right side										
.025	-1.051	-.074	.381	.577	.719	-1.085	-.193	.206	.378	.513
.075	-.854	-.169	.157	.342	.512	-.979	-.327	-.069	.087	.258
.150	-.368	-.169	.055	.205	.363	-.385	-.214	-.176	-.109	.063
.250	-.394	-.217	-.057	.071	.227					
.350	-.265	-.257	-.140	-.036	.092					
.450	-.231	-.245	-.182	-.117	-.007					
.550	-.179	-.198	-.163	-.130	-.066					
.650	-.099	-.114	-.108	-.093	-.082	-.071	-.069	-.067	-.077	-.132
.750	-.018	-.025	-.026	-.034	-.093	-.059	-.019	-.039	-.062	-.159
.850	.078	.073	.071	.055	-.066	.008	.062	.036	.007	-.124
.900	.112	.103	.095	.081	-.094	.018	.074	.050	.003	-.153

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(e) $\alpha = 0^{\circ}$; M = 0.92.

$\frac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9^{\circ}$	$\beta=0^{\circ}$	$\beta=3.9^{\circ}$	$\beta=7.9^{\circ}$	$\beta=12.7^{\circ}$	$\beta=-3.9^{\circ}$	$\beta=0^{\circ}$	$\beta=3.9^{\circ}$	$\beta=7.9^{\circ}$	$\beta=12.7^{\circ}$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.780	.694	.668	.467	.075	.604	.796	.676	.353	-.144
.025	.282	-.039	-.450	-.929	-1.013	.322	-.064	-.928	-1.245	-.763
.075	.153	-.080	-.380	-.755	-.897	.143	-.137	-.701	-1.126	-.735
.150	.039	-.143	-.370	-.664	-.793	.043	-.159	-.410	-1.040	-.717
.250	.008	-.119	-.257	-.556	-.674	-.035	-.182	-.316	-.716	-.700
.350	-.042	-.138	-.212	-.370	-.574	-.114	-.226	-.338	-.520	-.667
.450	-.112	-.176	-.205	-.298	-.515	-.194	-.277	-.296	-.380	-.614
.550	-.150	-.178	-.139	-.168	-.455	-.188	-.217	-.164	-.103	-.553
.650	-.091	-.094	-.058	-.070	-.361	-.131	-.127	-.092	-.032	-.493
.750	-.044	-.020	.009	-.001	-.095	-.055	-.026	-.011	.027	-.420
.850	.028	.056	.073	.059	-.176	.039	.061	.078	.092	-.349
.900	.060	.095	.102	.096	-.112	.085	.109	.118	.129	-.283
Right side										
.025	-.334	.029	.346	.620	.852	-.869	-.045	.362	.595	.760
.075	-.406	-.088	.151	.368	.580	-.650	-.132	.156	.364	.545
.150	-.368	-.127	.061	.234	.414			.225	.230	.227
.250	-.267	-.119	.018	.155	.297	-.314	-.176	-.024	.114	.259
.350	-.230	-.147	-.051	.062	.185	-.326	-.218	-.096	.019	.151
.450	-.211	-.176	-.104	-.020	.080	-.281	-.257	-.165	-.065	.042
.550	-.133	-.160	-.126	-.060	.016	-.169	-.209	-.177	-.108	-.026
.650	-.055	-.077	-.079	-.043	-.005	-.086	-.119	-.118	-.086	-.057
.750	.004	-.018	-.037	-.022	-.029	.001	-.008	-.026	-.020	-.050
.850	.079	.070	.048	.053	-.001	.087	.082	.067	.063	-.037
.900	.101	.096	.069	.078	-.008	.113	.102	.086	.081	-.062

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9^{\circ}$	$\beta=0^{\circ}$	$\beta=3.9^{\circ}$	$\beta=7.9^{\circ}$	$\beta=12.7^{\circ}$	$\beta=-3.9^{\circ}$	$\beta=0^{\circ}$	$\beta=3.9^{\circ}$	$\beta=7.9^{\circ}$	$\beta=12.7^{\circ}$
Left side										
.000	.550	.780	.629	.310	-.091	.562	.714	.514	.185	-.165
.025	.325	-.107	-1.036	-1.320	-.679	.140	-.284	-1.007	-1.279	-.507
.075	.139	-.161	-.878	-1.198	-.687	-.061	-.304	-.969	-1.201	-.473
.150	.018	-.204	-.581	-1.113	-.699	-.224	-.237	-.659	-1.015	-.515
.250	-.078	-.234	-.387	-1.068	-.683	-.111	-.107	-.138	-.900	-.510
.350	-.175	-.299	-.340	-.937	-.625					
.450	-.228	-.274	-.197	-.356	-.562					
.550	-.204	-.220	-.177	-.047	-.526					
.650	-.135	-.108	-.095	.003	-.494	-.081	-.050	-.089	-.436	-.421
.750	-.023	-.023	-.011	.040	-.451	-.039	-.006	-.051	-.351	-.399
.850	.029	.055	.063	.079	-.406	.005	.047	-.020	-.279	-.371
.900	.060	.086	.092	.098	-.372	.025	.069	-.007	-.234	-.355
Right side										
.025	-1.002	-.064	.365	.581	.719	-1.042	-.181	.194	.386	.515
.075	-.831	-.158	.145	.346	.516	-.963	-.361	-.083	.095	.258
.150	-.458	-.164	.046	.212	.364	-.585	-.237	-.215	-.108	.057
.250	-.387	-.227	-.066	.074	.225					
.350	-.362	-.301	-.163	-.041	.089					
.450	-.192	-.259	-.215	-.128	-.015					
.550	-.173	-.202	-.184	-.153	-.081					
.650	-.097	-.101	-.118	-.102	-.096	-.085	-.055	-.068	-.075	-.123
.750	-.010	-.017	-.026	-.035	-.106	-.060	-.010	-.037	-.064	-.174
.850	.081	.083	.071	.063	-.076	.009	.069	.039	.011	-.134
.900	.111	.114	.094	.095	-.106	.018	.089	.051	.014	-.156

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(f) $\alpha = 9.4^{\circ}$; M = 0.60.

$\dfrac{x}{c_v}$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.516	.818	.499	-.393	-1.164	.227	.667	.185	-.695	-1.564
.025	-.028	-.134	-.236	-.769	-2.346	.291	-.124	-.673	-1.141	-1.424
.075	.004	-.134	-.298	-.507	-.874	.121	-.154	-.453	-.976	-1.417
.150	-.023	-.152	-.298	-.407	-.608	.036	-.150	-.334	-.753	-1.479
.250	-.021	-.104	-.215	-.286	-.396	-.021	-.159	-.277	-.407	-1.146
.350	-.042	-.104	-.193	-.237	-.336	-.067	-.168	-.259	-.302	-.598
.450	-.058	-.115	-.181	-.207	-.318	-.095	-.179	-.236	-.262	-.364
.550	-.060	-.113	-.156	-.169	-.249	-.090	-.143	-.181	-.197	-.284
.650	-.035	-.065	-.099	-.132	-.166	-.067	-.104	-.128	-.137	-.215
.750	-.003	-.030	-.048	-.053	-.095	-.026	-.046	-.069	-.074	-.129
.850	.027	.022	.007	.005	-.026	.032	.018	.011	.005	-.056
.900	.055	.055	.039	.040	.020	.057	.050	.048	.042	-.017
	Right side									
.025	-.179	-.021	-.007	.531	.799	-.599	-.040	.352	.568	.687
.075	-.294	-.115	.007	.317	.528	-.402	-.111	.153	.347	.526
.150	-.273	-.122	-.003	.207	.374	-.312	-.127	.057	.214	.376
.250	-.207	-.097	-.014	.149	.275	-.257	-.136	-.003	.124	.257
.350	-.191	-.106	-.044	.077	.183	-.232	-.147	-.048	.059	.167
.450	-.177	-.118	-.062	.021	.114	-.214	-.154	-.080	.000	.091
.550	-.143	-.101	-.058	.003	.075	-.175	-.134	-.083	-.023	.054
.650	-.092	-.062	-.032	.014	.066	-.118	-.090	-.055	-.013	.041
.750	-.042	-.019	-.003	.028	.066	-.046	-.030	-.012	.017	.050
.850	.020	.038	.041	.070	.089	.032	.043	.043	.061	.073
.900	.052	.059	.057	.084	.096	.055	.052	.052	.061	.061
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.132	.628	.059	-.751	-1.047	.247	.568	-.003		-.798
.025	.293	-.163	-.920	-1.255	-.907	.151	-.234	-.806	-.707	-.594
.075	.126	-.170	-.511	-1.200	-.902	-.017	-.195	-.428	-.695	-.603
.150	.027	-.177	-.385	-1.083	-.874	-.092	-.168	-.286	-.646	-.603
.250	-.037	-.173	-.302	-.700	-.838	-.067	-.106	-.199	-.541	-.571
.350	-.088	-.179	-.266	-.325	-.812					
.450	-.115	-.182	-.234	-.225	-.762					
.550	-.118	-.163	-.197	-.197	-.677					
.650	-.090	-.108	-.133	-.130	-.552	-.083	-.062	-.110	-.332	-.449
.750	-.040	-.051	-.069	-.074	-.428	-.062	-.023	-.078	-.288	-.424
.850	-.001	-.003	-.012	-.023	-.318	-.030	.004	-.060	-.244	-.398
.900	.022	.025	.020	.003	-.261	-.005	.022	-.035	-.211	-.382
	Right side									
.025	-.705	-.060	.364	.554	.631	-.604	-.131	.196	.317	.397
.075	-.459	-.124	.146	.338	.484	-.386	-.182	-.021	.080	.197
.150	-.329	-.134	.050	.203	.346	-.260	-.166	-.101	-.048	.031
.250	-.285	-.159	-.028	.093	.222					
.350	-.257	-.166	-.080	.019	.126					
.450	-.216	-.163	-.108	-.023	.061					
.550	-.177	-.145	-.106	-.041	.018					
.650	-.118	-.111	-.078	-.034	.006	-.079	-.067	-.074	-.097	-.104
.750	-.058	-.033	-.039	-.013	-.005	-.067	-.037	-.078	-.097	-.139
.850	.025	.032	.032	.038	.006	-.010	.022	-.014	-.041	-.132
.900	.045	.055	.048	.054	-.003	-.001	.034	-.007	-.044	-.162

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(g) $\alpha = 9.6°$; $M = 0.80$.

$z/b_v = 0.11$ and $z/b_v = 0.38$

$\frac{x}{c_v}$	Cp for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	Cp for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.466	.864	.474	-.119	-.126	.414	.704	.442	-.127	-.781
.025	-.064	-.116	-.154	-.593	-1.593	.288	-.125	-.900	-1.389	-1.149
.075	-.003	-.128	-.313	-.496	-1.051	.119	-.160	-.482	-1.160	-1.095
.150	-.035	-.169	-.346	-.483	-.648	.036	-.152	-.362	-.813	-1.054
.250	-.031	-.119	-.244	-.348	-.445	-.028	-.166	-.307	-.442	-.968
.350	-.055	-.122	-.215	-.281	-.368	-.081	-.192	-.287	-.351	-.790
.450	-.076	-.139	-.208	-.243	-.349	-.124	-.207	-.256	-.296	-.579
.550	-.080	-.136	-.176	-.198	-.283	-.110	-.161	-.192	-.221	-.426
.650	-.046	-.076	-.105	-.145	-.216	-.083	-.111	-.130	-.154	-.338
.750	-.014	-.031	-.052	-.060	-.139	-.038	-.042	-.055	-.070	-.249
.850	.032	.031	.013	.012	-.060	.035	.036	.030	.012	-.149
.900	.065	.070	.056	.053	-.014	.068	.077	.074	.061	-.091
Right side										
.025	-.083	.005	-.038	.481	.754	-.881	-.028	.347	.568	.703
.075	-.308	-.111	.000	.311	.511	-.427	-.108	.143	.350	.520
.150	-.310	-.130	-.012	.213	.363	-.337	-.126	.053	.222	.377
.250	-.233	-.102	-.031	.152	.268	-.285	-.142	-.011	.125	.259
.350	-.218	-.122	-.058	.076	.175	-.262	-.164	-.063	.056	.170
.450	-.208	-.139	-.076	.012	.091	-.241	-.175	-.101	-.019	.076
.550	-.165	-.122	-.072	-.014	.050	-.192	-.154	-.108	-.041	.036
.650	-.098	-.069	-.043	.000	.039	-.121	-.101	-.070	-.029	.012
.750	-.047	-.028	-.011	.015	.035	-.040	-.023	-.015	.009	.018
.850	.033	.042	.050	.064	.059	.047	.055	.058	.059	.039
.900	.061	.066	.065	.082	.062	.074	.070	.070	.070	.024

$z/b_v = 0.66$ and $z/b_v = 0.93$

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.318	.657	.314	-.200	-.669	.335	.584	.193	-.188	-.607
.025	.297	-.172	-1.098	-1.035	-.776	.146	-.292	-.848	-.622	-.539
.075	.126	-.181	-.628	-.996	-.755	-.041	-.245	-.627	-.618	-.542
.150	.020	-.201	-.422	-.962	-.726	-.125	-.184	-.336	-.616	-.549
.250	-.052	-.192	-.320	-.868	-.700	-.073	-.107	-.227	-.551	-.546
.350	-.108	-.204	-.275	-.650	-.675					
.450	-.140	-.205	-.244	-.424	-.649					
.550	-.145	-.181	-.197	-.270	-.607					
.650	-.098	-.110	-.125	-.154	-.562	-.081	-.046	-.101	-.336	-.446
.750	-.041	-.037	-.049	-.061	-.501	-.043	-.001	-.066	-.297	-.420
.850	.013	.024	.017	.003	-.425	-.002	.039	-.023	-.250	-.390
.900	.039	.054	.052	.036	-.384	.018	.058	-.005	-.221	-.371
Right side										
.025	-.977	-.043	.353	.558	.656	-.948	-.146	.184	.346	.429
.075	-.514	-.123	.146	.347	.499	-.479	-.213	-.047	.100	.222
.150	-.365	-.137	.055	.218	.358	-.272	-.178	-.131	-.055	.039
.250	-.310	-.172	-.038	.100	.229					
.350	-.279	-.189	-.096	.012	.120					
.450	-.237	-.190	-.133	-.043	.042					
.550	-.188	-.164	-.127	-.064	-.010					
.650	-.115	-.114	-.084	-.047	-.030	-.052	-.049	-.067	-.080	-.123
.750	-.043	-.025	-.034	-.020	-.040	-.049	-.016	-.055	-.087	-.164
.850	.044	.051	.042	.042	-.033	.013	.051	.012	-.043	-.150
.900	.073	.077	.065	.061	-.057	.024	.069	.029	-.051	-.176

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(h) $\alpha = 9.7^\circ$; M = 0.85.

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
Left side										
.000	.436	.875	.448	-.129	.009	.478	.707	.510	.013	-.636
.025	-.101	-.116	-.108	-.543	-1.459	.273	-.130	-1.103	-1.524	-1.167
.075	-.019	-.142	-.314	-.505	-1.022	.105	-.174	-.472	-1.188	-1.107
.150	-.051	-.191	-.389	-.547	-.711	.022	-.174	-.387	-.752	-1.041
.250	-.048	-.137	-.271	-.415	-.493	-.042	-.190	-.337	-.520	-.920
.350	-.075	-.140	-.242	-.328	-.402	-.099	-.224	-.320	-.387	-.774
.450	-.099	-.173	-.232	-.274	-.380	-.146	-.242	-.287	-.340	-.621
.550	-.101	-.163	-.202	-.222	-.319	-.134	-.194	-.214	-.251	-.507
.650	-.062	-.101	-.122	-.159	-.251	-.102	-.135	-.144	-.165	-.415
.750	-.025	-.048	-.063	-.070	-.179	-.051	-.061	-.063	-.080	-.318
.850	.025	.022	.011	.009	-.094	.032	.028	.025	.009	-.208
.900	.062	.062	.053	.049	-.048	.070	.070	.068	.055	-.144
Right side										
.025	-.035	-.003	-.066	.447	.749	-.875	-.068	.336	.561	.700
.075	-.309	-.136	-.016	.301	.501	-.462	-.139	.131	.341	.511
.150	-.336	-.157	-.022	.203	.354	-.355	-.154	.040	.212	.372
.250	-.253	-.128	-.043	.144	.258	-.308	-.171	-.033	.113	.250
.350	-.241	-.146	-.072	.061	.158	-.291	-.197	-.082	.041	.157
.450	-.231	-.166	-.098	-.008	.080	-.265	-.208	-.126	-.040	.061
.550	-.181	-.150	-.093	-.035	.035	-.208	-.185	-.131	-.068	.012
.650	-.109	-.094	-.059	-.022	.023	-.128	-.126	-.093	-.056	-.009
.750	-.049	-.050	-.028	-.005	.015	-.038	-.043	-.035	-.012	-.007
.850	.030	.033	.040	.048	.041	.050	.048	.046	.047	.015
.900	.062	.057	.060	.069	.036	.080	.066	.060	.058	-.006

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
Left side										
.000	.376	.662	.378	-.112	-.601	.366	.589	.233	-.235	-.578
.025	.280	-.176	-1.221	-1.566	-.784	.125	-.297	-1.086	-.817	-.556
.075	.115	-.200	-.628	-1.347	-.756	-.071	-.314	-.675	-.793	-.550
.150	.006	-.221	-.472	-1.054	-.720	-.158	-.217	-.370	-.784	-.559
.250	-.072	-.228	-.360	-.783	-.695	-.094	-.135	-.230	-.660	-.563
.350	-.134	-.242	-.298	-.520	-.675					
.450	-.169	-.239	-.265	-.337	-.650					
.550	-.162	-.212	-.218	-.231	-.612	-.087	-.067	-.105	-.356	-.473
.650	-.118	-.129	-.139	-.136	-.578	-.045	-.019	-.063	-.300	-.447
.750	-.049	-.053	-.059	-.053	-.528	-.002	.031	-.025	-.239	-.419
.850	.013	.017	.014	.009	-.454					
.900	.045	.048	.046	.044	-.425	.020	.046	-.002	-.203	-.405
Right side										
.025	-1.129	-.077	.342	.545	.652	-1.141	-.193	.176	.341	.428
.075	-.472	-.157	.143	.334	.488	-.557	-.279	-.071	.085	.218
.150	-.395	-.168	.046	.207	.348	-.273	-.218	-.164	-.099	.022
.250	-.343	-.208	-.053	.084	.218					
.350	-.293	-.229	-.118	-.008	.107					
.450	-.251	-.228	-.157	-.073	.020					
.550	-.198	-.195	-.148	-.096	-.039					
.650	-.121	-.120	-.099	-.076	-.061	-.071	-.067	-.078	-.101	-.154
.750	-.041	-.048	-.046	-.041	-.071	-.052	-.027	-.058	-.104	-.196
.850	.050	.048	.044	.034	-.059	.016	.049	.015	-.047	-.176
.900	.088	.076	.070	.051	-.090	.033	.062	.027	-.045	-.205

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(i) $\alpha = 9.7°$; M = 0.90.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.340	.896	.413			.528	.714	.568		
.025	-.140	-.108	.000			.266	-.136	-1.001		
.075	-.030	-.140	-.239			.096	-.186	-.484		
.150	-.050	-.207	-.397			.011	-.191	-.466		
.250	-.057	-.151	-.291			-.052	-.207	-.353		
.350	-.092	-.159	-.264			-.116	-.240	-.369		
.450	-.117	-.196	-.266			-.183	-.282	-.343		
.550	-.119	-.196	-.223			-.171	-.220	-.210		
.650	-.077	-.111	-.121			-.131	-.140	-.136		
.750	-.042	-.052	-.055			-.061	-.057	-.054		
.850	.019	.020	.020			.026	.036	.039		
.900	.058	.067	.062			.073	.083	.088		
Right side										
.025	.075	.007	-.087			-.852	-.060	.330		
.075	-.256	-.140	-.009			-.534	-.144	.129		
.150	-.378	-.170	-.011			-.425	-.159	.034		
.250	-.293	-.141	-.039			-.339	-.183	-.031		
.350	-.277	-.163	-.073			-.359	-.219	-.089		
.450	-.277	-.187	-.102			-.334	-.244	-.144		
.550	-.214	-.175	-.099			-.213	-.214	-.153		
.650	-.120	-.104	-.069			-.132	-.143	-.105		
.750	-.056	-.051	-.031			-.038	-.033	-.036		
.850	.032	.034	.038			.053	.055	.047		
.900	.065	.064	.063			.088	.075	.065		
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.413	.667	.440			.382	.590	.280		
.025	.272	-.183	-1.157			.113	-.347	-1.153		
.075	.105	-.211	-.965			-.089	-.330	-1.073		
.150	-.009	-.234	-.459			-.295	-.258	-.502		
.250	-.092	-.251	-.443			-.140	-.121	-.156		
.350	-.170	-.294	-.385							
.450	-.221	-.274	-.229							
.550	-.202	-.227	-.202							
.650	-.136	-.127	-.126			-.093	-.051	-.105		
.750	-.056	-.043	-.043			-.046	-.002	-.062		
.850	.011	.028	.028			-.001	.046	-.019		
.900	.045	.060	.065			.025	.062	.000		
Right side										
.025	-1.070	-.073	.333			-1.131	-.196	.166		
.075	-.809	-.164	.128			-.972	-.338	-.091		
.150	-.407	-.178	.034			-.391	-.256	-.285		
.250	-.434	-.234	-.067							
.350	-.409	-.274	-.148							
.450	-.232	-.259	-.198							
.550	-.198	-.220	-.175							
.650	-.119	-.109	-.114			-.076	-.059	-.079		
.750	-.038	-.037	-.047			-.052	-.014	-.059		
.850	.056	.059	.046			.018	.063	.020		
.900	.091	.090	.071			.037	.079	.035		

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(j) $\alpha = 15.6°$; $M = 0.60$.

$\frac{x}{c_v}$	\multicolumn{5}{c}{C_p for–}	\multicolumn{5}{c}{C_p for–}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	\multicolumn{5}{c}{$z/b_v = 0.11$}	\multicolumn{5}{c}{$z/b_v = 0.38$}
	Left side
.000	.785	.809	.833	.170	-.458	.235	.676	.293	-.651	-1.487
.025	-.303	-.160	-.176	-.412	-1.149	.316	-.124	-.673	-1.487	-1.941
.075	-.141	-.155	-.267	-.385	-.739	.127	-.158	-.471	-.974	-2.067
.150	-.109	-.182	-.303	-.390	-.566	.030	-.164	-.353	-.507	-1.473
.250	-.073	-.135	-.222	-.290	-.394	-.039	-.167	-.297	-.401	-.521
.350	-.087	-.133	-.199	-.252	-.333	-.089	-.182	-.281	-.351	-.498
.450	-.105	-.144	-.194	-.232	-.297	-.123	-.191	-.260	-.304	-.435
.550	-.107	-.133	-.165	-.182	-.253	-.107	-.164	-.206	-.236	-.340
.650	-.073	-.088	-.108	-.121	-.185	-.087	-.124	-.147	-.175	-.267
.750	-.046	-.050	-.061	-.069	-.124	-.055	-.070	-.086	-.107	-.188
.850	-.003	.004	-.002	-.004	-.054	.006	.002	-.009	-.042	-.104
.900	.026	.037	.032	.033	-.011	.037	.033	.028	-.008	-.058
	Right side
.025	-.071	-.077	-.201	.245	.561	-.686	-.102	.345	.536	.699
.075	-.285	-.164	-.156	.109	.298	-.472	-.155	.125	.308	.477
.150	-.296	-.169	-.104	.062	.157	-.361	-.162	.035	.173	.307
.250	-.235	-.133	-.074	.035	.089	-.296	-.167	-.027	.078	.175
.350	-.220	-.135	-.088	-.006	.030	-.274	-.171	-.074	.005	.089
.450	-.208	-.151	-.106	-.044	-.015	-.253	-.182	-.104	-.055	.010
.550	-.166	-.126	-.097	-.049	-.024	-.211	-.162	-.104	-.076	-.031
.650	-.114	-.086	-.067	-.033	-.015	-.152	-.115	-.077	-.062	-.038
.750	-.060	-.043	-.036	-.008	-.002	-.078	-.054	-.029	-.033	-.029
.850	.008	.020	.014	.033	.028	.003	.017	.030	.014	.005
.900	.028	.042	.032	.046	.035	.024	.035	.037	.019	-.006
	\multicolumn{5}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
	Left side
.000	-.026	.616	.003	-.805	-1.317	.098	.544	-.111	-.566	-.845
.025	.339	-.142	-1.038	-1.320	-1.131	.175	-.218	-.868	-.746	-.741
.075	.148	-.169	-.555	-1.295	-1.140	-.015	-.200	-.462	-.724	-.739
.150	.030	-.182	-.410	-1.225	-1.108	-.116	-.178	-.308	-.679	-.725
.250	-.037	-.178	-.315	-.807	-1.077	-.091	-.108	-.219	-.590	-.689
.350	-.100	-.196	-.285	-.329	-1.009					
.450	-.136	-.198	-.256	-.245	-.866					
.550	-.143	-.180	-.215	-.220	-.693					
.650	-.114	-.137	-.151	-.159	-.505	-.123	-.081	-.133	-.403	-.575
.750	-.073	-.072	-.088	-.094	-.335	-.091	-.046	-.104	-.351	-.553
.850	-.028	-.021	-.029	-.033	-.208	-.055	-.007	-.074	-.293	-.521
.900	-.008	.002	.005	-.008	-.156	-.033	.011	-.049	-.254	-.494
	Right side
.025	-1.037	-.115	.359	.556	.665	-.828	-.178	.196	.321	.397
.075	-.569	-.164	.137	.346	.518	-.503	-.216	-.033	.087	.207
.150	-.393	-.162	.044	.204	.366	-.328	-.185	-.120	-.060	.032
.250	-.337	-.182	-.045	.087	.230					
.350	-.301	-.189	-.090	.001	.130					
.450	-.258	-.189	-.129	-.051	.057					
.550	-.215	-.167	-.124	-.074	.010					
.650	-.154	-.122	-.102	-.065	-.006	-.114	-.084	-.097	-.110	-.113
.750	-.093	-.066	-.061	-.042	-.018	-.116	-.050	-.090	-.119	-.147
.850	-.017	.011	.012	.012	.010	-.060	.011	-.031	-.071	-.131
.900	.019	.033	.032	.028	.003	-.037	.022	-.015	-.069	-.163

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Continued

(k) $\alpha = 15.8°$; M = 0.80.

z/bv = 0.11 and z/bv = 0.38

$\frac{x}{c_v}$	C_p for — (z/b_v = 0.11)					C_p for — (z/b_v = 0.38)				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.332	.837	.665	.540	-.437	.436	.729	.497	.156	-1.503
.025	-1.033	-.195	.270	-.101	-1.153	.299	-.148	-1.180	-1.651	-1.970
.075	-.308	-.203	-.109	-.265	-.724	.109	-.201	-.551	-.954	-2.095
.150	-.131	-.242	-.353	-.457	-.548	.021	-.207	-.430	-.688	-1.489
.250	-.089	-.191	-.282	-.353	-.367	-.050	-.215	-.365	-.460	-.501
.350	-.092	-.171	-.258	-.306	-.301	-.118	-.237	-.339	-.427	-.480
.450	-.118	-.192	-.256	-.290	-.268	-.158	-.251	-.309	-.370	-.409
.550	-.127	-.181	-.218	-.247	-.224	-.145	-.210	-.239	-.284	-.315
.650	-.098	-.117	-.137	-.180	-.151	-.119	-.153	-.171	-.211	-.233
.750	-.071	-.064	-.078	-.104	-.092	-.066	-.083	-.085	-.122	-.153
.850	-.027	-.002	-.014	-.018	-.020	.002	.000	-.005	-.042	-.067
.900	.009	.042	.030	.022	.034	.036	.042	.040	.004	-.015
Right side										
.025	.359	-.088	-.979	-.100	.619	-1.167	-.129	.346	.493	.762
.075	-.116	-.212	-.303	-.088	.347	-.529	-.194	.116	.252	.530
.150	-.324	-.224	-.118	-.054	.201	-.422	-.201	.019	.121	.358
.250	-.276	-.186	-.081	-.045	.131	-.351	-.204	-.041	.035	.222
.350	-.264	-.181	-.093	-.057	.067	-.321	-.218	-.093	-.036	.124
.450	-.246	-.195	-.123	-.082	.025	-.290	-.234	-.143	-.101	.048
.550	-.196	-.168	-.123	-.076	.016	-.232	-.204	-.144	-.119	.006
.650	-.127	-.109	-.097	-.048	.027	-.155	-.139	-.111	-.100	.006
.750	-.068	-.059	-.064	-.030	.032	-.069	-.064	-.053	-.063	.013
.850	.011	.016	-.003	.025	.067	.017	.021	.019	-.006	.041
.900	.032	.047	.019	.037	.077	.042	.041	.034	-.005	.034

z/bv = 0.66 and z/bv = 0.93

$\frac{x}{c_v}$	C_p for — (z/b_v = 0.66)					C_p for — (z/b_v = 0.93)				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.234	.655	.269	-.213	-1.322	.229	.563	.083	-.400	-.834
.025	.332	-.160	-1.365	-1.385	-1.130	.169	-.280	-1.073	-.740	-.726
.075	.151	-.197	-.760	-1.302	-1.135	-.035	-.265	-.758	-.746	-.724
.150	.026	-.218	-.504	-1.104	-1.106	-.152	-.218	-.451	-.725	-.710
.250	-.054	-.219	-.377	-.898	-1.078	-.099	-.124	-.294	-.657	-.672
.350	-.122	-.239	-.327	-.651	-1.001					
.450	-.169	-.246	-.289	-.412	-.853					
.550	-.179	-.219	-.239	-.281	-.677					
.650	-.137	-.151	-.168	-.180	-.480	-.119	-.085	-.149	-.451	-.550
.750	-.078	-.076	-.088	-.097	-.301	-.081	-.042	-.112	-.391	-.534
.850	-.024	-.014	-.017	-.027	-.175	-.048	.001	-.073	-.330	-.494
.900	.000	.021	.016	.001	-.123	-.027	.024	-.049	-.293	-.475
Right side										
.025	-1.468	-.135	.367	.554	.729	-.900	-.224	.198	.349	.450
.075	-.652	-.203	.149	.337	.574	-.724	-.278	-.066	.099	.255
.150	-.452	-.195	.043	.200	.419	-.452	-.227	-.164	-.079	.072
.250	-.379	-.218	-.053	.083	.281					
.350	-.329	-.246	-.121	-.015	.170					
.450	-.279	-.234	-.161	-.079	.095					
.550	-.231	-.204	-.162	-.110	.051					
.650	-.157	-.139	-.125	-.100	.034	-.119	-.086	-.108	-.130	-.076
.750	-.075	-.076	-.072	-.066	.020	-.112	-.049	-.090	-.137	-.106
.850	.014	.018	.010	-.003	.046	-.047	.025	-.023	-.081	-.097
.900	.044	.048	.031	.014	.044	-.029	.042	-.007	-.087	-.123

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

AND VERTICAL TAIL - Concluded

(*l*) $\alpha = 15.9°$; M = 0.85.

$\frac{x}{c_v}$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for— $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.384	.852	.699	.482		.487	.736	.552	.251	
.025	-1.035	-.220	.301	-.155		.273	-.183	-1.230	-1.497	
.075	-.535	-.238	-.073	-.289		.078	-.244	-.614	-.922	
.150	-.160	-.292	-.379	-.474		-.015	-.245	-.493	-.806	
.250	-.118	-.234	-.323	-.450		-.087	-.248	-.399	-.592	
.350	-.123	-.202	-.291	-.359		-.161	-.281	-.388	-.416	
.450	-.142	-.231	-.294	-.310		-.206	-.303	-.341	-.423	
.550	-.143	-.219	-.247	-.264		-.189	-.248	-.257	-.317	
.650	-.108	-.141	-.155	-.191		-.153	-.183	-.182	-.226	
.750	-.071	-.090	-.094	-.120		-.092	-.103	-.098	-.131	
.850	-.021	-.022	-.021	-.039		-.016	-.013	-.010	-.043	
.900	.017	.029	.023	.009		.027	.033	.037	.005	
Right side										
.025	.393	-.101	-.982	-.073		-1.108	-.145	.319	.462	
.075	-.064	-.254	-.465	-.094		-.635	-.223	.091	.227	
.150	-.371	-.268	-.131	-.076		-.535	-.230	-.010	.098	
.250	-.350	-.227	-.105	-.070		-.406	-.240	-.075	.013	
.350	-.324	-.217	-.115	-.087		-.379	-.260	-.128	-.059	
.450	-.296	-.228	-.135	-.107		-.327	-.274	-.179	-.130	
.550	-.237	-.202	-.129	-.099		-.260	-.238	-.178	-.147	
.650	-.154	-.131	-.098	-.070		-.173	-.169	-.136	-.124	
.750	-.094	-.083	-.064	-.046		-.083	-.087	-.073	-.075	
.850	-.007	.001	-.002	.006		.005	.006	.006	-.017	
.900	.020	.035	.027	.023		.037	.033	.024	-.009	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.306	.654	.348	-.086		.254	.552	.131	-.365	
.025	.299	-.200	-1.402	-1.388		.134	-.343	-1.332	-.831	
.075	.116	-.233	-1.168	-1.283		-.077	-.344	-.915	-.838	
.150	-.009	-.258	-.520	-1.127		-.249	-.260	-.531	-.800	
.250	-.092	-.268	-.422	-.881		-.136	-.146	-.276	-.735	
.350	-.174	-.291	-.340	-.712						
.450	-.223	-.292	-.311	-.504						
.550	-.226	-.255	-.263	-.337						
.650	-.173	-.176	-.182	-.209		-.139	-.103	-.162	-.440	
.750	-.102	-.094	-.098	-.110		-.095	-.053	-.115	-.376	
.850	-.038	-.028	-.023	-.034		-.053	-.006	-.078	-.304	
.900	-.008	.011	.014	.000		-.028	.021	-.050	-.267	
Right side										
.025	-1.374	-.149	.332	.537		-1.403	-.254	.166	.342	
.075	-1.137	-.226	.114	.313		-.981	-.344	-.104	.084	
.150	-.472	-.220	.011	.179		-.564	-.267	-.247	-.119	
.250	-.443	-.260	-.087	.056						
.350	-.347	-.293	-.165	-.051						
.450	-.308	-.278	-.206	-.116						
.550	-.251	-.235	-.200	-.144						
.650	-.173	-.163	-.152	-.126		-.144	-.104	-.124	-.151	
.750	-.091	-.090	-.097	-.087		-.121	-.064	-.102	-.150	
.850	.005	.009	-.004	-.017		-.050	.016	-.026	-.089	
.900	.037	.039	.025	.003		-.032	.032	-.014	-.092	

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL

(a) $\alpha = 0°$; M = 0.60.

$\dfrac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.559	.657	.242	-.658	-1.359	.171	.736	.197	-.642	-1.274
.025	.317	-.106	-.632	-1.400	-1.573	.374	-.090	-.771	-1.297	-1.021
.075	.169	-.110	-.422	-.797	-1.518	.178	-.130	-.483	-1.256	-1.009
.150	.068	-.130	-.334	-.555	-1.267	.078	-.140	-.348	-1.158	-.989
.250	.043	-.094	-.224	-.343	-.783	.002	-.142	-.283	-.731	-.950
.350	-.005	-.106	-.186	-.260	-.498	-.053	-.160	-.260	-.352	-.872
.450	-.041	-.117	-.168	-.219	-.370	-.103	-.176	-.233	-.256	-.749
.550	-.055	-.103	-.136	-.176	-.290	-.085	-.144	-.179	-.196	-.619
.650	-.028	-.069	-.084	-.130	-.256	-.069	-.108	-.123	-.146	-.512
.750	-.016	-.026	-.039	-.087	-.206	-.028	-.049	-.060	-.082	-.397
.850	.020	.019	.017	-.028	-.142	.025	.019	.017	.000	-.283
.900	.050	.051	.044	.004	-.094	.057	.057	.055	.039	-.215
Right side										
.025	-.553	-.035	.366	.669	.877	-.703	-.069	.413	.635	.753
.075	-.429	-.119	.163	.395	.591	-.464	-.124	.188	.390	.550
.150	-.322	-.121	.082	.253	.422	-.352	-.135	.080	.240	.390
.250	-.224	-.097	.044	.167	.297	-.272	-.137	.010	.128	.256
.350	-.199	-.108	-.006	.089	.196	-.247	-.149	-.037	.062	.162
.450	-.171	-.117	-.039	.036	.119	-.222	-.160	-.073	.000	.089
.550	-.130	-.094	-.039	.016	.082	-.180	-.137	-.080	-.023	.039
.650	-.073	-.058	-.017	.030	.066	-.117	-.092	-.053	-.012	.018
.750	-.030	-.017	-.001	.034	.055	-.048	-.031	-.010	.014	.014
.850	.027	.035	.042	.064	.066	.032	.041	.051	.059	.020
.900	.055	.055	.057	.057	.059	.057	.060	.057	.059	-.007
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.151	.718	.172	-.295	-.731	.347	.675	.154	-.142	-.525
.025	.379	-.126	-1.003	-.877	-.724	.230	-.176	-.837	-.662	-.528
.075	.178	-.151	-.528	-.845	-.715	.048	-.167	-.420	-.651	-.548
.150	.062	-.162	-.393	-.806	-.715	-.046	-.146	-.292	-.639	-.580
.250	-.016	-.164	-.301	-.735	-.687	-.028	-.094	-.213	-.566	-.573
.350	-.073	-.178	-.262	-.628	-.667					
.450	-.105	-.180	-.235	-.507	-.637					
.550	-.105	-.160	-.193	-.375	-.594					
.650	-.080	-.119	-.129	-.270	-.543	-.085	-.074	-.118	-.388	-.436
.750	-.039	-.047	-.064	-.171	-.486	-.064	-.042	-.082	-.356	-.416
.850	.009	.005	.001	-.089	-.425	-.032	-.011	-.073	-.315	-.391
.900	.036	.028	.028	-.059	-.386	-.009	.014	-.060	-.283	-.379
Right side										
.025	-.868	-.087	.416	.598	.689	-.630	-.146	.251	.363	.457
.075	-.507	-.149	.179	.356	.507	-.407	-.183	.019	.116	.224
.150	-.356	-.146	.073	.215	.347	-.283	-.158	-.055	-.009	.073
.250	-.299	-.162	-.010	.105	.210					
.350	-.263	-.171	-.069	.020	.107					
.450	-.222	-.176	-.093	-.028	.039					
.550	-.174	-.153	-.100	-.059	-.009					
.650	-.110	-.106	-.073	-.043	-.032	-.053	-.063	-.073	-.087	-.110
.750	-.053	-.031	-.046	-.025	-.055	-.085	-.051	-.071	-.096	-.153
.850	.023	.030	.044	.023	-.059	-.032	.010	-.010	-.062	-.142
.900	.057	.060	.060	.030	-.103	-.028	.028	-.006	-.073	-.174

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(b) $\alpha = 0°$; M = 0.80.

$\frac{x}{c_v}$	\multicolumn{5}{c}{C_p for - ($z/b_v = 0.11$)}	\multicolumn{5}{c}{C_p for - ($z/b_v = 0.38$)}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.701	.682	.465	.090	-.399	.412	.771	.483	-.073	-.642
.025	.328	-.067	-.633	-1.441	-1.116	.371	-.073	-1.211	-1.769	-1.572
.075	.180	-.096	-.428	-1.067	-1.054	.180	-.131	-.484	-.830	-.863
.150	.077	-.132	-.363	-.703	-.992	.079	-.141	-.362	-.801	-.830
.250	.044	-.102	-.240	-.320	-.857	.007	-.156	-.300	-.437	-.798
.350	-.002	-.112	-.193	-.259	-.687	-.062	-.173	-.277	-.352	-.749
.450	-.048	-.132	-.179	-.228	-.511	-.118	-.203	-.247	-.298	-.684
.550	-.065	-.118	-.143	-.181	-.373	-.100	-.161	-.182	-.222	-.604
.650	-.041	-.072	-.087	-.137	-.313	-.077	-.115	-.124	-.145	-.531
.750	-.025	-.038	-.036	-.082	-.181	-.034	-.047	-.047	-.065	-.443
.850	.018	.016	.024	-.009	-.176	.028	.030	.033	.020	-.364
.900	.044	.052	.060	.026	-.126	.063	.077	.077	.064	-.302
Right side
.025	-.513	-.005	.374	.659	.869	-1.216	-.059	.407	.627	.764
.075	-.459	-.105	.175	.399	.605	-.481	-.126	.189	.393	.566
.150	-.343	-.118	.091	.263	.433	-.363	-.135	.091	.253	.404
.250	-.236	-.103	.047	.183	.317	-.287	-.144	.014	.145	.275
.350	-.203	-.115	-.004	.097	.207	-.256	-.155	-.044	.062	.174
.450	-.170	-.129	-.045	.037	.119	-.229	-.174	-.092	.003	.078
.550	-.126	-.100	-.045	.015	.077	-.177	-.152	-.093	-.029	.024
.650	-.070	-.055	-.021	.027	.057	-.106	-.099	-.063	-.023	-.001
.750	-.024	-.023	-.010	.024	.034	-.028	-.026	-.009	.006	-.013
.850	.036	.037	.039	.055	.031	.054	.052	.057	.058	-.017
.900	.063	.065	.054	.062	.016	.074	.077	.075	.067	-.057

$\frac{x}{c_v}$	\multicolumn{5}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.385	.745	.434	-.058	-.430	.456	.703	.350	-.033	-.310
.025	.371	-.111	-1.175	-1.131	-.670	.207	-.221	-.986	-.729	-.443
.075	.183	-.149	-.661	-1.081	-.655	.016	-.200	-.578	-.717	-.458
.150	.057	-.168	-.425	-.970	-.651	-.073	-.164	-.320	-.723	-.483
.250	-.027	-.182	-.315	-.794	-.626	-.034	-.091	-.225	-.635	-.495
.350	-.089	-.194	-.270	-.627	-.605					
.450	-.131	-.203	-.241	-.470	-.573					
.550	-.128	-.171	-.191	-.336	-.539					
.650	-.092	-.120	-.118	-.221	-.511	-.079	-.067	-.109	-.399	-.407
.750	-.042	-.026	-.044	-.126	-.478	-.047	-.020	-.071	-.349	-.390
.850	.021	.030	.027	-.049	-.434	-.013	.018	-.054	-.295	-.366
.900	.039	.051	.059	-.015	-.419	.005	.043	-.039	-.263	-.357
Right side
.025	-1.250	-.066	.413	.607	.717	-1.148	-.156	.243	.396	.499
.075	-.576	-.144	.190	.378	.525	-.527	-.206	.002	.125	.254
.150	-.378	-.140	.082	.234	.374	-.282	-.167	-.071	-.026	.080
.250	-.313	-.168	-.017	.108	.233					
.350	-.274	-.193	-.080	.020	.119					
.450	-.235	-.190	-.116	-.046	.031					
.550	-.187	-.159	-.115	-.068	-.026					
.650	-.115	-.109	-.077	-.049	-.051	-.089	-.062	-.066	-.082	-.110
.750	-.041	-.025	-.029	-.020	-.075	-.074	-.026	-.054	-.090	-.160
.850	.048	.057	.054	.035	-.081	-.015	.045	.011	-.040	-.146
.900	.077	.086	.080	.044	-.120	-.013	.057	.021	-.055	-.175

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(c) $\alpha = 0°$; M = 0.85.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.742	.688	.528	.217	-.237	.483	.787	.555	.088	-.500
.025	.334	-.061	-.596	-1.290	-1.076	.369	-.072	-1.174	-1.556	-.848
.075	.183	-.093	-.457	-.980	-.988	.183	-.137	-.393	-.823	-.847
.150	.075	-.140	-.386	-.844	-.926	.078	-.152	-.360	-1.273	-.815
.250	.044	-.109	-.252	-.420	-.833	-.002	-.166	-.312	-.439	-.781
.350	-.007	-.122	-.208	-.260	-.707	-.077	-.193	-.293	-.255	-.736
.450	-.062	-.145	-.192	-.204	-.565	-.134	-.225	-.263	-.281	-.679
.550	-.077	-.129	-.154	-.179	-.436	-.124	-.180	-.198	-.220	-.613
.650	-.050	-.084	-.092	-.141	-.362	-.099	-.132	-.126	-.149	-.547
.750	-.039	-.049	-.042	-.081	-.296	-.052	-.057	-.057	-.069	-.460
.850	.010	.014	.026	-.010	-.207	.027	.030	.038	.025	-.387
.900	.039	.054	.063	.033	-.154	.068	.075	.080	.068	-.330
Right side										
.025	-.485	-.001	.374	.659	.873	-1.154	-.058	.401	.624	.769
.075	-.498	-.105	.173	.404	.608	-.427	-.129	.186	.388	.567
.150	-.370	-.123	.088	.263	.439	-.369	-.145	.083	.246	.405
.250	-.256	-.109	.043	.177	.318	-.302	-.157	.003	.139	.274
.350	-.217	-.126	-.015	.094	.210	-.272	-.174	-.059	.057	.176
.450	-.188	-.140	-.062	.025	.116	-.241	-.193	-.108	-.016	.074
.550	-.130	-.109	-.064	.005	.068	-.185	-.163	-.111	-.048	.017
.650	-.074	-.064	-.035	.015	.047	-.111	-.106	-.075	-.038	-.016
.750	-.032	-.037	-.021	.013	.014	-.032	-.032	-.025	-.002	-.027
.850	.044	.038	.033	.047	.010	.057	.053	.053	.056	-.037
.900	.071	.060	.052	.053	-.009	.084	.077	.069	.061	-.074
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.443	.759	.501	.064	-.333	.483	.711	.403	.004	-.279
.025	.371	-.113	-1.253	-1.599	-.692	.196	-.254	-1.108	-.864	-.467
.075	.183	-.160	-.988	-1.436	-.689	-.010	-.240	-.654	-.814	-.474
.150	.055	-.187	-.403	-1.076	-.695	-.101	-.188	-.340	-.813	-.500
.250	-.040	-.203	-.316	-.757	-.665	-.050	-.112	-.225	-.689	-.518
.350	-.108	-.218	-.279	-.606	-.631					
.450	-.151	-.227	-.255	-.436	-.589					
.550	-.148	-.196	-.205	-.283	-.557					
.650	-.110	-.137	-.129	-.162	-.525	-.089	-.075	-.115	-.422	-.430
.750	-.053	-.031	-.047	-.071	-.490	-.052	-.030	-.078	-.369	-.409
.850	.024	.027	.026	-.002	-.444	-.017	.016	-.059	-.310	-.392
.900	.042	.058	.056	.025	-.424	.010	.043	-.044	-.268	-.376
Right side										
.025	-1.272	-.074	.403	.604	.719	-1.237	-.177	.223	.394	.507
.075	-.968	-.156	.173	.369	.526	-.770	-.248	-.039	.113	.257
.150	-.357	-.154	.072	.227	.371	-.244	-.186	-.108	-.052	.069
.250	-.313	-.191	-.032	.103	.224					
.350	-.282	-.218	-.105	.004	.103					
.450	-.248	-.211	-.145	-.068	.012					
.550	-.191	-.180	-.141	-.088	-.046					
.650	-.114	-.116	-.095	-.065	-.074	-.097	-.075	-.081	-.092	-.136
.750	-.040	-.038	-.044	-.032	-.094	-.091	-.032	-.059	-.091	-.182
.850	.052	.057	.052	.035	-.094	-.026	.048	.016	-.042	-.161
.900	.085	.082	.078	.051	-.128	-.023	.060	.028	-.055	-.191

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(d) $\alpha = 0°$; M = 0.90.

$z/b_v = 0.11$ — Left side and **$z/b_v = 0.38$ — Left side**

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.773	.700	.586	.342	-.084	.546	.796	.625	.233	-.347
.025	.340	-.047	-.520	-1.149	-1.034	.372	-.067	-1.037	-1.151	-1.130
.075	.198	-.088	-.449	-.873	-.908	.187	-.141	-.406	-.487	-.584
.150	.084	-.144	-.425	-.806	-.844	.083	-.160	-.429	-1.182	-.753
.250	.048	-.120	-.282	-.650	-.771	.002	-.183	-.322	-1.065	-.738
.350	-.006	-.137	-.223	-.376	-.673	-.079	-.219	-.341	-.455	-.702
.450	-.071	-.165	-.215	-.259	-.578	-.156	-.271	-.275	-.221	-.646
.550	-.091	-.156	-.159	-.165	-.475	-.144	-.216	-.194	-.147	-.593
.650	-.063	-.103	-.095	-.119	-.404	-.117	-.149	-.124	-.100	-.536
.750	-.049	-.064	-.040	-.063	-.339	-.055	-.063	-.044	-.039	-.464
.850	.004	.004	.028	-.001	-.239	.026	.032	.046	.042	-.400
.900	.038	.050	.068	.035	-.183	.075	.080	.092	.084	-.348

$z/b_v = 0.11$ — Right side and **$z/b_v = 0.38$ — Right side**

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-.425	.005	.377	.662	.882	-1.035	-.064	.392	.623	.782
.075	-.488	-.108	.179	.409	.622	-.835	-.141	.181	.390	.577
.150	-.418	-.131	.087	.271	.459	-.408	-.155	.076	.249	.416
.250	-.278	-.123	.036	.187	.337	-.310	-.173	-.004	.143	.287
.350	-.232	-.145	-.022	.097	.219	-.298	-.201	-.073	.047	.185
.450	-.195	-.161	-.071	.023	.124	-.236	-.232	-.131	-.033	.076
.550	-.128	-.131	-.080	-.001	.069	-.179	-.195	-.141	-.064	.015
.650	-.067	-.077	-.047	.008	.049	-.103	-.125	-.100	-.052	-.020
.750	-.020	-.045	-.039	-.001	.009	-.016	-.039	-.037	-.012	-.037
.850	.047	.028	.026	.045	-.005	.068	.050	.050	.051	-.049
.900	.079	.057	.042	.051	-.032	.097	.080	.068	.061	-.086

$z/b_v = 0.66$ — Left side and **$z/b_v = 0.93$ — Left side**

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.503	.769	.574	.203	-.219	.523	.712	.474	.103	-.211
.025	.371	-.104	-1.133	-1.454	-.678	.182	-.276	-1.073	-1.185	-.447
.075	.181	-.165	-.976	-1.321	-.686	-.029	-.325	-1.037	-1.053	-.460
.150	.051	-.208	-.795	-1.236	-.698	-.175	-.215	-.572	-.996	-.487
.250	-.044	-.239	-.369	-1.068	-.659	-.071	-.117	-.155	-.880	-.501
.350	-.129	-.265	-.226	-.708	-.622					
.450	-.182	-.264	-.234	-.455	-.584					
.550	-.178	-.229	-.207	-.232	-.549					
.650	-.123	-.165	-.127	-.098	-.525	-.087	-.076	-.117	-.467	-.435
.750	-.060	-.023	-.039	-.028	-.493	-.039	-.019	-.077	-.403	-.420
.850	.031	.032	.038	.033	-.451	.002	.028	-.061	-.338	-.396
.900	.054	.062	.066	.053	-.428	.020	.053	-.044	-.305	-.383

$z/b_v = 0.66$ — Right side and **$z/b_v = 0.93$ — Right side**

$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-1.151	-.077	.391	.596	.731	-1.162	-.203	.203	.390	.517
.075	-.969	-.168	.166	.363	.536	-1.049	-.346	-.069	.104	.270
.150	-.663	-.169	.062	.224	.382	-.448	-.216	-.194	-.094	.071
.250	-.350	-.221	-.052	.094	.238					
.350	-.234	-.268	-.137	-.020	.117					
.450	-.223	-.253	-.186	-.099	.017					
.550	-.183	-.208	-.172	-.126	-.054					
.650	-.108	-.128	-.117	-.090	-.080	-.087	-.073	-.083	-.108	-.134
.750	-.024	-.037	-.053	-.044	-.102	-.079	-.028	-.060	-.096	-.186
.850	.068	.062	.059	.039	-.090	-.020	.052	.023	-.031	-.156
.900	.101	.094	.079	.058	-.128	-.014	.069	.031	-.041	-.185

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(e) $\alpha = 0°$; M = 0.92.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.785	.690	.604	.383	-.063	.569	.799	.648	.283	-.327
.025	.331	-.044	-.491	-1.098	-1.057	.361	-.070	-.970	-1.315	-.763
.075	.190	-.087	-.442	-.830	-.928	.176	-.144	-.775	-1.075	-.784
.150	.076	-.152	-.429	-.775	-.860	.073	-.169	-.444	-1.124	-.767
.250	.036	-.127	-.342	-.683	-.787	-.015	-.188	-.379	-1.041	-.754
.350	-.024	-.147	-.247	-.436	-.688	-.102	-.236	-.361	-.654	-.716
.450	-.092	-.187	-.238	-.343	-.598	-.192	-.300	-.361	-.427	-.668
.550	-.122	-.190	-.182	-.218	-.500	-.190	-.255	-.175	-.140	-.611
.650	-.089	-.122	-.095	-.128	-.425	-.152	-.173	-.112	-.070	-.557
.750	-.072	-.075	-.040	-.061	-.360	-.072	-.063	-.046	-.018	-.484
.850	-.016	-.002	.025	.002	-.273	.015	.030	.046	.055	-.425
.900	.023	.043	.064	.044	-.212	.067	.081	.093	.093	-.368
	Right side									
.025	-.399	.008	.372	.657	.886	-.946	-.069	.383	.620	.784
.075	-.480	-.113	.175	.409	.623	-.806	-.152	.172	.387	.578
.150	-.447	-.138	.085	.269	.457	-.490	-.166	.068	.248	.418
.250	-.339	-.132	.037	.183	.336	-.344	-.187	-.017	.132	.282
.350	-.266	-.158	-.036	.085	.213	-.349	-.221	-.085	.040	.173
.450	-.226	-.182	-.087	.009	.113	-.306	-.274	-.153	-.042	.066
.550	-.147	-.157	-.100	-.020	.062	-.168	-.229	-.175	-.087	.000
.650	-.070	-.089	-.066	-.007	.039	-.101	-.136	-.127	-.078	-.035
.750	-.031	-.054	-.058	.016	-.004	-.019	-.040	-.050	-.033	-.055
.850	.043	.021	.008	.028	-.026	.062	.048	.041	.042	-.066
.900	.071	.052	.030	.043	-.055	.091	.078	.061	.057	-.110
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.528	.769	.597	.249	-.208	.536	.712	.498	.150	-.218
.025	.345	-.109	-1.063	-1.376	-.692	.160	-.282	-1.013	-1.327	-.473
.075	.164	-.169	-.926	-1.246	-.700	-.046	-.309	-.992	-1.108	-.478
.150	.036	-.213	-.827	-1.187	-.710	-.229	-.282	-.832	-1.024	-.506
.250	-.065	-.255	-.399	-1.128	-.692	-.142	-.136	-.414	-.947	-.522
.350	-.168	-.310	-.412	-1.020	-.647					
.450	-.241	-.317	-.178	-.529	-.601					
.550	-.229	-.261	-.171	-.287	-.573					
.650	-.156	-.167	-.115	-.081	-.554	-.100	-.075	-.115	-.560	-.461
.750	-.050	-.030	-.039	.010	-.523	-.046	-.023	-.084	-.461	-.442
.850	.022	.028	.038	.060	-.478	-.010	.029	-.067	-.373	-.420
.900	.045	.063	.071	.080	-.461	.019	.052	-.048	-.?26	-.408
	Right side									
.025	-1.065	-.083	.379	.591	.729	-1.089	-.209	.197	.391	.520
.075	-.916	-.180	.158	.362	.530	-1.008	-.375	-.079	.101	.264
.150	-.777	-.183	.051	.223	.379	-.819	-.275	-.235	-.105	.059
.250	-.380	-.242	-.062	.086	.232					
.350	-.401	-.320	-.164	-.029	.104					
.450	-.168	-.294	-.236	-.112	.000					
.550	-.168	-.229	-.216	-.151	-.074					
.650	-.110	-.126	-.144	-.109	-.103	-.106	-.076	-.082	-.109	-.160
.750	-.032	-.035	-.046	-.057	-.133	-.093	-.031	-.057	-.099	-.210
.850	.065	.063	.056	.043	-.111	-.029	.052	.021	-.028	-.178
.900	.096	.098	.081	.066	-.148	-.020	.069	.032	-.033	-.207

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(f) $\alpha = 9.4^\circ$; $M = 0.60$.

$\frac{x}{c_v}$	C_p for – $\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	C_p for – $\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
		$z/b_v = 0.11$					$z/b_v = 0.38$			
Left side										
.000	.385	.806		-.650	-1.449	.002	.661		-1.036	-1.512
.025		-.130		-.972	-2.578	.349	-.109		-1.368	-1.289
.075	.037	-.127		-.622	-1.446	.169	-.143		-1.318	-1.273
.150	.021	-.155		-.472	-.695	.068	-.150		-1.159	-1.285
.250	.009	-.114		-.325	-.501	-.002	-.162		-.384	-1.253
.350	-.025	-.118		-.270	-.433	-.059	-.184		-.288	-.959
.450	-.057	-.139		-.241	-.346	-.100	-.205		-.295	-.631
.550	-.073	-.150		-.218	-.280	-.096	-.178		-.245	-.442
.650	-.068	-.123		-.186	-.239	-.084	-.141		-.193	-.346
.750	-.055	-.105		-.138	-.196	-.055	-.091		-.120	-.269
.850	-.016	-.043		-.061	-.114	-.002	-.018		-.036	-.173
.900	.011	-.014		-.027	-.059	.025	.016		.007	-.118
Right side										
.025	-.155	-.057		.589	.870	-.761	-.068		.589	.677
.075	-.319	-.132		.373	.590	-.483	-.125		.378	.535
.150	-.305	-.134		.259	.426	-.364	-.137		.239	.390
.250	-.225	-.109		.187	.317	-.287	-.146		.137	.269
.350	-.209	-.114		.109	.214	-.255	-.164		.064	.180
.450	-.203	-.137		.041	.135	-.241	-.178		.003	.098
.550	-.173	-.132		.016	.096	-.200	-.164		-.022	.048
.650	-.127	-.107		.005	.078	-.146	-.125		-.022	.025
.750	-.096	-.084		.000	.062	-.077	-.066		.000	.023
.850	-.023	-.018		.037	.075	.002	.005		.037	.039
.900	.007	.005		.050	.071	.030	.027		.041	.018
		$z/b_v = 0.66$					$z/b_v = 0.93$			
Left side										
.000	-.039	.613		-.732	-1.141	.137	.549		-.434	-.822
.025	.335	-.155		-1.072	-.856	.176	-.241		-.632	-.601
.075	.155	-.171		-1.045	-.840	-.009	-.205		-.629	-.606
.150	.043	-.182		-1.000	-.818	-.100	-.182		-.602	-.615
.250	-.030	-.184		-.904	-.790	-.077	-.116		-.536	-.601
.350	-.084	-.196		-.702	-.786					
.450	-.118	-.203		-.488	-.761					
.550	-.134	-.187		-.325	-.706	-.102	-.082		-.379	-.487
.650	-.100	-.141		-.213	-.629	-.075	-.041		-.338	-.465
.750	-.055	-.077		-.129	-.544	-.052	-.011		-.309	-.437
.850	-.014	-.023		-.068	-.465	-.027	.002		-.268	-.412
.900	.000	.000		-.038	-.426					
Right side										
.025	-.998	-.086		.548	.611	-.717	-.159		.318	.371
.075	-.537	-.146		.353	.476	-.435	-.198		.087	.178
.150	-.371	-.153		.214	.340	-.291	-.178		-.057	.005
.250	-.317	-.173		.103	.214					
.350	-.285	-.182		.021	.112					
.450	-.241	-.182		-.022	.041					
.550	-.203	-.164		-.057	-.007					
.650	-.146	-.125		-.047	-.030	-.107	-.075		-.116	-.157
.750	-.084	-.064		-.029	-.057	-.091	-.059		-.127	-.194
.850	-.004	.002		.018	-.055	-.032	.007		-.088	-.184
.900	.037	.032		.030	-.082	-.016	.025		-.097	-.209

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(g) $\alpha = 9.6^\circ$; $M = 0.80$.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	$z/b_v=0.11$					$z/b_v=0.38$				
Left side										
.000	.350	.856	.414	-.228	-.308	.284	.689	.329	-.357	-1.016
.025		-.113	-.153	-.745	-1.646	.341	-.109	-1.110	-1.477	-1.059
.075	.019	-.127	-.330	-.599	-1.326	.161	-.154	-.554	-1.419	-1.026
.150	.003	-.162	-.369	-.545	-.834	.069	-.157	-.387	-1.194	-.981
.250	-.004	-.115	-.256	-.371	-.557	-.001	-.168	-.320	-.259	-.941
.350	-.041	-.124	-.224	-.293	-.474	-.071	-.197	-.305	-.325	-.854
.450	-.073	-.151	-.221	-.262	-.365	-.121	-.231	-.282	-.326	-.715
.550	-.091	-.163	-.208	-.240	-.335	-.121	-.206	-.230	-.280	-.592
.650	-.079	-.139	-.163	-.219	-.318	-.111	-.163	-.171	-.225	-.498
.750	-.074	-.124	-.127	-.160	-.294	-.071	-.106	-.102	-.150	-.407
.850	-.027	-.061	-.054	-.082	-.191	-.003	-.020	-.011	-.057	-.303
.900	.005	-.008	-.006	-.033	-.131	.031	.030	.035	-.006	-.238
Right side										
.025	-.077	-.020	-.041	.524	.836	-1.214	-.056	.388	.583	.697
.075	-.339	-.130	.027	.361	.590	-.500	-.120	.178	.374	.540
.150	-.348	-.141	.023	.257	.437	-.386	-.135	.082	.247	.400
.250	-.261	-.109	.001	.195	.329	-.314	-.150	.015	.146	.281
.350	-.237	-.123	-.031	.112	.230	-.282	-.172	-.043	.069	.190
.450	-.220	-.144	-.060	.046	.145	-.266	-.201	-.090	-.003	.096
.550	-.185	-.138	-.069	.014	.096	-.219	-.184	-.108	-.040	.042
.650	-.144	-.113	-.060	.004	.066	-.155	-.139	-.084	-.039	.012
.750	-.111	-.100	-.047	-.008	.036	-.077	-.071	-.040	-.014	.001
.850	-.030	-.023	.004	.027	.040	.012	.012	.029	.029	.000
.900	.005	.004	.024	.040	.028	.041	.036	.042	.032	-.026
	$z/b_v=0.66$					$z/b_v=0.93$				
Left side										
.000	.205	.634	.212	-.320	-.808	.253	.562	.123	-.253	-.669
.025	.330	-.151	-1.220	-.928	-.746	.154	-.295	-.807	-.596	-.541
.075	.157	-.177	-.735	-.890	-.727	-.035	-.254	-.691	-.585	-.541
.150	.038	-.197	-.451	-.857	-.705	-.140	-.209	-.439	-.590	-.547
.250	-.039	-.203	-.343	-.842	-.681	-.088	-.127	-.272	-.542	-.550
.350	-.103	-.221	-.301	-.774	-.665					
.450	-.149	-.233	-.272	-.626	-.645					
.550	-.158	-.213	-.227	-.465	-.615					
.650	-.120	-.148	-.156	-.326	-.583	-.105	-.071	-.130	-.359	-.480
.750	-.065	-.073	-.076	-.208	-.545	-.068	-.027	-.096	-.331	-.457
.850	-.010	-.008	-.009	-.113	-.498	-.030	.015	-.060	-.302	-.433
.900	.011	.025	.024	-.074	-.483	-.009	.038	-.032	-.283	-.418
Right side										
.025	-1.150	-.073	.378	.552	.627	-.980	-.177	.193	.327	.407
.075	-.661	-.145	.166	.352	.490	-.585	-.242	-.044	.084	.207
.150	-.404	-.153	.067	.220	.357	-.313	-.203	-.145	-.071	.016
.250	-.333	-.189	-.028	.103	.228					
.350	-.298	-.212	-.090	.014	.118					
.450	-.258	-.216	-.128	-.051	.033					
.550	-.203	-.188	-.134	-.080	-.025					
.650	-.143	-.139	-.101	-.071	-.055	-.108	-.067	-.089	-.125	-.171
.750	-.068	-.064	-.052	-.050	-.081	-.082	-.038	-.072	-.140	-.220
.850	.023	.027	.026	.004	-.084	-.009	.036	-.003	-.103	-.214
.900	.055	.054	.047	.018	-.123	.006	.050	.007	-.117	-.233

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(h) $\alpha = 9.7°$; M = 0.85.

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
					Left side					
.000	.311	.877	.390	-.245	-.143	.349	.702	.417	-.180	-.861
.025		-.106	-.112	-.633	-1.521	.335	-.107	-1.227	-1.658	-1.011
.075	.013	-.124	-.323	-.587	-1.248	.157	-.155	-.503	-1.315	-.984
.150	.000	-.172	-.418	-.609	-.831	.063	-.163	-.401	-.856	-.936
.250	-.016	-.126	-.282	-.447	-.574	-.013	-.178	-.350	-.580	-.900
.350	-.050	-.130	-.249	-.345	-.480	-.088	-.215	-.336	-.457	-.833
.450	-.081	-.164	-.247	-.298	-.394	-.142	-.254	-.309	-.395	-.716
.550	-.103	-.184	-.230	-.266	-.356	-.148	-.225	-.257	-.321	-.613
.650	-.098	-.160	-.188	-.254	-.346	-.135	-.181	-.194	-.248	-.530
.750	-.096	-.140	-.150	-.177	-.348	-.096	-.117	-.116	-.159	-.440
.850	-.046	-.069	-.072	-.095	-.230	-.018	-.020	-.021	-.062	-.347
.900	-.008	-.013	-.020	-.049	-.160	.025	.028	.031	-.011	-.281
					Right side					
.025	-.049	-.016	-.066	.465	.826	-1.182	-.076	.375	.569	.706
.075	-.341	-.136	.013	.333	.593	-.475	-.133	.168	.361	.546
.150	-.386	-.150	.007	.247	.442	-.401	-.150	.070	.231	.404
.250	-.285	-.118	-.014	.179	.333	-.340	-.164	-.001	.135	.284
.350	-.261	-.136	-.048	.098	.232	-.313	-.191	-.057	.051	.193
.450	-.245	-.158	-.076	.031	.143	-.290	-.221	-.108	-.028	.099
.550	-.204	-.154	-.089	-.001	.093	-.242	-.204	-.133	-.066	.036
.650	-.165	-.126	-.081	-.014	.063	-.175	-.158	-.113	-.069	-.001
.750	-.129	-.113	-.071	-.033	.026	-.088	-.086	-.058	-.041	-.011
.850	-.045	-.033	-.015	.011	.024	.004	.011	.016	.011	-.017
.900	-.005	.001	.010	.018	.012	.038	.035	.029	.011	-.051
	$z/b_v = 0.66$					$z/b_v = 0.93$				
					Left side					
.000	.256	.644	.291	-.254	-.726	.272	.561	.152	-.332	-.646
.025	.322	-.153	-1.326	-1.438	-.764	.141	-.289	-1.218	-.824	-.544
.075	.152	-.184	-.815	-1.315	-.733	-.058	-.303	-.779	-.803	-.541
.150	.028	-.207	-.498	-1.137	-.701	-.185	-.231	-.444	-.788	-.547
.250	-.056	-.221	-.380	-.876	-.680	-.108	-.150	-.255	-.694	-.557
.350	-.125	-.246	-.315	-.716	-.664					
.450	-.177	-.258	-.292	-.500	-.644					
.550	-.189	-.239	-.249	-.335	-.616					
.650	-.145	-.167	-.170	-.215	-.588	-.119	-.080	-.137	-.421	-.493
.750	-.082	-.077	-.086	-.121	-.553	-.076	-.035	-.098	-.367	-.473
.850	-.019	-.012	-.013	-.052	-.510	-.033	.012	-.059	-.313	-.448
.900	.011	1.098	.029	-.020	-.496	-.008	.034	-.031	-.285	-.434
					Right side					
.025	-1.350	-.093	.362	.536	.637	-1.320	-.215	.180	.328	.417
.075	-.629	-.167	.155	.338	.492	-.702	-.291	-.074	.073	.219
.150	-.430	-.170	.058	.208	.359	-.338	-.228	-.193	-.122	.014
.250	-.371	-.212	-.042	.085	.230					
.350	-.323	-.238	-.118	-.011	.117					
.450	-.281	-.244	-.162	-.083	.029					
.550	-.232	-.215	-.162	-.117	-.040	-.111	-.084	-.099	-.156	-.198
.650	-.155	-.158	-.125	-.105	-.068	-.091	-.040	-.086	-.151	-.237
.750	-.072	-.070	-.068	-.073	-.097	-.022	.037	-.008	-.096	-.227
.850	.018	.025	.019	-.005	-.098	.000	.052	.007	-.107	-.253
.900	.054	.055	.047	.008	-.140					

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(i) $\alpha = 9.7°$; M = 0.90.

Cp for z/bv = 0.11 (Left columns) and z/bv = 0.38 (Right columns)

Left side

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.000	.281	.895	.394	-.229						
.025		-.105	-.039	-.464		.417	.704	-.489	.016	
.075	.014	-.126	-.258	-.458		.321	-.109	-1.078	-1.413	
.150	.001	-.193	-.421	-.547		.147	-.167	-.642	-1.275	
.250	-.013	-.142	-.318	-.499		.054	-.177	-.506	-.796	
.350	-.049	-.149	-.265	-.414		-.021	-.195	-.357	-.686	
.450	-.088	-.190	-.270	-.345		-.096	-.237	-.363	-.537	
.550	-.117	-.233	-.263	-.277		-.164	-.302	-.375	-.357	
.650	-.125	-.199	-.209	-.272		-.185	-.283	-.269	-.337	
.750	-.120	-.187	-.171	-.256		-.178	-.243	-.201	-.357	
.850	-.060	-.094	-.077	-.124		-.120	-.140	-.119	-.217	
.900	-.014	-.032	-.021	-.058		-.027	-.029	-.017	-.078	
						.022	.027	.039	-.020	

Right side

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.025	.025	-.016	-.062	.417		-.991	-.081	.371	.567	
.075	-.277	-.152	.018	.331		-.662	-.153	.170	.362	
.150	-.413	-.173	.022	.250		-.487	-.165	.073	.238	
.250	-.319	-.142	-.001	.192		-.364	-.186	-.001	.140	
.350	-.295	-.157	-.041	.101		-.363	-.222	-.063	.053	
.450	-.276	-.183	-.075	.028		-.353	-.265	-.123	-.035	
.550	-.234	-.194	-.087	-.012		-.258	-.266	-.159	-.083	
.650	-.182	-.161	-.086	-.029		-.186	-.203	-.139	-.099	
.750	-.154	-.152	-.079	-.057		-.093	-.104	-.071	-.070	
.850	-.056	-.058	-.022	-.012		.002	-.003	.007	-.010	
.900	-.015	-.016	.006	.000		.041	.032	.029	-.001	

Cp for z/bv = 0.66 (Left columns) and z/bv = 0.93 (Right columns)

Left side

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.000	.320	.640	.363	-.066		.320	.560	.218	-.179	
.025	.309	-.152	-1.209	-1.357		.114	-.344	-1.194	-.775	
.075	.141	-.195	-1.030	-1.276		-.066	-.315	-1.119	-.783	
.150	.019	-.223	-.550	-1.124		-.309	-.330	-.674	-.803	
.250	-.074	-.250	-.426	-.746		-.201	-.129	-.209	-.762	
.350	-.160	-.314	-.449	-.603						
.450	-.236	-.331	-.249	-.559						
.550	-.242	-.303	-.239	-.515						
.650	-.172	-.187	-.166	-.371		-.122	-.086	-.137	-.459	
.750	-.093	-.085	-.081	-.223		-.078	-.034	-.097	-.397	
.850	-.022	-.008	-.003	-.109		-.030	.021	-.057	-.329	
.900	.014	.025	.038	-.057		-.003	.039	-.030	-.297	

Right side

x/c_v	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°	β=-3.9°	β=0°	β=3.9°	β=7.9°	β=12.7°
.025	-1.154	-.106	.353	.528		-1.201	-.234	.174	.330	
.075	-.965	-.181	.153	.330		-1.078	-.374	-.081	.077	
.150	-.417	-.189	.053	.209		-.619	-.308	-.299	-.152	
.250	-.432	-.243	-.054	.088						
.350	-.447	-.303	-.142	-.024						
.450	-.246	-.303	-.206	-.114						
.550	-.226	-.263	-.203	-.163						
.650	-.158	-.171	-.142	-.144		-.114	-.082	-.103	-.160	
.750	-.074	-.073	-.075	-.102		-.090	-.041	-.081	-.168	
.850	.026	.028	.019	-.020		-.017	.044	-.002	-.113	
.900	.062	.059	.047	-.004		.003	.056	.011	-.115	

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(j) $\alpha = 9.7^\circ$; $M = 0.92$.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.297	.912	.408	-.254		.437	.678	.512	.033	
.025		-.123	-.003	-.430		.318	-.139	-1.018	-1.388	
.075	.008	-.154	-.215	-.440		.148	-.205	-.566	-1.244	
.150	.010	-.230	-.391	-.532		.054	-.212	-.507	-.796	
.250	-.007	-.183	-.360	-.506		-.022	-.225	-.433	-.711	
.350	-.045	-.187	-.302	-.441		-.099	-.268	-.392	-.591	
.450	-.088	-.239	-.306	-.374		-.175	-.322	-.408	-.391	
.550	-.126	-.268	-.307	-.295		-.209	-.340	-.390	-.328	
.650	-.147	-.248	-.264	-.273		-.243	-.364	-.247	-.371	
.750	-.154	-.329	-.215	-.279		-.167	-.288	-.123	-.256	
.850	-.086	-.157	-.092	-.148		-.039	-.046	-.021	-.099	
.900	-.031	-.067	-.035	-.074		.016	.022	.028	-.032	
Right side										
.025	.058	-.033	-.084	.418		-.976	-.106	.351	.575	
.075	-.235	-.179	.005	.340		-.631	-.183	.153	.366	
.150	-.394	-.213	.015	.258		-.507	-.195	.057	.244	
.250	-.345	-.184	-.006	.191		-.393	-.213	-.012	.142	
.350	-.316	-.195	-.048	.104		-.393	-.255	-.078	.050	
.450	-.304	-.217	-.089	.026		-.393	-.301	-.144	-.040	
.550	-.271	-.240	-.108	-.019		-.355	-.314	-.189	-.093	
.650	-.222	-.218	-.118	-.041		-.201	-.338	-.206	-.119	
.750	-.179	-.277	-.123	-.077		-.091	-.201	-.119	-.091	
.850	-.071	-.114	-.055	-.036		-.002	-.015	-.011	-.029	
.900	-.024	-.059	-.019	-.020		.040	.024	.011	-.018	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.340	.623	.396	-.048		.331	.549	.250	-.163	
.025	.305	-.175	-1.147	-1.367		.132	-.369	-1.173	-.771	
.075	.140	-.222	-.980	-1.279		-.058	-.309	-1.076	-.780	
.150	.021	-.248	-.649	-1.115		-.312	-.424	-.914	-.792	
.250	-.071	-.273	-.489	-.744		-.260	-.295	-.433	-.746	
.350	-.166	-.330	-.472	-.614						
.450	-.268	-.413	-.485	-.561						
.550	-.341	-.443	-.247	-.528						
.650	-.256	-.342	-.140	-.416		-.109	-.082	-.126	-.477	
.750	-.092	-.096	-.068	-.277		-.074	-.027	-.095	-.416	
.850	-.018	-.001	-.003	-.157		-.026	.021	-.054	-.351	
.900	.014	.038	.035	-.101		.003	.048	-.029	-.318	
Right side										
.025	-1.133	-.123	.331	.531		-1.180	-.251	.164	.332	
.075	-.951	-.201	.135	.335		-1.064	-.364	-.083	.081	
.150	-.545	-.212	.041	.209		-.874	-.411	-.310	-.155	
.250	-.451	-.266	-.061	.087						
.350	-.486	-.330	-.165	-.027						
.450	-.456	-.403	-.255	-.120						
.550	-.207	-.408	-.314	-.185						
.650	-.133	-.309	-.216	-.168		-.104	-.084	-.101	-.181	
.750	-.062	-.066	-.089	-.127		-.086	-.045	-.079	-.189	
.850	.029	.035	.015	-.043		-.013	.041	-.003	-.128	
.900	.063	.061	.043	-.027		.004	.059	.011	-.132	

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(k) $\alpha = 15.6°$; $M = 0.60$.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.621	.807	.829	.123	-1.083	.002	.659	.105	-1.228	-1.884
.025	-.450	-.151	-.119	-.430	-2.349	.383	-.104	-.845	-1.868	-1.755
.075	-.176	-.151	-.262	-.421	-.866	.180	-.146	-.520	-1.467	-1.834
.150	-.110	-.178	-.309	-.430	-.646	.075	-.160	-.386	-.505	-1.761
.250	-.073	-.137	-.237	-.308	-.451	-.002	-.173	-.321	-.448	-1.006
.350	-.096	-.137	-.216	-.263	-.383	-.064	-.196	-.309	-.408	-.560
.450	-.126	-.169	-.219	-.247	-.358	-.107	-.220	-.296	-.353	-.503
.550	-.149	-.182	-.214	-.228	-.335	-.114	-.200	-.246	-.308	-.444
.650	-.144	-.155	-.185	-.192	-.306	-.107	-.169	-.201	-.269	-.383
.750	-.139	-.142	-.155	-.156	-.256	-.080	-.117	-.137	-.222	-.287
.850	-.103	-.093	-.101	-.095	-.163	-.025	-.048	-.065	-.158	-.192
.900	-.078	-.048	-.058	-.052	-.106	-.180	-.178	-.160	-.176	-.170
Right side										
.025	.046	-.075	-.287	.180	.774	-.881	-.104	.399	.533	.640
.075	-.258	-.164	-.169	.125	.531	-.546	-.153	.173	.345	.518
.150	-.308	-.169	-.108	.114	.377	-.407	-.158	.066	.209	.381
.250	-.251	-.140	-.081	.093	.284	-.327	-.169	-.002	.116	.259
.350	-.238	-.151	-.092	.050	.186	-.299	-.185	-.056	.044	.171
.450	-.238	-.169	-.124	-.002	.103	-.285	-.207	-.097	-.024	.082
.550	-.210	-.162	-.137	-.024	.046	-.247	-.193	-.110	-.054	.030
.650	-.171	-.144	-.124	-.045	.014	-.192	-.155	-.097	-.058	.005
.750	-.142	-.124	-.110	-.056	-.011	-.119	-.104	-.058	-.038	-.004
.850	-.075	-.063	-.069	-.018	.010	-.037	-.028	.001	.005	.016
.900	-.053	-.039	-.049	-.006	.003	-.014	-.014	.008	.001	-.002
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	-.169	.578	-.137	-1.063	-1.126	.062	.508	-.235	-.646	-.777
.025	.356	-.128	-1.063	-1.294	-1.020	.153	-.227	-1.029	-.721	-.757
.075	.169	-.164	-.683	-1.272	-1.027	-.023	-.211	-.477	-.702	-.764
.150	.048	-.189	-.449	-1.272	-1.013	-.130	-.193	-.318	-.664	-.752
.250	.148	-.046	-.273	-.920	-1.031	-.091	-.115	-.235	-.585	-.707
.350	-.094	-.207	-.305	-.612	-1.020					
.450	-.137	-.216	-.275	-.287	-.925					
.550	-.149	-.202	-.241	-.240	-.791					
.650	-.123	-.167	-.185	-.192	-.625	-.130	-.095	-.155	-.435	-.587
.750	-.080	-.086	-.117	-.131	-.469	-.098	-.061	-.121	-.408	-.560
.850	-.041	-.050	-.060	-.072	-.337	-.066	-.023	-.092	-.358	-.521
.900	-.028	-.023	-.033	-.047	-.278	-.046	-.001	-.069	-.321	-.494
Right side										
.025	-1.238	-.128	.370	.527	.576	-.804	-.198	.186	.291	.325
.075	-.758	-.176	.159	.343	.477	-.607	-.229	-.049	.064	.152
.150	-.418	-.169	.055	.207	.343	-.391	-.202	-.135	-.086	-.020
.250	-.349	-.191	-.029	.093	.218					
.350	-.317	-.207	-.087	.007	.121					
.450	-.279	-.214	-.126	-.045	.046					
.550	-.240	-.193	-.137	-.077	-.004					
.650	-.180	-.151	-.108	-.072	-.024	-.128	-.088	-.106	-.133	-.158
.750	-.117	-.086	-.074	-.054	-.040	-.128	-.063	-.106	-.147	-.194
.850	-.032	-.014	-.008	-.002	-.027	-.066	.004	-.038	-.108	-.188
.900	-.005	.011	.019	.016	-.047	-.046	.017	-.026	-.115	-.208

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(*l*) $\alpha = 15.8°$; M = 0.80.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.370	.849	.728	.166	-.237	.308	.703	.377	-.450	-1.158
.025	-.842	-.178	.181	-.294	-1.468	.377	-.115	-1.429	-1.638	-1.272
.075	-.204	-.185	-.189	-.367	-.761	.175	-.170	-.582	-1.361	-1.279
.150	-.110	-.225	-.388	-.497	-.689	.080	-.185	-.449	-.741	-1.264
.250	-.092	-.190	-.293	-.389	-.456	-.003	-.201	-.375	-.523	-1.113
.350	-.122	-.167	-.263	-.320	-.402	-.087	-.241	-.363	-.462	-.786
.450	-.157	-.201	-.268	-.286	-.375	-.146	-.273	-.343	-.418	-.576
.550	-.188	-.229	-.262	-.257	-.374	-.158	-.256	-.292	-.385	-.506
.650	-.187	-.204	-.225	-.221	-.389	-.157	-.229	-.239	-.352	-.455
.750	-.198	-.198	-.204	-.201	-.338	-.119	-.163	-.165	-.297	-.359
.850	-.152	-.133	-.133	-.121	-.215	-.050	-.077	-.071	-.230	-.252
.900	-.110	-.080	-.079	-.069	-.140	-.170	-.187	-.160	-.145	-.143
	Right side									
.025	.293	-.078	-.807	.192	.715	-1.369	-.124	.411	.516	.667
.075	-.195	-.205	-.197	.120	.505	-.578	-.182	.184	.336	.521
.150	-.377	-.219	-.098	.107	.380	-.454	-.185	.078	.221	.386
.250	-.302	-.176	-.094	.093	.295	-.377	-.195	.004	.120	.274
.350	-.285	-.179	-.120	.051	.190	-.348	-.223	-.058	.048	.181
.450	-.270	-.204	-.154	-.006	.102	-.330	-.256	-.117	-.033	.081
.550	-.240	-.207	-.175	-.043	.043	-.288	-.246	-.142	-.077	.016
.650	-.207	-.182	-.172	-.066	.000	-.228	-.204	-.130	-.089	-.015
.750	-.188	-.175	-.174	-.086	-.042	-.145	-.136	-.085	-.065	-.030
.850	-.112	-.099	-.117	-.046	-.030	-.054	-.045	-.017	-.022	-.017
.900	-.075	-.065	-.088	-.031	-.026	-.017	-.023	-.006	-.021	-.039
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.083	.600	.118	-.594	-1.141	.154	.514	-.044	-.524	-.746
.025	.350	-.143	-1.292	-1.241	-.948	.152	-.282	-1.325	-.734	-.659
.075	.169	-.187	-.891	-1.198	-.930	-.048	-.285	-.784	-.713	-.654
.150	.045	-.217	-.546	-1.191	-.900	-.188	-.238	-.410	-.685	-.651
.250	-.101	-.163	-.339	-.826	-.871	-.112	-.134	-.305	-.626	-.639
.350	-.119	-.258	-.354	-.820	-.849					
.450	-.176	-.274	-.319	-.491	-.795					
.550	-.195	-.262	-.280	-.329	-.726					
.650	-.164	-.199	-.207	-.241	-.644	-.145	-.109	-.177	-.464	-.548
.750	-.107	-.119	-.132	-.165	-.561	-.110	-.063	-.136	-.427	-.531
.850	-.050	-.051	-.056	-.086	-.473	-.069	-.010	-.089	-.386	-.494
.900	-.024	-.020	-.020	-.054	-.431	-.041	.009	-.061	-.358	-.477
	Right side									
.025	-1.305	-.145	.377	.526	.601	-.863	-.246	.180	.315	.374
.075	-.882	-.201	.172	.345	.490	-.754	-.296	-.073	.082	.200
.150	-.517	-.195	.063	.218	.367	-.550	-.252	-.194	-.106	.003
.250	-.398	-.231	-.039	.098	.238					
.350	-.354	-.264	-.113	-.006	.132					
.450	-.309	-.262	-.162	-.072	.045					
.550	-.261	-.237	-.165	-.110	-.012					
.650	-.193	-.181	-.138	-.106	-.044	-.116	-.096	-.120	-.162	-.188
.750	-.116	-.115	-.095	-.086	-.069	-.133	-.069	-.110	-.177	-.228
.850	-.023	-.018	-.015	-.028	-.063	-.068	.009	-.042	-.133	-.219
.900	.011	.012	.015	-.012	-.090	-.044	.024	-.026	-.136	-.242

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE LOW WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Concluded

(m) $\alpha = 15.9^{\circ}$; M = 0.85.

$\dfrac{x}{c_v}$	C_p for $\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	C_p for $\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
		$z/b_v = 0.11$					$z/b_v = 0.38$			
Left side										
.000	.290	.880	.687	.128		.379	.723	.471	-.279	
.025	-.963	-.193	.289	-.208		.360	-.144	-1.276	-1.512	
.075	-.379	-.216	-.119	-.290		.163	-.204	-.612	-1.137	
.150	-.143	-.267	-.403	-.467		.064	-.209	-.557	-.856	
.250	-.092	-.220	-.334	-.491		-.021	-.220	-.397	-.677	
.350	-.120	-.189	-.297	-.385		-.105	-.268	-.409	-.519	
.450	-.157	-.223	-.310	-.320		-.170	-.312	-.385	-.474	
.550	-.188	-.252	-.303	-.284		-.187	-.293	-.335	-.457	
.650	-.198	-.228	-.263	-.262		-.188	-.264	-.284	-.419	
.750	-.213	-.234	-.265	-.259		-.149	-.189	-.200	-.348	
.850	-.162	-.153	-.174	-.145		-.069	-.087	-.096	-.270	
.900	-.118	-.095	-.112	-.085		-.179	-.192	-.171	-.147	
Right side										
.025	.387	-.103	-.975	.139		-1.257	-.112	.381	.481	
.075	-.129	-.247	-.376	.076		-.632	-.186	.151	.306	
.150	-.425	-.250	-.149	.080		-.502	-.196	.048	.193	
.250	-.339	-.204	-.113	.068		-.394	-.206	-.030	.102	
.350	-.308	-.192	-.146	.029		-.374	-.240	-.092	.022	
.450	-.289	-.220	-.187	-.026		-.350	-.279	-.161	-.060	
.550	-.259	-.226	-.209	-.062		-.308	-.277	-.197	-.110	
.650	-.223	-.203	-.209	-.097		-.245	-.237	-.184	-.126	
.750	-.215	-.211	-.219	-.124		-.156	-.156	-.132	-.104	
.850	-.126	-.121	-.149	-.077		-.061	-.062	-.053	-.055	
.900	-.089	-.081	-.120	-.060		-.023	-.030	-.036	-.046	
		$z/b_v = 0.66$					$z/b_v = 0.93$			
Left side										
.000	.156	.608	.224	-.413		.152	.509	.038	-.532	
.025	.331	-.180	-1.440	-1.324		.136	-.357	-1.351	-.808	
.075	.152	-.217	-1.214	-1.293		-.072	-.364	-1.040	-.787	
.150	.026	-.241	-.531	-1.157		-.257	-.277	-.594	-.753	
.250	-.332	-.275	-.386	-.796		-.129	-.152	-.265	-.695	
.350	-.145	-.299	-.363	-.852						
.450	-.213	-.320	-.355	-.640						
.550	-.240	-.301	-.318	-.430						
.650	-.194	-.223	-.239	-.289		-.157	-.118	-.191	-.494	
.750	-.129	-.138	-.152	-.191		-.120	-.070	-.143	-.447	
.850	-.064	-.056	-.067	-.111		-.071	-.016	-.096	-.396	
.900	-.035	-.022	-.030	-.077		-.042	.006	-.070	-.364	
Right side										
.025	-1.478	-.131	.344	.512		-1.360	-.251	.152	.305	
.075	-1.235	-.203	.138	.329		-.940	-.356	-.113	.066	
.150	-.479	-.209	.029	.197		-.583	-.281	-.304	-.150	
.250	-.427	-.252	-.075	.077						
.350	-.367	-.293	-.160	-.032						
.450	-.335	-.303	-.222	-.117						
.550	-.288	-.274	-.231	-.158						
.650	-.214	-.207	-.187	-.153		-.153	-.105	-.139	-.194	
.750	-.129	-.129	-.126	-.127		-.137	-.081	-.128	-.209	
.850	-.030	-.022	-.033	-.058		-.061	.006	-.047	-.155	
.900	.007	.011	-.006	-.043		-.038	.022	-.030	-.161	

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL

(a) $\alpha = 0°$; M = 0.60.

$\frac{x}{c_v}$	C_p for– $z/b_v = 0.11$					C_p for– $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.608	.653	.363	-.424	-1.296	.279	.732	.320	-.426	-1.210
.025	.273	-.089	-.541	-1.138	-2.278	.334	-.087	-.628	-1.067	-1.158
.075	.134	-.105	-.369	-.680	-1.585	.148	-.125	-.437	-.943	-1.112
.150	.039	-.128	-.299	-.478	-.662	.053	-.132	-.321	-.716	-1.060
.250	.021	-.092	-.199	-.301	-.425	-.011	-.141	-.263	-.439	-1.011
.350	-.013	-.098	-.167	-.235	-.341	-.058	-.157	-.251	-.335	-.868
.450	-.047	-.110	-.156	-.201	-.311	-.104	-.177	-.228	-.278	-.669
.550	-.061	-.103	-.131	-.163	-.321	-.081	-.141	-.174	-.217	-.508
.650	-.031	-.067	-.077	-.122	-.239	-.063	-.101	-.122	-.151	-.391
.750	-.013	-.035	-.045	-.090	-.137	-.027	-.051	-.063	-.083	-.271
.850	.021	.010	.010	-.033	-.090	.030	.019	.016	-.006	-.151
.900	.044	.046	.035	.005	-.042	.055	.060	.055	.037	-.085
Right side										
.025	-.451	-.028	.320	.611	.840	-.612	-.060	.370	.601	.750
.075	-.371	-.107	.125	.347	.566	-.417	-.116	.159	.366	.541
.150	-.283	-.119	.048	.214	.392	-.312	-.128	.053	.225	.383
.250	-.197	-.096	.025	.146	.284	-.253	-.139	-.004	.128	.256
.350	-.176	-.107	-.022	.082	.186	-.226	-.141	-.049	.057	.168
.450	-.154	-.114	-.052	.030	.121	-.208	-.153	-.086	.003	.098
.550	-.113	-.094	-.052	.016	.093	-.172	-.132	-.083	-.011	.053
.650	-.072	-.053	-.024	.037	.089	-.115	-.089	-.058	-.006	.039
.750	-.029	-.026	.001	.039	.084	-.045	-.031	-.018	.023	.039
.850	.025	.030	.044	.075	.100	.032	.039	.048	.066	.057
.900	.048	.055	.050	.073	.089	.057	.055	.053	.066	.044

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.239	.721	.261	-.430	-.678	.393	.676	.223	-.260	-.472
.025	.345	-.114	-.836	-1.120	-.737	.202	-.202	-.693	-.800	-.524
.075	.159	-.139	-.492	-1.040	-.714	.028	-.168	-.394	-.775	-.533
.150	.039	-.155	-.374	-.886	-.707	-.049	-.139	-.274	-.702	-.569
.250	-.022	-.159	-.285	-.609	-.671	-.027	-.083	-.199	-.560	-.558
.350	-.079	-.171	-.253	-.389	-.642					
.450	-.108	-.171	-.228	-.281	-.615					
.550	-.111	-.153	-.192	-.217	-.569					
.650	-.079	-.110	-.129	-.154	-.520	-.074	-.069	-.113	-.371	-.407
.750	-.045	-.044	-.061	-.088	-.461	-.047	-.035	-.072	-.337	-.384
.850	.012	.010	.001	-.029	-.388	-.024	-.004	-.056	-.299	-.352
.900	.030	.028	.030	-.002	-.352	-.015	.021	-.049	-.263	-.341
Right side										
.025	-.709	-.074	.388	.588	.698	-.566	-.130	.239	.388	.469
.075	-.471	-.137	.152	.357	.508	-.380	-.173	.016	.128	.229
.150	-.331	-.134	.048	.211	.354	-.267	-.146	-.054	.003	.073
.250	-.278	-.155	-.020	.096	.225					
.350	-.249	-.168	-.081	.016	.121					
.450	-.217	-.164	-.104	-.024	.053					
.550	-.174	-.139	-.099	-.047	.012					
.650	-.111	-.103	-.077	-.027	-.006	-.083	-.069	-.072	-.077	-.081
.750	-.049	-.033	-.029	-.006	-.024	-.070	-.040	-.061	-.074	-.119
.850	.028	.033	.037	.041	-.026	-.020	.019	-.009	-.031	-.115
.900	.055	.062	.057	.053	-.060	-.011	.035	.005	-.033	-.142

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(b) $\alpha = 0°$; $M = 0.80$.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.721	.677	.550	.198	-.299	.472	.773	.534	.028	-.591
.025	.284	-.066	-.541	-1.146	-1.535	.341	-.073	-1.067	-1.715	-1.029
.075	.151	-.096	-.369	-.897	-1.278	.158	-.129	-.445	-1.511	-1.009
.150	.052	-.134	-.319	-.541	-1.012	.063	-.140	-.337	-.534	-.947
.250	.030	-.100	-.209	-.305	-.665	-.010	-.152	-.282	-.385	-.869
.350	-.013	-.108	-.174	-.244	-.433	-.072	-.174	-.264	-.348	-.775
.450	-.059	-.129	-.164	-.216	-.351	-.123	-.197	-.240	-.293	-.662
.550	-.072	-.117	-.133	-.180	-.299	-.106	-.159	-.179	-.218	-.551
.650	-.037	-.073	-.083	-.137	-.222	-.079	-.112	-.118	-.148	-.460
.750	-.019	-.040	-.038	-.094	-.097	-.032	-.044	-.048	-.068	-.360
.850	.024	.021	.022	-.028	-.129	.037	.034	.034	.019	-.267
.900	.049	.057	.053	.021	-.076	.072	.075	.078	.059	-.208
	Right side									
.025	-.413	.000	.338	.612	.839	-.978	-.052	.379	.605	.753
.075	-.391	-.099	.145	.357	.569	-.426	-.120	.166	.365	.549
.150	-.303	-.115	.063	.221	.402	-.326	-.132	.069	.229	.391
.250	-.209	-.097	.028	.152	.292	-.265	-.144	.000	.128	.262
.350	-.184	-.111	-.018	.077	.190	-.244	-.161	-.057	.051	.175
.450	-.162	-.123	-.053	.022	.116	-.221	-.170	-.092	-.010	.084
.550	-.117	-.099	-.051	.008	.087	-.173	-.146	-.095	-.034	.037
.650	-.067	-.055	-.018	.027	.085	-.105	-.097	-.059	-.021	.023
.750	-.029	-.023	-.001	.036	.070	-.026	-.022	-.009	.011	.022
.850	.034	.040	.047	.068	.076	.057	.059	.061	.062	.026
.900	.069	.062	.061	.074	.066	.083	.074	.070	.068	-.003
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.446	.748	.482	-.013	-.427	.487	.702	.386	-.016	-.311
.025	.348	-.112	-1.117	-1.334	-.699	.193	-.221	-1.056	-.781	-.483
.075	.163	-.150	-.556	-1.201	-.697	.006	-.197	-.504	-.753	-.481
.150	.048	-.173	-.401	-.982	-.697	-.070	-.159	-.298	-.735	-.507
.250	-.032	-.182	-.305	-.771	-.665	-.037	-.090	-.214	-.623	-.509
.350	-.096	-.194	-.267	-.556	-.623					
.450	-.132	-.199	-.238	-.369	-.583					
.550	-.128	-.171	-.194	-.244	-.541					
.650	-.088	-.117	-.121	-.143	-.503	-.073	-.059	-.106	-.378	-.404
.750	-.040	-.034	-.047	-.063	-.454	-.040	-.019	-.064	-.334	-.380
.850	.031	.028	.022	-.001	-.404	-.007	.024	-.045	-.282	-.354
.900	.046	.056	.050	.025	-.378	.015	.045	-.032	-.244	-.343
	Right side									
.025	-1.117	-.066	.391	.603	.718	-1.076	-.150	.228	.392	.505
.075	-.503	-.143	.164	.363	.520	-.480	-.205	-.012	.119	.256
.150	-.353	-.140	.063	.222	.368	-.267	-.161	-.074	-.025	.081
.250	-.302	-.170	-.027	.100	.227					
.350	-.268	-.188	-.089	.008	.119					
.450	-.227	-.188	-.121	-.044	.038					
.550	-.177	-.158	-.114	-.068	-.012					
.650	-.105	-.105	-.080	-.050	-.032	-.041	-.056	-.065	-.077	-.098
.750	-.035	-.022	-.032	-.019	-.048	-.061	-.025	-.054	-.073	-.138
.850	.052	.063	.052	.044	'-.048	.000	.049	.015	-.025	-.123
.900	.087	.084	.076	.062	-.085	.009	.063	.023	-.031	-.155

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(c) $\alpha = 0°$; $M = 0.85$.

$\dfrac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.744	.682	.601	.317	-.026	.527	.785	.592	.173	-.292
.025	.280	-.061	-.533	-1.086	-1.135	.330	-.075	-1.115	-1.551	-.784
.075	.148	-.088	-.379	-.867	-.943	.152	-.136	-.413	-1.400	-.803
.150	.045	-.136	-.344	-.692	-.798	.054	-.152	-.347	-1.226	-.749
.250	.018	-.108	-.227	-.350	-.623	-.019	-.163	-.300	-.334	-.689
.350	-.027	-.122	-.189	-.256	-.438	-.087	-.190	-.284	-.291	-.620
.450	-.073	-.139	-.181	-.210	-.319	-.145	-.221	-.259	-.277	-.528
.550	-.087	-.130	-.148	-.169	-.241	-.130	-.179	-.195	-.217	-.444
.650	-.047	-.082	-.095	-.139	-.161	-.101	-.127	-.127	-.149	-.364
.750	-.033	-.047	-.047	-.090	-.132	-.044	-.054	-.054	-.065	-.275
.850	.014	.013	.018	-.023	-.052	.028	.030	.031	.026	-.188
.900	.042	.055	.057	.021	.000	.068	.073	.079	.072	-.134
Right side										
.025	-.390	.010	.344	.615	.865	-.999	-.048	.375	.598	.790
.075	-.406	-.099	.148	.361	.618	-.454	-.125	.162	.367	.595
.150	-.322	-.122	.062	.229	.466	-.345	-.139	.064	.232	.452
.250	-.229	-.108	.024	.157	.362	-.286	-.156	-.009	.126	.336
.350	-.198	-.126	-.030	.076	.266	-.262	-.176	-.068	.046	.246
.450	-.175	-.137	-.071	.015	.189	-.238	-.188	-.114	-.025	.157
.550	-.127	-.113	-.067	.002	.163	-.187	-.162	-.118	-.045	.114
.650	-.079	-.062	-.034	.024	.157	-.114	-.113	-.075	-.032	.097
.750	-.035	-.034	-.013	.031	.144	-.032	-.031	-.021	.006	.093
.850	.034	.033	.040	.066	.142	.051	.055	.054	.063	.094
.900	.062	.061	.051	.066	.131	.084	.073	.071	.069	.066

$\dfrac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.490	.760	.538	.120	-.201	.505	.707	.430	.029	-.176
.025	.338	-.119	-1.212	-1.645	-.600	.177	-.215	-.706	-.758	-.593
.075	.153	-.160	-.622	-1.465	-.608	-.022	-.235	-.605	-.851	-.390
.150	.037	-.186	-.415	-1.247	-.618	-.103	-.179	-.307	-.840	-.427
.250	-.056	-.201	-.315	-.620	-.583	-.052	-.103	-.220	-.713	-.421
.350	-.117	-.218	-.280	-.475	-.526					
.450	-.157	-.224	-.254	-.318	-.477					
.550	-.153	-.193	-.203	-.209	-.426					
.650	-.111	-.133	-.128	-.126	-.385	-.084	-.074	-.112	-.415	-.313
.750	-.053	-.040	-.043	-.052	-.331	-.046	-.024	-.070	-.364	-.294
.850	.022	.024	.030	.015	-.275	-.013	.023	-.051	-.304	
.900	.045	.058	.061	.042	-.250	.010	.047	-.036	-.266	-.253
Right side										
.025	-1.178	-.064	.383	.592	.754	-1.167	-.170	.213	.393	.558
.075	-.421	-.147	.159	.357	.570	-.601	-.235	-.043	.106	.323
.150	-.373	-.149	.057	.217	.430	-.237	-.181	-.107	-.059	.152
.250	-.316	-.190	-.040	.095	.298					
.350	-.285	-.210	-.108	-.004	.193					
.450	-.244	-.210	-.145	-.066	.111					
.550	-.192	-.177	-.139	-.083	.061					
.650	-.116	-.119	-.098	-.060	.040	-.091	-.068	-.071	-.085	-.024
.750	-.037	-.028	-.038	-.026	.029	-.074	-.030	-.057	-.078	-.056
.850	.058	.055	.051	.046	.036	-.005	.048	.016	-.021	-.038
.900	.086	.084	.077	.066	.000	.001	.063	.025	-.029	-.065

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(d) $\alpha = 0^{\circ}$; M = 0.90.

$\frac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.777	.688	.649	.416	.042	.590	.794	.652	.293	-.220
.025	.297	-.051	-.486	-.993	-1.168	.340	-.074	-.996	-1.329	-.811
.075	.169	-.088	-.392	-.785	-.980	.162	-.141	-.673	-1.207	-.816
.150	.063	-.146	-.371	-.684	-.838	.068	-.162	-.375	-1.089	-.795
.250	.032	-.121	-.242	-.481	-.717	-.009	-.183	-.313	-.579	-.759
.350	-.018	-.138	-.209	-.312	-.598	-.085	-.216	-.335	-.398	-.723
.450	-.073	-.161	-.200	-.242	-.514	-.155	-.261	-.274	-.235	-.658
.550	-.092	-.150	-.154	-.163	-.434	-.140	-.206	-.200	-.159	-.582
.650	-.049	-.098	-.099	-.120	-.353	-.108	-.147	-.130	-.108	-.510
.750	-.033	-.064	-.047	-.074	-.313	-.041	-.056	-.051	-.046	-.432
.850	.020	.005	.024	-.005	-.215	.039	.032	.041	.042	-.358
.900	.052	.054	.062	.035	-.152	.083	.080	.087	.080	-.301
Right side										
.025	-.339	.020	.349	.620	.852	-.913	-.048	.369	.596	.763
.075	-.397	-.095	.151	.366	.582	-.569	-.126	.162	.363	.554
.150	-.333	-.123	.062	.234	.422	-.361	-.147	.060	.232	.402
.250	-.227	-.111	.023	.160	.305	-.289	-.167	-.017	.123	.271
.350	-.200	-.137	-.036	.073	.200	-.281	-.200	-.082	.035	.176
.450	-.175	-.153	-.080	.010	.114	-.236	-.218	-.134	-.035	.076
.550	-.124	-.125	-.083	-.008	.073	-.178	-.189	-.142	-.059	.020
.650	-.063	-.071	-.046	.021	.073	-.103	-.125	-.096	-.042	-.003
.750	-.020	-.043	-.029	.021	.045	-.014	-.029	-.031	.000	-.012
.850	.050	.031	.028	.060	.037	.071	.056	.052	.063	-.016
.900	.083	.058	.044	.067	.010	.105	.078	.071	.069	-.055
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.540	.771	.600	.241	-.169	.542	.708	.487	.122	-.204
.025	.347	-.117	-1.114	-1.414	-.741	-.521	-.506	-.507	-.530	-.497
.075	.166	-.167	-.918	-1.288	-.754	-.037	-.327	-1.020	-1.278	-.504
.150	.044	-.206	-.398	-1.173	-.767	-.152	-.209	-.411	-1.032	-.541
.250	-.053	-.232	-.406	-1.071	-.725	-.059	-.117	-.142	-.671	-.546
.350	-.131	-.256	-.237	-.548	-.663					
.450	-.179	-.260	-.250	-.172	-.607					
.550	-.170	-.220	-.212	-.120	-.574					
.650	-.115	-.154	-.129	-.081	-.533	-.076	-.075	-.115	-.343	-.454
.750	-.048	-.031	-.040	-.028	-.486	-.033	-.017	-.067	-.323	-.429
.850	.043	.035	.039	.031	-.429	.012	.031	-.048	-.281	-.406
.900	.066	.066	.071	.054	-.400	.034	.056	-.035	-.237	-.389
Right side										
.025	-1.061	-.063	.376	.582	.723	-1.077	-.183	.198	.382	.518
.075	-.835	-.154	.154	.345	.522	-.887	-.315	-.074	.094	.261
.150	-.362	-.162	.051	.213	.375	-.331	-.208	-.189	-.101	.064
.250	-.363	-.216	-.056	.084	.233					
.350	-.251	-.249	-.142	-.023	.113					
.450	-.228	-.241	-.182	-.098	.016					
.550	-.182	-.204	-.169	-.112	-.048					
.650	-.100	-.130	-.115	-.077	-.068	-.073	-.068	-.076	-.096	-.136
.750	-.018	-.031	-.043	-.032	-.091	-.059	-.028	-.054	-.077	-.180
.850	.076	.060	.055	.052	-.072	.010	.056	.025	-.012	-.148
.900	.113	.092	.079	.069	-.105	.020	.070	.032	-.017	-.175

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(e) $\alpha = 0^{\circ}$; M = 0.92.

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.783	.687	.665	.465	.087	.605	.799	.674	.352	-.172
.025	.283	-.046	-.457	-.962	-1.138	.330	-.072	-.959	-1.282	-.756
.075	.157	-.084	-.376	-.752	-.949	.150	-.145	-.723	-1.171	-.783
.150	.046	-.148	-.363	-.673	-.806	.052	-.166	-.408	-1.075	-.756
.250	.018	-.119	-.252	-.558	-.687	-.041	-.181	-.311	-.649	-.704
.350	-.035	-.136	-.211	-.382	-.584	-.112	-.225	-.338	-.534	-.705
.450	-.098	-.171	-.208	-.321	-.522	-.191	-.283	-.326	-.437	-.647
.550	-.125	-.165	-.159	-.226	-.454	-.179	-.221	-.181	-.165	-.576
.650	-.079	-.103	-.092	-.134	-.375	-.139	-.153	-.122	-.076	-.517
.750	-.057	-.064	-.039	-.076	-.337	-.067	-.058	-.042	-.024	-.440
.850	-.006	.006	.030	-.007	-.245	.023	.034	.047	.052	-.372
.900	.033	.053	.064	.035	-.176	.073	.087	.096	.093	-.316
Right side										
.025	-.335	.028	.352	.618	.849	-.860	-.041	.367	.597	.770
.075	-.407	-.090	.159	.367	.585	-.660	-.124	.164	.364	.553
.150	-.363	-.120	.071	.236	.420	-.409	-.149	.065	.224	.402
.250	-.264	-.114	.025	.158	.312	-.306	-.170	-.016	.120	.274
.350	-.230	-.144	-.035	.068	.204	-.324	-.209	-.084	.031	.177
.450	-.207	-.165	-.082	-.003	.112	-.303	-.238	-.146	-.049	.076
.550	-.142	-.140	-.088	-.021	.074	-.179	-.202	-.153	-.080	.019
.650	-.076	-.080	-.043	.005	.071	-.110	-.135	-.106	-.066	-.010
.750	-.031	-.049	-.033	.008	.044	-.025	-.030	-.032	-.013	-.019
.850	.039	.027	.032	.051	.024	.065	.058	.053	.052	-.024
.900	.067	.060	.045	.055	-.006	.093	.084	.074	.069	-.062
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.558	.777	.626	.303	-.122	.557	.712	.512	.182	-.176
.025	.330	-.116	-1.071	-1.359	-.732	-.451	-.449	-.446	-.444	-.439
.075	.143	-.171	-.897	-1.226	-.739	-.059	-.313	-.990	-1.264	-.503
.150	.020	-.208	-.637	-1.156	-.754	-.223	-.230	-.732	-1.070	-.533
.250	-.075	-.244	-.385	-1.109	-.726	-.122	-.115	-.157	-.981	-.545
.350	-.173	-.294	-.382	-1.029	-.665					
.450	-.230	-.277	-.193	-.442	-.601					
.550	-.208	-.236	-.197	-.071	-.568					
.650	-.142	-.148	-.119	-.024	-.542	-.091	-.072	-.110	-.468	-.458
.750	-.050	-.030	-.030	.010	-.494	-.040	-.021	-.065	-.384	-.435
.850	.026	.041	.045	.057	-.439	-.001	.035	-.046	-.315	-.410
.900	.053	.071	.077	.073	-.411	.024	.060	-.030	-.274	-.394
Right side										
.025	-1.001	-.059	.374	.582	.725	-1.045	-.183	.197	.381	.519
.075	-.830	-.153	.155	.350	.530	-.960	-.358	-.077	.094	.266
.150	-.511	-.168	.053	.212	.377	-.649	-.228	-.214	-.108	.066
.250	-.380	-.223	-.062	.080	.234					
.350	-.404	-.286	-.155	-.034	.110					
.450	-.196	-.258	-.204	-.118	.016					
.550	-.187	-.214	-.181	-.146	-.054					
.650	-.112	-.135	-.120	-.104	-.078	-.096	-.064	-.077	-.102	-.141
.750	-.027	-.026	-.042	-.045	-.096	-.068	-.026	-.049	-.084	-.192
.850	.067	.070	.062	.050	-.079	-.003	.064	.032	-.011	-.150
.900	.101	.099	.086	.075	-.113	.006	.077	.040	-.008	-.179

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(f) $\alpha = 9.4°$; M = 0.60.

$\frac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.512	.797	.500	-.461	-1.196	.233	.660	.112	-.775	-1.558
.025	-.044	-.145	-.274	-.822	-2.592	.287	-.143	-.770	-1.218	-1.385
.075	-.001	-.134	-.324	-.534	-.845	.116	-.161	-.484	-1.100	-1.373
.150	-.026	-.154	-.313	-.432	-.619	.028	-.161	-.356	-.877	-1.455
.250	-.021	-.107	-.226	-.304	-.412	-.026	-.165	-.299	-.407	-1.194
.350	-.055	-.116	-.201	-.257	-.364	-.080	-.188	-.285	-.318	-.717
.450	-.073	-.139	-.199	-.234	-.351	-.116	-.203	-.265	-.293	-.412
.550	-.087	-.141	-.187	-.211	-.296	-.109	-.174	-.212	-.238	-.330
.650	-.073	-.114	-.153	-.184	-.232	-.089	-.141	-.160	-.186	-.269
.750	-.057	-.090	-.121	-.134	-.178	-.060	-.094	-.107	-.120	-.198
.850	-.019	-.047	-.062	-.068	-.107	.001	-.018	-.023	-.029	-.102
.900	.010	-.007	-.018	-.025	-.057	.033	.018	.016	.007	-.059
Right side										
.025	-.172	-.027	.007	.546	.806	-.600	-.040	.379	.578	.690
.075	-.294	-.119	.023	.332	.538	-.404	-.110	.162	.362	.524
.150	-.271	-.123	.009	.223	.383	-.312	-.123	.064	.225	.378
.250	-.208	-.094	-.005	.164	.289	-.265	-.141	.002	.130	.251
.350	-.195	-.110	-.034	.084	.189	-.244	-.157	-.046	.062	.169
.450	-.193	-.127	-.057	.032	.119	-.240	-.170	-.085	.005	.098
.550	-.168	-.119	-.062	.009	.087	-.204	-.157	-.091	-.025	.052
.650	-.134	-.094	-.053	.007	.075	-.150	-.123	-.071	-.020	.043
.750	-.098	-.081	-.034	.009	.059	-.078	-.061	-.034	.000	.039
.850	-.033	-.020	.007	.039	.078	.001	.006	.025	.046	.057
.900	-.006	.000	.018	.041	.068	.026	.020	.032	.046	.039
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.132	.618	-.023	-.779	-1.112	.244	.558	-.046	-.461	-.824
.025	.285	-.188	-1.064	-1.263	-.932	.148	-.264	-.888	-.711	-.617
.075	.114	-.181	-.553	-1.225	-.913	-.026	-.210	-.489	-.700	-.613
.150	.010	-.188	-.413	-1.152	-.891	-.111	-.183	-.317	-.663	-.617
.250	-.055	-.183	-.327	-.838	-.847	-.069	-.121	-.226	-.570	-.594
.350	-.100	-.197	-.283	-.434	-.822					
.450	-.132	-.201	-.263	-.263	-.768					
.550	-.134	-.188	-.226	-.222	-.692					
.650	-.109	-.141	-.160	-.166	-.588	-.100	-.081	-.137	-.382	-.476
.750	-.060	-.083	-.096	-.102	-.481	-.075	-.047	-.110	-.336	-.449
.850	-.012	-.020	-.037	-.047	-.378	-.042	-.009	-.087	-.286	-.424
.900	.010	.002	.000	-.020	-.335	-.017	.009	-.064	-.259	-.408
Right side										
.025	-.699	-.054	.372	.562	.629	-.596	-.132	.201	.323	.383
.075	-.465	-.130	.160	.353	.481	-.388	-.188	.018	.089	.187
.150	-.330	-.139	.059	.221	.340	-.267	-.170	-.101	-.047	.027
.250	-.289	-.165	-.023	.107	.217					
.350	-.269	-.177	-.080	.028	.119					
.450	-.235	-.183	-.110	-.022	.057					
.550	-.195	-.165	-.114	-.047	.007					
.650	-.138	-.134	-.091	-.038	-.009	-.048	-.078	-.107	-.107	-.127
.750	-.078	-.054	-.057	-.018	-.016	-.078	-.061	-.091	-.113	-.159
.850	.008	.011	.023	.025	-.009	-.021	.006	-.032	-.070	-.159
.900	.035	.038	.036	.043	-.030	-.008	.020	-.021	-.075	-.184

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(g) $\alpha = 9.6°$; M = 0.80.

$\dfrac{x}{c_v}$	C_p for —					C_p for —				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.482	.844	.492	-.106	-.123	.432	.701	.413	-.156	-.769
.025	-.075	-.126	-.185	-.635	-1.586	.269	-.126	-1.003	-1.437	-1.104
.075	-.012	-.130	-.337	-.529	-1.079	.107	-.157	-.514	-1.313	-1.065
.150	-.040	-.166	-.362	-.501	-.644	.022	-.154	-.383	-1.012	-1.046
.250	-.034	-.117	-.258	-.355	-.435	-.039	-.168	-.322	-.383	-.982
.350	-.062	-.123	-.224	-.290	-.370	-.095	-.209	-.311	-.331	-.811
.450	-.091	-.148	-.223	-.260	-.351	-.147	-.225	-.293	-.325	-.589
.550	-.106	-.160	-.209	-.234	-.304	-.142	-.198	-.243	-.272	-.437
.650	-.089	-.138	-.172	-.203	-.277	-.121	-.160	-.189	-.211	-.367
.750	-.078	-.117	-.145	-.159	-.243	-.077	-.100	-.119	-.136	-.304
.850	-.036	-.053	-.072	-.082	-.162	-.006	-.011	-.025	-.042	-.207
.900	-.002	-.005	-.025	-.035	-.104	.037	.035	.024	.011	-.138
Right side										
.025	-.080	.001	-.031	.500	.781	-.823	-.035	.363	.573	.708
.075	-.304	-.110		.321	.527	-.417	-.109	.157	.354	.527
.150	-.303	-.126	-.002	.219	.382	-.338	-.127	.064	.227	.383
.250	-.230	-.098	-.025	.163	.290	-.286	-.148	-.005	.128	.269
.350	-.219	-.118	-.049	.084	.190	-.276	-.172	-.061	.058	.178
.450	-.215	-.141	-.080	.022	.109	-.263	-.191	-.105	-.016	.081
.550	-.186	-.136	-.080	-.009	.065	-.226	-.178	-.121	-.050	.032
.650	-.151	-.112	-.067	-.007	.051	-.165	-.144	-.095	-.045	.012
.750	-.124	-.098	-.055	-.015	.029	-.087	-.068	-.049	-.019	.006
.850	-.048	-.029	-.009	.017	.035	.004	.012	.018	.032	.016
.900	-.007	.001	.009	.028	.030	.029	.029	.030	.037	-.006
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.324	.652	.273	-.214	-.686	.336	.577	.157	-.217	-.618
.025	.275	-.181	-1.166	-.969	-.781	.114	-.284	-.869	-.618	-.544
.075	.108	-.188	-.708	-.934	-.755	-.056	-.251	-.691	-.620	-.546
.150	.002	-.200	-.456	-.907	-.724	-.139	-.198	-.415	-.629	-.549
.250	-.068	-.203	-.349	-.872	-.695	-.080	-.121	-.270	-.579	-.549
.350	-.125	-.221	-.307	-.744	-.677					
.450	-.163	-.227	-.279	-.561	-.653					
.550	-.165	-.209	-.233	-.398	-.613					
.650	-.127	-.145	-.165	-.266	-.573	-.103	-.068	-.139	-.363	-.473
.750	-.065	-.068	-.084	-.156	-.523	-.068	-.026	-.101	-.326	-.447
.850	-.009	-.002	-.017	-.068	-.462	-.025	.013	-.067	-.287	-.419
.900	.020	.029	.020	-.032	-.428	-.001	.036	-.040	-.269	-.399
Right side										
.025	-.952	-.049	.367	.560	.657	-.999	-.154	.187	.335	.429
.075	-.502	-.132	.157	.350	.500	-.470	-.225	-.051	.088	.223
.150	-.364	-.142	.058	.219	.362	-.286	-.195	-.145	-.065	.036
.250	-.320	-.180	-.040	.101	.231					
.350	-.294	-.203	-.105	.008	.125					
.450	-.260	-.210	-.142	-.051	.044					
.550	-.212	-.188	-.145	-.079	-.015					
.650	-.144	-.147	-.108	-.068	-.038	-.060	-.062	-.099	-.123	-.147
.750	-.068	-.050	-.060	-.044	-.055	-.069	-.036	-.081	-.126	-.194
.850	.020	.032	.021	.015	-.052	-.007	.036	-.014	-.082	-.184
.900	.057	.056	.042	.035	-.083	.011	.050	.000	-.095	-.206

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(h) $\alpha = 9.7°$; $M = 0.85$.

$\frac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.418	.864	.465	-.111	.039	.452	.709	.488	.004	-.609
.025	-.137	-.131	-.120	-.557	-1.437	.244	-.138	-1.180	-1.576	-1.153
.075	-.054	-.145	-.325	-.513	-1.039	.075	-.179	-.476	-1.240	-1.103
.150	-.080	-.192	-.393	-.554	-.723	-.010	-.179	-.391	-.776	-1.036
.250	-.080	-.139	-.268	-.414	-.501	-.083	-.196	-.341	-.543	-.923
.350	-.109	-.144	-.239	-.327	-.398	-.140	-.237	-.331	-.407	-.783
.450	-.135	-.176	-.242	-.280	-.373	-.202	-.267	-.312	-.373	-.636
.550	-.144	-.189	-.225	-.248	-.328	-.196	-.238	-.256	-.301	-.524
.650	-.127	-.162	-.185	-.216	-.301	-.176	-.196	-.200	-.224	-.447
.750	-.127	-.145	-.159	-.163	-.291	-.127	-.122	-.125	-.141	-.371
.850	-.077	-.074	-.080	-.086	-.209	-.042	-.030	-.026	-.043	-.269
.900	-.041	-.020	-.030	-.037	-.144	-.003	.025	.026	.013	-.198
	Right side									
.025	-.061	-.008	-.056	.463	.767	-.949	-.056	.349	.564	.704
.075	-.348	-.135	-.004	.310	.517	-.502	-.134	.144	.344	.523
.150	-.377	-.149	-.012	.210	.372	-.391	-.151	.053	.216	.379
.250	-.292	-.122	-.030	.152	.282	-.348	-.183	-.017	.116	.257
.350	-.278	-.146	-.064	.070	.178	-.336	-.204	-.074	.043	.169
.450	-.263	-.168	-.090	.006	.093	-.319	-.228	-.123	-.038	.070
.550	-.231	-.165	-.090	-.023	.050	-.275	-.219	-.137	-.068	.015
.650	-.192	-.135	-.076	-.024	.037	-.212	-.180	-.115	-.066	-.012
.750	-.170	-.125	-.064	-.033	.005	-.128	-.095	-.062	-.034	-.019
.850	-.086	-.050	-.013	.010	.012	-.031	-.004	.013	.022	-.015
.900	-.047	-.018	.009	.022	-.001	.000	.018	.024	.027	-.039

$\frac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.344	.658	.352	-.120	-.599	.331	.574	.197	-.258	-.597
.025	.251	-.190	-1.263	-1.598	-.791	.077	-.339	-1.112	-.821	-.553
.075	.084	-.209	-.705	-1.387	-.757	-.115	-.333	-.741	-.821	-.556
.150	-.037	-.230	-.481	-1.100	-.722	-.224	-.241	-.434	-.814	-.566
.250	-.119	-.241	-.366	-.836	-.701	-.135	-.156	-.270	-.701	-.570
.350	-.185	-.267	-.319	-.571	-.677					
.450	-.233	-.277	-.295	-.370	-.655					
.550	-.234	-.252	-.250	-.268	-.621					
.650	-.185	-.176	-.176	-.176	-.592	-.146	-.093	-.142	-.396	-.497
.750	-.118	-.088	-.092	-.094	-.548	-.103	-.043	-.103	-.340	-.474
.850	-.048	-.013	-.020	-.026	-.491	-.060	.005	-.064	-.281	-.446
.900	-.016	.021	.020	.014	-.471	-.037	.030	-.042	-.248	-.429
	Right side									
.025	-1.204	-.071	.355	.547	.656	-1.224	-.194	.176	.339	.430
.075	-.537	-.159	.142	.337	.495	-.647	-.293	-.080	.082	.223
.150	-.445	-.172	.047	.207	.359	-.327	-.235	-.182	-.106	.019
.250	-.397	-.219	-.056	.087	.223					
.350	-.358	-.250	-.123	-.013	.110					
.450	-.324	-.260	-.167	-.080	.023					
.550	-.274	-.231	-.167	-.108	-.045					
.650	-.195	-.182	-.125	-.093	-.069	-.135	-.088	-.109	-.130	-.182
.750	-.111	-.071	-.067	-.058	-.090	-.112	-.054	-.084	-.128	-.225
.850	-.009	.019	.019	.012	-.080	-.038	.025	-.010	-.071	-.209
.900	.020	.045	.040	.033	-.114	-.028	.040	.006	-.071	-.235

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(i) $\alpha = 9.7°$; M = 0.90.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.340	.876	.417	-.021		.521	.696	.564	.145	
.025	-.149	-.128	.002	-.448		.254	-.156	-1.015	-1.357	
.075	-.035	-.162	-.231	-.435		.086	-.212	-.492	-1.211	
.150	-.055	-.228	-.392	-.536		.004	-.214	-.486	-.750	
.250	-.064	-.177	-.301	-.482		-.061	-.225	-.358	-.667	
.350	-.094	-.185	-.264	-.406		-.129	-.270	-.378	-.529	
.450	-.123	-.227	-.277	-.347		-.203	-.324	-.393	-.365	
.550	-.137	-.258	-.271	-.276		-.208	-.333	-.290	-.336	
.650	-.125	-.232	-.213	-.242		-.186	-.322	-.221	-.277	
.750	-.121	-.241	-.195	-.207		-.123	-.161	-.139	-.172	
.850	-.060	-.114	-.099	-.112		-.025	-.032	-.030	-.056	
.900	-.018	-.040	-.040	-.052		.021	.024	.026	.000	
Right side										
.025	.062	-.019	-.095	.436		-.844	-.072	.327	.555	
.075	-.266	-.164	-.008	.298		-.535	-.161	.126	.338	
.150	-.375	-.189	-.011	.203		-.415	-.182	.034	.211	
.250	-.289	-.162	-.036	.144		-.337	-.204	-.032	.109	
.350	-.271	-.187	-.072	.059		-.358	-.242	-.097	.030	
.450	-.274	-.211	-.101	-.011		-.361	-.292	-.157	-.059	
.550	-.240	-.223	-.109	-.038		-.271	-.308	-.184	-.095	
.650	-.189	-.202	-.099	-.039		-.207	-.290	-.159	-.097	
.750	-.172	-.208	-.095	-.056		-.111	-.140	-.089	-.063	
.850	-.071	-.092	-.032	-.013		-.009	-.015	-.003	.000	
.900	-.028	-.046	-.010	-.001		.027	.016	.012	.008	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.406	.646	.435	.020		.378	.569	.273	-.145	
.025	.263	-.200	-.169	-1.427		.089	-.388	-1.203	-.960	
.075	.094	-.229	-.978	-1.299		-.096	-.333	-1.101	-.879	
.150	-.016	-.256	-.489	-1.105		-.319	-.405	-.662	-.857	
.250	-.104	-.279	-.444	-.852		-.139	-.219	-.219	-.751	
.350	-.190	-.341	-.472	-.676						
.450	-.254	-.416	-.285	-.529						
.550	-.247	-.364	-.256	-.404		-.126	-.088	-.144	-.435	
.650	-.177	-.210	-.184	-.261		-.078	-.040	-.099	-.370	
.750	-.094	-.084	-.093	-.140		-.028	.011	-.063	-.304	
.850	-.021	-.009	-.015	-.051		-.002	.041	-.039	-.264	
.900	.014	.031	.024	-.008						
Right side										
.025	-1.079	-.090	.327	.532		-1.150	-.221	.153	.330	
.075	-.779	-.179	.121	.324		-1.000	-.357	-.099	.071	
.150	-.412	-.199	.028	.198		-.396	-.389	-.312	-.161	
.250	-.436	-.261	-.075	.070						
.350	-.447	-.324	-.168	-.034						
.450	-.274	-.383	-.240	-.118						
.550	-.255	-.340	-.231	-.163						
.650	-.170	-.225	-.152	-.140		-.112	-.084	-.127	-.149	
.750	-.086	-.076	-.088	-.090		-.086	-.055	-.091	-.151	
.850	.019	.020	.010	-.008		-.006	.028	-.011	-.091	
.900	.054	.053	.038	.012		.008	.046	.001	-.091	

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(j) $\alpha = 9.7^\circ$; M = 0.92.

	C_p for-					C_p for-				
$\frac{x}{c_v}$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.322	.889	.430			.536	.714	.577		
.025	-.147	-.115	.006			.261	-.141	-1.018		
.075	-.026	-.145	-.228			.094	-.194	-.479		
.150	-.043	-.210	-.386			.010	-.198	-.465		
.250	-.055	-.156	-.287			-.055	-.208	-.347		
.350	-.087	-.163	-.255			-.124	-.252	-.366		
.450	-.121	-.204	-.271			-.198	-.307	-.382		
.550	-.135	-.236	-.252			-.221	-.307	-.265		
.650	-.134	-.207	-.199			-.215	-.271	-.211		
.750	-.131	-.199	-.185			-.129	-.136	-.129		
.850	-.071	-.088	-.092			-.028	-.022	-.020		
.900	-.023	-.026	-.030			.024	.037	.038		
Right side										
.025	.093	-.003	-.086			-.818	-.057	.335		
.075	-.232	-.146	-.007			-.534	-.142	.136		
.150	-.360	-.170	-.003			-.455	-.165	.042		
.250	-.301	-.145	-.033			-.339	-.188	-.026		
.350	-.278	-.170	-.065			-.363	-.229	-.085		
.450	-.279	-.194	-.092			-.375	-.271	-.145		
.550	-.256	-.202	-.100			-.324	-.282	-.171		
.650	-.201	-.175	-.088			-.201	-.239	-.140		
.750	-.182	-.170	-.081			-.109	-.105	-.078		
.850	-.079	-.070	-.022			-.006	.000	.009		
.900	-.028	-.028	-.001			.032	.028	.025		
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.418	.660	.444			.385	.581	.284		
.025	.265	-.190	-1.177			.098	-.385	-1.249		
.075	.098	-.215	-.975			-.084	-.325	-1.097		
.150	-.010	-.240	-.450			-.328	-.381	-.568		
.250	-.098	-.262	-.438			-.225	-.177	-.185		
.350	-.192	-.329	-.451							
.450	-.282	-.393	-.255							
.550	-.287	-.324	-.255							
.650	-.188	-.182	-.176			-.117	-.080	-.134		
.750	-.092	-.079	-.086			-.071	-.029	-.093		
.850	-.015	-.001	-.005			-.024	.020	-.052		
.900	.019	.039	.035			.006	.050	-.025		
Right side										
.025	-1.054	-.073	.340			-1.125	-.207	.161		
.075	-.816	-.165	.134			-1.011	-.351	-.092		
.150	-.422	-.183	.038			-.512	-.364	-.306		
.250	-.426	-.246	-.066							
.350	-.468	-.310	-.156							
.450	-.373	-.352	-.225							
.550	-.227	-.299	-.211							
.650	-.162	-.186	-.149			-.105	-.047	-.110		
.750	-.079	-.071	-.078			-.081	-.046	-.085		
.850	.026	.029	.019			-.007	.039	.001		
.900	.063	.062	.051			.013	.058	.014		

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(k) $\alpha = 15.6°$; $M = 0.60$.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.790	.804	.825	.164	-.446	.238	.673	.281	-.667	-1.469
.025	-.300	-.169	-.186	-.409	-1.144	.302	-.133	-.697	-1.522	-1.949
.075	-.146	-.169	-.277	-.391	-.728	.114	-.172	-.487	-1.013	-2.078
.150	-.117	-.194	-.313	-.402	-.563	.021	-.181	-.372	-.520	-1.408
.250	-.083	-.147	-.238	-.300	-.394	-.054	-.192	-.320	-.425	-.527
.350	-.097	-.147	-.218	-.268	-.338	-.106	-.210	-.308	-.391	-.516
.450	-.126	-.176	-.227	-.259	-.320	-.146	-.235	-.297	-.345	-.471
.550	-.144	-.183	-.218	-.241	-.304	-.149	-.212	-.250	-.291	-.399
.650	-.137	-.156	-.186	-.194	-.265	-.133	-.181	-.207	-.239	-.326
.750	-.126	-.145	-.159	-.162	-.225	-.101	-.129	-.141	-.167	-.243
.850	-.083	-.089	-.096	-.097	-.155	-.045	-.062	-.067	-.101	-.157
.900	-.054	-.048	-.055	-.060	-.098	-.178	-.190	-.184	-.183	-.162
Right side										
.025	-.081	-.084	-.202	.241	.563	-.678	-.113	.337	.535	.694
.075	-.282	-.176	-.166	.105	.299	-.468	-.165	.125	.299	.473
.150	-.291	-.174	-.114	.051	.159	-.368	-.178	.021	.159	.297
.250	-.235	-.149	-.087	.023	.084	-.309	-.185	-.042	.066	.166
.350	-.226	-.151	-.105	-.017	.021	-.287	-.201	-.094	-.017	.066
.450	-.226	-.176	-.130	-.060	-.031	-.275	-.219	-.132	-.085	-.015
.550	-.203	-.169	-.137	-.083	-.055	-.246	-.210	-.146	-.115	-.062
.650	-.169	-.147	-.123	-.087	-.062	-.189	-.169	-.123	-.106	-.076
.750	-.142	-.127	-.114	-.087	-.071	-.121	-.111	-.083	-.083	-.067
.850	-.078	-.066	-.071	-.047	-.046	-.045	-.037	-.017	-.026	-.040
.900	-.045	-.044	-.049	-.029	-.028	-.022	-.019	-.015	-.029	-.044
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	-.013	.608	-.010	-.814	-1.304	.103	.536	-.130	-.601	-.886
.025	.317	-.160	-1.069	-1.354	-1.134	.168	-.237	-.979	-.780	-.748
.075	.134	-.187	-.577	-1.334	-1.141	-.029	-.226	-.493	-.764	-.751
.150	.014	-.205	-.432	-1.253	-1.116	-.124	-.203	-.326	-.712	-.742
.250	.431		-.042	-.490	-.981	-.101	-.129	-.238	-.628	-.703
.350	-.121	-.226	-.317	-.373	-1.017					
.450	-.158	-.237	-.286	-.282	-.873					
.550	-.173	-.219	-.250	-.253	-.701					
.650	-.142	-.174	-.186	-.196	-.523	-.133	-.111	-.159	-.420	-.588
.750	-.097	-.104	-.123	-.128	-.365	-.103	-.077	-.132	-.364	-.561
.850	-.060	-.062	-.065	-.069	-.250	-.074	-.044	-.101	-.309	-.523
.900	-.042	-.030	-.042	-.042	-.202	-.047	-.019	-.076	-.275	-.498
Right side										
.025	-.990	-.124	.355	.548	.665	-.827	-.201	.188	.313	.387
.075	-.565	-.181	.127	.336	.509	-.495	-.241	-.051	.062	.200
.150	-.397	-.183	.026	.184	.362	-.327	-.210	-.139	-.078	.021
.250	-.352	-.205	-.062	.069	.220					
.350	-.314	-.221	-.121	-.022	.112					
.450	-.280	-.226	-.150	-.078	.033					
.550	-.235	-.203	-.155	-.101	-.010					
.650	-.178	-.165	-.130	-.097	-.033	-.135	-.115	-.119	-.126	-.119
.750	-.117	-.100	-.092	-.074	-.040	-.126	-.089	-.114	-.137	-.159
.850	-.035	-.023	-.022	-.017	-.017	-.074	-.019	-.055	-.090	-.150
.900	-.008	.004	-.001	-.002	-.019	-.051	-.010	-.042	-.087	-.180

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(l) $\alpha = 15.8°$; M = 0.80.

$\frac{x}{c_v}$	C_p for-					C_p for-				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.358	.839	.695	.547	.146	.442	.729	.511	.125	-.197
.025	-1.020	-.199	.246	-.075	-.897	.294	-.158	-1.145	-1.677	-1.658
.075	-.287	-.208	-.128	-.260	-.646	.098	-.213	-.540	-.955	-1.445
.150	-.138	-.252	-.355	-.453	-.595	.004	-.217	-.428	-.676	-1.191
.250	-.101	-.199	-.283	-.348	-.430	-.070	-.231	-.373	-.473	-.724
.350	-.107	-.184	-.257	-.305	-.367	-.144	-.261	-.361	-.447	-.545
.450	-.145	-.208	-.261	-.305	-.348	-.200	-.293	-.346	-.402	-.522
.550	-.181	-.228	-.255	-.293	-.334	-.201	-.268	-.299	-.348	-.486
.650	-.188	-.204	-.222	-.263	-.331	-.192	-.235	-.246	-.292	-.445
.750	-.200	-.199	-.210	-.230	-.310	-.148	-.167	-.170	-.206	-.360
.850	-.151	-.128	-.137	-.147	-.214	-.077	-.077	-.081	-.116	-.251
.900	-.106	-.072	-.086	-.095	-.155	-.171	-.185	-.170	-.152	-.138
Right side										
.025	.343	-.095	-.957	-.111	.339	-1.118	-.134	.333	.500	.624
.075	-.124	-.222	-.267	-.092	.107	-.520	-.201	.104	.253	.371
.150	-.325	-.232	-.125	-.051	.018	-.420	-.213	.004	.122	.214
.250	-.278	-.196	-.089	-.045	-.004	-.360	-.220	-.061	.030	.112
.350	-.269	-.188	-.105	-.060	-.019	-.342	-.244	-.119	-.046	.030
.450	-.262	-.214	-.149	-.087	-.040	-.323	-.268	-.176	-.122	-.046
.550	-.233	-.207	-.169	-.104	-.057	-.287	-.258	-.195	-.156	-.084
.650	-.203	-.182	-.175	-.110	-.064	-.225	-.211	-.176	-.147	-.096
.750	-.192	-.182	-.183	-.123	-.086	-.145	-.143	-.125	-.119	-.092
.850	-.112	-.099	-.125	-.081	-.064	-.053	-.054	-.052	-.063	-.069
.900	-.073	-.062	-.093	-.061	-.058	-.021	-.027	-.034	-.060	-.090

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.251	.649	.286	-.233	-.744	.223	.557	.081	-.439	-.933
.025	.319	-.178	-1.343	-1.486	-.962	.155	-.306	-1.170	-.802	-.697
.075	.131	-.214	-.742	-1.396	-.955	-.056	-.294	-.767	-.798	-.694
.150	.007	-.243	-.514	-1.146	-.912	-.177	-.249	-.411	-.769	-.700
.250	-.218	-.219	-.275	-.590	-.843	-.117	-.146	-.289	-.695	-.697
.350	-.156	-.277	-.360	-.628	-.852					
.450	-.209	-.290	-.336	-.411	-.792					
.550	-.224	-.274	-.292	-.314	-.720					
.650	-.188	-.213	-.220	-.227	-.631	-.145	-.121	-.183	-.486	-.589
.750	-.127	-.131	-.140	-.143	-.530	-.109	-.066	-.143	-.430	-.552
.850	-.067	-.059	-.066	-.070	-.421	-.071	-.024	-.101	-.364	-.515
.900	-.035	-.026	-.031	-.039	-.370	-.042	.000	-.073	-.326	-.497
Right side										
.025	-1.429	-.143	.351	.552	.686	-1.000	-.241	.174	.341	.436
.075	-.641	-.216	.125	.332	.506	-.727	-.303	-.087	.083	.235
.150	-.467	-.210	.019	.189	.357	-.426	-.246	-.193	-.098	.039
.250	-.399	-.243	-.084	.064	.217					
.350	-.361	-.279	-.157	-.039	.103					
.450	-.320	-.282	-.204	-.110	.017					
.550	-.271	-.255	-.210	-.141	-.037					
.650	-.206	-.195	-.173	-.135	-.060	-.118	-.112	-.131	-.152	-.150
.750	-.127	-.124	-.123	-.102	-.069	-.138	-.078	-.120	-.161	-.192
.850	-.030	-.029	-.033	-.034	-.054	-.070	-.003	-.051	-.101	-.174
.900	.007	.009	-.007	-.015	-.076	-.044	.014	-.034	-.105	-.201

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE MIDWING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Concluded

(m) $\alpha = 15.9°$; M = 0.85.

$\dfrac{x}{c_v}$	C_p for -					C_p for -				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.375	.858	.699	.464		.471	.736	.548	.248	
.025	-1.031	-.225	.304	-.164		.265	-.190	-1.215	-1.482	
.075	-.496	-.239	-.066	-.289		.071	-.250	-.607	-.908	
.150	-.153	-.296	-.380	-.469		-.018	-.252	-.486	-.793	
.250	-.120	-.237	-.316	-.428		-.095	-.260	-.396	-.601	
.350	-.126	-.210	-.288	-.356		-.177	-.296	-.398	-.406	
.450	-.154	-.243	-.296	-.322		-.234	-.332	-.371	-.462	
.550	-.183	-.263	-.279	-.305		-.245	-.313	-.322	-.397	
.650	-.194	-.239	-.242	-.301		-.238	-.280	-.275	-.332	
.750	-.206	-.242	-.249	-.284		-.186	-.202	-.194	-.232	
.850	-.156	-.159	-.166	-.181		-.103	-.101	-.099	-.130	
.900	-.109	-.100	-.106	-.118		-.179	-.198	-.160	-.157	
Right side										
.025	.390	-.108	-.967	-.058		-1.127	-.146	.321	.461	
.075	-.072	-.253	-.445	-.088		-.628	-.230	.091	.226	
.150	-.370	-.271	-.127	-.071		-.501	-.239	-.010	.097	
.250	-.332	-.230	-.102	-.072		-.401	-.246	-.078	.007	
.350	-.313	-.216	-.116	-.089		-.386	-.277	-.142	-.074	
.450	-.296	-.239	-.145	-.116		-.357	-.308	-.208	-.153	
.550	-.259	-.239	-.163	-.129		-.319	-.301	-.231	-.186	
.650	-.230	-.213	-.171	-.133		-.259	-.256	-.211	-.180	
.750	-.234	-.223	-.180	-.154		-.174	-.175	-.154	-.145	
.850	-.146	-.132	-.123	-.108		-.078	-.074	-.072	-.084	
.900	-.108	-.091	-.092	-.091		-.040	-.048	-.049	-.074	

$\dfrac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.304	.649	.349	-.074		.247	.551	.131	-.376	
.025	.294	-.205	-1.379	-1.613		.124	-.361	-1.329	-.952	
.075	.108	-.244	-1.145	-1.377		-.091	-.378	-1.020	-.929	
.150	-.021	-.273	-.529	-1.130		-.274	-.288	-.557	-.865	
.250	-.569	-.519	-.489	-.664		-.160	-.173	-.262	-.801	
.350	-.197	-.325	-.363	-.702						
.450	-.261	-.350	-.362	-.478						
.550	-.279	-.327	-.323	-.354						
.650	-.231	-.243	-.240	-.254		-.170	-.139	-.194	-.478	
.750	-.156	-.156	-.153	-.162		-.125	-.088	-.153	-.406	
.850	-.084	-.075	-.074	-.079		-.081	-.037	-.109	-.333	
.900	-.054	-.043	-.035	-.042		-.055	-.009	-.085	-.296	
Right side										
.025	-1.374	-.158	.332	.525		-1.413	-.266	.153	.328	
.075	-1.118	-.236	.114	.302		-1.021	-.375	-.122	.061	
.150	-.482	-.234	.003	.163		-.529	-.291	-.285	-.147	
.250	-.468	-.281	-.101	.036						
.350	-.379	-.324	-.191	-.072						
.450	-.354	-.332	-.245	-.152						
.550	-.310	-.301	-.252	-.186						
.650	-.232	-.230	-.208	-.170		-.184	-.142	-.153	-.180	
.750	-.146	-.146	-.146	-.132		-.154	-.097	-.135	-.176	
.850	-.044	-.040	-.048	-.052		-.081	-.014	-.054	-.108	
.900	-.006	-.006	-.021	-.031		-.058	.000	-.041	-.110	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL

(a) $\alpha = 0°$; M = 0.60.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.750	.643	.601	-.006	-.868	.403	.734	.497	-.247	-1.071
.025	.209	-.072	-.437	-.809	-1.847	.285	-.072	-.503	-1.156	-1.326
.075	.089	-.087	-.281	-.505	-.788	.116	-.115	-.362	-.829	-1.282
.150	.012	-.119	-.240	-.376	-.553	.039	-.126	-.272	-.476	-1.285
.250	.003	-.085	-.158	-.240	-.362	-.019	-.137	-.240	-.351	-1.030
.350	-.022	-.097	-.140	-.206	-.298	-.065	-.153	-.235	-.303	-.558
.450	-.053	-.106	-.140	-.183	-.260	-.105	-.169	-.219	-.263	-.317
.550	-.058	-.097	-.117	-.151	-.219	-.085	-.133	-.167	-.199	-.239
.650	-.022	-.058	-.081	-.115	-.194	-.065	-.097	-.117	-.138	-.198
.750	-.004	-.031	-.045	-.077	-.146	-.024	-.042	-.056	-.072	-.141
.850	.030	.012	.010	-.022	-.073	.030	.017	.014	.007	-.052
.900	.053	.048	.044	.019	-.027	.060	.055	.053	.048	-.002
	Right side									
.025	-.347	-.026	.252	.474	.706	-.627	-.063	.313	.565	.740
.075	-.297	-.101	.080	.261	.467	-.349	-.119	.121	.323	.504
.150	-.229	-.103	.019	.162	.330	-.279	-.124	.035	.189	.351
.250	-.159	-.083	.001	.114	.248	-.227	-.133	-.018	.100	.232
.350	-.148	-.097	-.029	.057	.164	-.213	-.146	-.058	.044	.150
.450	-.139	-.106	-.061	.016	.109	-.193	-.149	-.083	-.006	.087
.550	-.105	-.087	-.049	.014	.091	-.162	-.133	-.077	-.020	.055
.650	-.067	-.045	-.018	.039	.105	-.107	-.085	-.058	-.004	.050
.750	-.031	-.017	.012	.053	.105	-.040	-.031	-.018	.030	.062
.850	.026	.032	.057	.089	.121	.039	.041	.071	.073	.087
.900	.053	.055	.064	.089	.109	.062	.060	.062	.073	.080
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.297	.718	.386	-.235	-.658	.414	.677	.316	-.163	-.437
.025	.310	-.106	-.666	-.993	-.840	.188	-.192	-.605	-.793	-.567
.075	.134	-.135	-.430	-.886	-.815	.019	-.160	-.369	-.770	-.556
.150	.033	-.155	-.335	-.702	-.786	-.053	-.142	-.258	-.655	-.594
.250	-.033	-.158	-.267	-.448	-.731	-.033	-.090	-.181	-.458	-.563
.350	-.083	-.169	-.240	-.328	-.679					
.450	-.110	-.178	-.222	-.267	-.631					
.550	-.112	-.151	-.185	-.208	-.563					
.650	-.085	-.110	-.117	-.149	-.481	-.076	-.076	-.102	-.267	-.401
.750	-.042	-.045	-.058	-.083	-.394	-.055	-.031	-.067	-.253	-.369
.850	.021	.008	-.002	-.020	-.303	-.028	-.004	-.049	-.240	-.335
.900	.035	.030	.030	.007	-.260	-.010	.019	-.031	-.206	-.328
	Right side									
.025	-.631	-.083	.338	.565	.688	-.532	-.142	.209	.377	.463
.075	-.435	-.137	.123	.327	.485	-.365	-.176	-.002	.125	.219
.150	-.306	-.137	.032	.196	.328	-.252	-.146	-.065	-.006	.071
.250	-.263	-.160	-.033	.084	.203					
.350	-.241	-.164	-.081	.014	.105					
.450	-.209	-.167	-.106	-.027	.041					
.550	-.164	-.144	-.099	-.047	.005					
.650	-.112	-.103	-.074	-.029	.002	-.080	-.069	-.067	-.072	-.075
.750	-.046	-.035	-.033	-.002	.005	-.060	-.038	-.070	-.070	-.102
.850	.028	.032	.053	.055	.018	-.015	.014	-.011	-.020	-.086
.900	.057	.055	.053	.066	-.004	-.006	.030	.003	-.020	-.118

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(b) $\alpha = 0°$; M = 0.80.

	C_p for – $z/b_v = 0.11$					C_p for – $z/b_v = 0.38$				
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.831	.655	.709	.411	-.077	.539	.772	.633	.225	-.345
.025	.222	-.045	-.419	-.741	-1.500	.304	-.052	-.577	-1.166	-1.228
.075	.107	-.076	-.285	-.540	-1.061	.130	-.114	-.377	-.883	-1.181
.150	.025	-.116	-.253	-.423	-.633	.046	-.131	-.291	-.587	-1.039
.250	.007	-.087	-.167	-.263	-.402	-.018	-.146	-.256	-.384	-.878
.350	-.024	-.101	-.148	-.213	-.323	-.075	-.167	-.248	-.318	-.708
.450	-.065	-.117	-.152	-.195	-.281	-.124	-.198	-.234	-.271	-.519
.550	-.069	-.108	-.131	-.161	-.240	-.109	-.152	-.180	-.202	-.377
.650	-.030	-.067	-.088	-.123	-.205	-.075	-.111	-.120	-.135	-.275
.750	-.009	-.036	-.046	-.082	-.161	-.028	-.045	-.049	-.061	-.179
.850	.037	.020	.018	-.017	-.083	.042	.034	.030	.023	-.076
.900	.058	.055	.050	.026	-.035	.076	.075	.078	.073	-.023
Right side										
.025	-.320	-.004	.271	.478	.683	-.717	-.061	.323	.557	.731
.075	-.306	-.090	.090	.269	.464	-.377	-.123	.129	.323	.509
.150	-.245	-.110	.024	.167	.339	-.295	-.133	.038	.195	.360
.250	-.171	-.092	.002	.116	.256	-.250	-.143	-.018	.105	.242
.350	-.159	-.107	-.038	.058	.170	-.227	-.157	-.067	.038	.154
.450	-.148	-.114	-.067	.015	.108	-.215	-.166	-.097	-.021	.084
.550	-.113	-.090	-.058	.011	.091	-.168	-.142	-.093	-.029	.046
.650	-.066	-.051	-.020	.037	.107	-.104	-.093	-.062	-.008	.043
.750	-.034	-.023	.008	.053	.105	-.027	-.020	-.006	.026	.050
.850	.037	.039	.056	.090	.120	.057	.058	.067	.079	.078
.900	.060	.066	.069	.097	.107	.081	.072	.078	.085	.063

	$z/b_v = 0.66$					$z/b_v = 0.93$				
$\frac{x}{c_v}$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.474	.749	.553	.122	-.320	.498	.707	.451	.029	-.282
.025	.324	-.087	-.935	-1.125	-.817	.180	-.210	-1.027	-.851	-.547
.075	.143	-.134	-.449	-.985	-.810	.004	-.186	-.399	-.819	-.568
.150	.035	-.167	-.362	-.785	-.788	-.072	-.154	-.269	-.728	-.611
.250	-.042	-.176	-.285	-.519	-.732	-.036	-.087	-.190	-.531	-.550
.350	-.095	-.190	-.253	-.361	-.659					
.450	-.135	-.196	-.233	-.268	-.588					
.550	-.128	-.167	-.183	-.201	-.516					
.650	-.092	-.122	-.119	-.132	-.448	-.071	-.061	-.090	-.294	-.407
.750	-.045	-.022	-.043	-.058	-.374	-.036	-.013	-.049	-.268	-.371
.850	.042	.028	.026	.006	-.299	-.007	.027	-.021	-.230	-.337
.900	.057	.049	.061	.037	-.264	.016	.051	-.003	-.196	-.322
Right side										
.025	-.949	-.073	.347	.572	.708	-1.028	-.166	.204	.382	.496
.075	-.459	-.143	.135	.333	.499	-.421	-.208	-.020	.110	.242
.150	-.332	-.134	.043	.199	.348	-.260	-.167	-.075	-.030	.076
.250	-.286	-.164	-.043	.084	.219					
.350	-.260	-.195	-.096	.002	.111					
.450	-.226	-.183	-.125	-.049	.035					
.550	-.172	-.152	-.119	-.061	-.004					
.650	-.104	-.098	-.081	-.043	-.010	-.059	-.058	-.062	-.067	-.079
.750	-.027	-.031	-.037	-.003	-.013	-.054	-.020	-.046	-.052	-.111
.850	.054	.058	.069	.059	.005	.013	.046	.023	.000	-.089
.900	.087	.086	.084	.076	-.018	.019	.061	.034	.002	-.115

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(c) $\alpha = 0°$; M = 0.85.

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v=0.11$					$z/b_v=0.38$				
	Left side									
.000	.852	.657	.744	.509		.582	.785	.674	.338	-.163
.025	.224	-.038	-.406	-.659	-1.334	.300	-.051	-.729	-1.439	-1.148
.075	.109	-.074	-.285	-.606	-.984	.126	-.119	-.380	-1.251	-1.136
.150	.016	-.126	-.259	-.476	-.683	.044	-.137	-.303	-.421	-.999
.250	-.002	-.098	-.180	-.286	-.450	-.029	-.159	-.266	-.370	-.840
.350	-.036	-.109	-.163	-.234	-.362	-.090	-.186	-.266	-.310	-.690
.450	-.078	-.132	-.163	-.203	-.315	-.148	-.217	-.249	-.274	-.574
.550	-.085	-.126	-.141	-.169	-.260	-.125	-.177	-.191	-.206	-.452
.650	-.044	-.081	-.100	-.129	-.219	-.091	-.127	-.133	-.143	-.341
.750	-.020	-.047	-.052	-.093	-.173	-.040	-.057	-.054	-.063	-.223
.850	.030	.014	.015	-.022	-.094	.040	.036	.034	.025	-.104
.900	.052	.051	.054	.024	-.040	.078	.080	.079	.075	-.041
	Right side									
.025	-.308	.000	.273	.478	.682	-.766	-.069	.315	.549	.734
.075	-.321	-.096	.086	.266	.467	-.418	-.132	.123	.317	.510
.150	-.260	-.119	.020	.169	.339	-.314	-.143	.032	.196	.362
.250	-.188	-.106	-.002	.116	.260	-.263	-.156	-.026	.100	.247
.350	-.176	-.122	-.046	.052	.173	-.250	-.180	-.081	.029	.165
.450	-.161	-.133	-.079	.002	.108	-.232	-.188	-.118	-.029	.078
.550	-.122	-.106	-.072	-.004	.087	-.183	-.167	-.114	-.042	.043
.650	-.078	-.065	-.025	.032	.102	-.112	-.109	-.079	-.019	.037
.750	-.041	-.030	-.006	.053	.102	-.033	-.025	-.022	.024	.048
.850	.028	.034	.048	.088	.112	.055	.057	.068	.081	.072
.900	.062	.064	.058	.095	.102	.085	.077	.076	.089	.055
	$z/b_v=0.66$					$z/b_v=0.93$				
	Left side									
.000	.513	.759	.598	.234	-.198	.520	.705	.486	.092	-.250
.025	.321	-.089	-1.020	-1.569	-.803	.162	-.234	-1.124	-.966	-.590
.075	.141	-.143	-.443	-1.388	-.811	-.027	-.225	-.419	-.937	-.635
.150	.027	-.177	-.389	-1.196	-.815	-.097	-.179	-.268	-.817	-.656
.250	-.058	-.197	-.299	-.286	-.753	-.054	-.098	-.198	-.580	-.571
.350	-.118	-.214	-.269	-.297	-.669					
.450	-.158	-.220	-.249	-.260	-.580					
.550	-.149	-.183	-.198	-.194	-.500					
.650	-.104	-.137	-.121	-.126	-.429	-.080	-.074	-.091	-.317	-.425
.750	-.054	-.020	-.039	-.051	-.352	-.043	-.024	-.046	-.283	-.389
.850	.035	.029	.029	.018	-.279	-.006	.024	-.017	-.243	-.350
.900	.054	.054	.062	.053	-.243	.020	.046	.000	-.204	-.331
	Right side									
.025	-1.061	-.079	.341	.566	.714	-1.118	-.191	.190	.380	.504
.075	-.435	-.159	.125	.330	.504	-.511	-.242	-.044	.095	.246
.150	-.358	-.153	.039	.197	.354	-.242	-.181	-.101	-.062	.069
.250	-.306	-.193	-.057	.078	.220					
.350	-.273	-.217	-.118	-.012	.111					
.450	-.237	-.208	-.148	-.063	.035					
.550	-.186	-.177	-.141	-.079	-.013					
.650	-.108	-.109	-.096	-.051	-.020	-.075	-.064	-.067	-.066	-.088
.750	-.033	-.035	-.047	-.009	-.024	-.063	-.028	-.049	-.055	-.117
.850	.058	.055	.074	.066	-.003	.003	.051	.027	.006	-.091
.900	.094	.088	.086	.079	-.027	.018	.067	.039	-.001	-.119

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(d) $\alpha = 0°$; M = 0.90.

$\frac{x}{c_v}$	C_p for- $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for- $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.879	.659	.776	.581	.180	.625	.794	.714	.420	.027
.025	.232	-.031	-.387	-.624	-1.153	.295	-.052	-.811	-1.281	-1.106
.075	.113	-.076	-.279	-.626	-.879	.122	-.124	-.371	-1.133	-1.036
.150	.013	-.129	-.266	-.519	-.705	.034	-.148	-.322	-.794	-.878
.250	-.010	-.105	-.192	-.338	-.483	-.038	-.171	-.290	-.431	-.784
.350	-.049	-.124	-.182	-.272	-.358	-.104	-.207	-.306	-.379	-.729
.450	-.097	-.156	-.186	-.240	-.404	-.174	-.256	-.272	-.305	-.625
.550	-.109	-.147	-.153	-.185	-.331	-.156	-.207	-.203	-.204	-.521
.650	-.054	-.100	-.105	-.141	-.262	-.113	-.144	-.136	-.144	-.424
.750	-.033	-.060	-.057	-.100	-.222	-.052	-.064	-.053	-.073	-.310
.850	.019	.010	.016	-.031	-.140	.032	.031	.039	.019	-.195
.900	.048	.048	.060	.015	-.078	.076	.079	.092	.059	-.121
Right side										
.025	-.283	.007	.281	.474	.674	-.767	-.068	.309	.536	.735
.075	-.315	-.096	.091	.259	.467	-.452	-.137	.122	.303	.509
.150	-.280	-.124	.020	.159	.346	-.338	-.153	.030	.180	.364
.250	-.207	-.113	-.005	.100	.265	-.283	-.171	-.035	.085	.245
.350	-.198	-.135	-.057	.037	.176	-.289	-.204	-.096	.008	.156
.450	-.185	-.153	-.093	-.012	.101	-.252	-.219	-.141	-.053	.071
.550	-.138	-.125	-.087	-.010	.079	-.194	-.188	-.136	-.064	.028
.650	-.087	-.075	-.040	.019	.096	-.123	-.124	-.099	-.039	.020
.750	-.046	-.041	-.013	.031	.087	-.033	-.031	-.031	.002	.025
.850	.030	.028	.046	.075	.092	.057	.056	.072	.067	.041
.900	.061	.059	.059	.068	.079	.088	.079	.080	.065	.017
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.562	.770	.649	.320	-.033	.559	.713	.536	.175	-.142
.025	.315	-.085	-.976	-1.402	-.800	.148	-.263	-1.077	-1.329	-.560
.075	.140	-.151	-.605	-1.259	-.819	-.057	-.311	-.926	-1.232	-.592
.150	.018	-.194	-.418	-1.148	-.828	-.167	-.204	-.218	-.958	-.620
.250	-.073	-.228	-.389	-.785	-.761	-.066	-.112	-.170	-.635	-.574
.350	-.151	-.248	-.251	-.289	-.665					
.450	-.194	-.254	-.263	-.204	-.590					
.550	-.182	-.218	-.208	-.178	-.536					
.650	-.128	-.164	-.125	-.126	-.460	-.088	-.077	-.096	-.284	-.455
.750	-.068	-.014	-.037	-.057	-.386	-.041	-.020	-.051	-.273	-.420
.850	.037	.034	.040	.008	-.311	.002	.027	-.018	-.243	-.383
.900	.057	.059	.075	.035	-.281	.029	.054	.003	-.208	-.367
Right side										
.025	-1.002	-.084	.330	.551	.714	-1.089	-.222	.170	.356	.509
.075	-.693	-.170	.118	.306	.501	-.924	-.335	-.092	.063	.249
.150	-.383	-.167	.031	.178	.354	-.253	-.208	-.168	-.113	.057
.250	-.389	-.216	-.075	.057	.217					
.350	-.258	-.255	-.148	-.045	.100					
.450	-.250	-.244	-.183	-.112	.012					
.550	-.199	-.202	-.167	-.124	-.038					
.650	-.111	-.120	-.115	-.084	-.048	-.081	-.072	-.076	-.108	-.121
.750	-.029	-.031	-.053	-.031	-.050	-.060	-.029	-.048	-.078	-.146
.850	.066	.059	.075	.052	-.023	.013	.051	.032	-.003	-.108
.900	.104	.097	.091	.063	-.053	.019	.071	.046	-.016	-.132

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(e) α = 0°; M = 0.92.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.889	.653	.785	.640	.204	.639	.799	.736	.492	.061
.025	.237	-.023	-.374	-.564	-1.117	.298	-.044	-.764	-1.182	-1.186
.075	.117	-.070	-.280	-.584	-.856	.124	-.123	-.374	-1.061	-1.049
.150	.012	-.135	-.283	-.505	-.734	.038	-.151	-.341	-.933	-.857
.250	-.008	-.112	-.200	-.393	-.486	-.036	-.174	-.305	-.512	-.776
.350	-.049	-.132	-.198	-.294	-.340	-.109	-.218	-.333	-.440	-.726
.450	-.104	-.170	-.212	-.280	-.410	-.188	-.281	-.344	-.420	-.627
.550	-.119	-.166	-.179	-.216	-.368	-.173	-.231	-.202	-.190	-.528
.650	-.065	-.109	-.115	-.135	-.268	-.125	-.156	-.139	-.101	-.437
.750	-.037	-.066	-.066	-.085	-.244	-.057	-.056	-.058	-.039	-.329
.850	.016	.008	.010	-.010	-.163	.034	.038	.040	.049	-.218
.900	.047	.052	.050	.037	-.096	.078	.082	.090	.092	-.148
Right side										
.025	-.270	.013	.274	.487	.677	-.760	-.065	.301	.543	.739
.075	-.322	-.096	.087	.271	.470	-.494	-.143	.108	.314	.516
.150	-.290	-.123	.015	.170	.352	-.352	-.157	.023	.191	.367
.250	-.217	-.117	-.015	.109	.269	-.299	-.178	-.048	.093	.249
.350	-.202	-.141	-.066	.039	.175	-.306	-.213	-.113	.015	.161
.450	-.191	-.165	-.110	-.014	.097	-.286	-.247	-.168	-.060	.060
.550	-.141	-.138	-.104	-.021	.077	-.191	-.204	-.164	-.080	.026
.650	-.086	-.080	-.053	.020	.094	-.119	-.128	-.121	-.051	.016
.750	-.044	-.045	-.027	.034	.085	-.026	-.028	-.040	.008	.018
.850	.030	.030	.033	.075	.084	.063	.060	.065	.075	.030
.900	.057	.063	.049	.079	.065	.092	.084	.073	.088	.006
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.576	.773	.673	.395	-.004	.566	.714	.561	.253	-.114
.025	.317	-.085	-.942	-1.289	-.813	.137	-.265	-.997	-1.314	-.547
.075	.136	-.151	-.702	-1.156	-.831	-.059	-.297	-.917	-1.228	-.570
.150	.014	-.193	-.414	-1.090	-.839	-.208	-.244	-.457	-1.059	-.599
.250	-.079	-.239	-.401	-1.014	-.776	-.096	-.114	-.147	-.934	-.566
.350	-.166	-.292	-.422	-.623	-.675					
.450	-.219	-.283	-.229	-.175	-.596					
.550	-.198	-.239	-.215	-.088	-.553					
.650	-.141	-.177	-.132	-.063	-.493	-.090	-.067	-.100	-.199	-.463
.750	-.059	-.015	-.040	-.013	-.417	-.038	-.015	-.053	-.209	-.430
.850	.038	.037	.039	.045	-.337	.003	.037	-.023	-.201	-.391
.900	.057	.068	.075	.072	-.306	.025	.058	-.001	-.171	-.375
Right side										
.025	-.971	-.078	.323	.547	.717	-1.044	-.222	.156	.360	.511
.075	-.726	-.171	.105	.312	.506	-.943	-.374	-.110	.072	.251
.150	-.377	-.169	.018	.189	.358	-.410	-.247	-.234	-.119	.059
.250	-.398	-.232	-.089	.059	.221					
.350	-.354	-.308	-.179	-.046	.101					
.450	-.235	-.265	-.225	-.129	.010					
.550	-.194	-.216	-.191	-.144	-.045					
.650	-.112	-.122	-.136	-.090	-.055	-.090	-.065	-.079	-.078	-.121
.750	-.030	-.033	-.050	-.025	-.060	-.065	-.026	-.048	-.057	-.157
.850	.070	.068	.073	.066	-.032	.009	.056	.035	.017	-.113
.900	.104	.097	.088	.084	-.057	.017	.076	.043	.016	-.143

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(f) $\alpha = 9.4°$; $M = 0.60$.

$\frac{x}{c_v}$	\multicolumn{5}{c}{C_p for— $z/b_v = 0.11$}	\multicolumn{5}{c}{C_p for— $z/b_v = 0.38$}
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side
.000	.567	.805	.557	.174	-.502	.395	.667	.387	-.398	-1.188
.025	.028	-.123	-.218	-.396	-.909	.227	-.110	-.513	-1.180	-1.641
.075	.014	-.112	-.216	-.323	-.591	.082	-.137	-.366	-.686	-1.618
.150	-.015	-.132	-.218	-.295	-.477	.010	-.137	-.279	-.421	-1.195
.250	-.011	-.089	-.157	-.219	-.338	-.047	-.148	-.247	-.346	-.307
.350	-.038	-.103	-.143	-.198	-.293	-.090	-.180	-.245	-.318	-.363
.450	-.072	-.119	-.157	-.203	-.272	-.126	-.193	-.236	-.281	-.357
.550	-.081	-.123	-.161	-.194	-.252	-.115	-.162	-.193	-.231	-.300
.650	-.063	-.107	-.134	-.169	-.213	-.099	-.130	-.152	-.178	-.247
.750	-.054	-.089	-.104	-.136	-.182	-.067	-.089	-.093	-.116	-.179
.850	-.022	-.040	-.052	-.074	-.116	-.002	-.017	-.018	-.035	-.088
.900	.003	-.004	-.009	-.033	-.070	.032	.021	.023	.008	-.041
Right side
.025	-.131	-.022	.068	.248	.571	-.523	-.051	.300	.510	.696
.075	-.208	-.105	.023	.176	.359	-.326	-.107	.109	.287	.478
.150	-.192	-.103	.012	.133	.250	-.265	-.121	.030	.162	.328
.250	-.149	-.078	.005	.098	.182	-.231	-.132	-.025	.082	.207
.350	-.149	-.096	-.027	.043	.100	-.222	-.150	-.068	.022	.123
.450	-.158	-.114	-.052	.004	.041	-.219	-.164	-.095	-.031	.053
.550	-.145	-.110	-.057	-.005	.023	-.192	-.153	-.102	-.051	.014
.650	-.122	-.092	-.041	-.008	.023	-.138	-.119	-.082	-.042	.005
.750	-.097	-.076	-.034	-.003	.021	-.077	-.062	-.043	-.010	.014
.850	-.036	-.017	.009	.031	.043	-.002	-.001	.028	.034	.053
.900	-.004	.008	.028	.043	.048	.028	.019	.030	.036	.039
\multicolumn{6}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
Left side
.000	.248	.626	.218	-.424	-1.109	.295	.572	.091	-.451	-.813
.025	.254	-.144	-.716	-1.026	-1.195	.125	-.241	-.707	-.849	-.670
.075	.096	-.162	-.450	-.904	-1.172	-.027	-.193	-.397	-.785	-.670
.150	-.002	-.171	-.347	-.762	-1.120	-.104	-.175	-.275	-.651	-.650
.250	-.061	-.175	-.284	-.511	-1.013	-.077	-.119	-.188	-.562	-.607
.350	-.111	-.189	-.259	-.346	-.872					
.450	-.136	-.191	-.234	-.267	-.693					
.550	-.140	-.175	-.200	-.221	-.525					
.650	-.111	-.134	-.141	-.164	-.379	-.102	-.083	-.107	-.290	-.484
.750	-.065	-.074	-.077	-.100	-.252	-.074	-.046	-.070	-.240	-.463
.850	-.013	-.024	-.029	-.047	-.152	-.038	-.008	.057	-.196	-.434
.900	.010	.005	.005	-.012	-.111	-.013	.012	-.036	-.162	-.411
Right side
.025	-.585	-.074	.323	.523	.684	-.560	-.155	.178	.328	.439
.075	-.417	-.134	.114	.305	.500	-.367	-.193	-.029	.077	.221
.150	-.303	-.144	.028	.174	.350	-.258	-.171	-.104	-.044	.055
.250	-.272	-.166	-.052	.071	.218					
.350	-.260	-.175	-.093	.002	.123					
.450	-.231	-.182	-.116	-.049	.050					
.550	-.188	-.162	-.120	-.067	.018					
.650	-.133	-.130	-.097	-.049	.005	-.090	-.074	-.088	-.102	-.086
.750	-.074	-.058	-.066	-.026	.009	-.070	-.053	-.079	-.095	-.118
.850	.005	.014	.023	.025	.032	-.015	.010	-.018	-.035	-.102
.900	.032	.037	.032	.041	.034	-.006	.026	.000	-.026	-.129

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(g) $\alpha = 9.6°$; M = 0.80.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.558	.846	.598	.542	.016	.519	.704	.560	.189	-.230
.025	-.059	-.117	-.130	-.319	-.517	.222	-.102	-.491	-1.271	-1.477
.075		-.117	-.219	-.294	-.467	.073	-.143	-.380	-.707	-1.376
.150	-.024	-.143	-.251	-.326	-.487	.002	-.140	-.303	-.499	-1.080
.250	-.027	-.099	-.179	-.236	-.382	-.054	-.158	-.271	-.375	-.407
.350	-.054	-.114	-.171	-.216	-.321	-.106	-.201	-.276	-.337	-.410
.450	-.085	-.137	-.182	-.219	-.301	-.156	-.223	-.271	-.308	-.392
.550	-.102	-.155	-.189	-.212	-.280	-.147	-.198	-.230	-.252	-.348
.650	-.088	-.134	-.166	-.189	-.250	-.126	-.163	-.182	-.198	-.305
.750	-.079	-.117	-.139	-.161	-.230	-.080	-.101	-.112	-.125	-.227
.850	-.030	-.057	-.068	-.089	-.155	-.006	-.016	-.019	-.028	-.126
.900	.003	-.010	-.018	-.037	-.096	.035	.030	.032	.021	-.062
Right side										
.025	-.036	-.007	.002	.233	.301	-.550	-.054	.280	.501	.685
.075	-.211	-.111	.016	.129	.255	-.355	-.116	.098	.279	.465
.150	-.217	-.113	-.001	.093	.221	-.281	-.130	.023	.163	.322
.250	-.173	-.092	-.012	.073	.176	-.249	-.149	-.033	.083	.213
.350	-.174	-.113	-.046	.028	.110	-.244	-.172	-.081	.015	.128
.450	-.184	-.133	-.075	-.011	.055	-.247	-.196	-.122	-.045	.040
.550	-.168	-.128	-.080	-.024	.031	-.217	-.184	-.133	-.068	.000
.650	-.146	-.114	-.068	-.019	.024	-.162	-.145	-.104	-.054	-.012
.750	-.126	-.099	-.060	-.019	.018	-.085	-.081	-.051	-.019	-.003
.850	-.047	-.031	-.009	.015	.034	.006	.011	.020	.036	.025
.900	-.010	.001	.016	.030	.042	.034	.028	.034	.039	.018

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.394	.657	.422	-.016	-.487	.374	.587	.255	-.129	-.576
.025	.246	-.155	-.936	-1.201	-.939	.113	-.249	-.878	-.758	-.592
.075	.091	-.175	-.494	-1.138	-.912	-.061	-.243	-.512	-.712	-.598
.150	-.012	-.192	-.388	-.955	-.868	-.135	-.198	-.300	-.658	-.604
.250	-.077	-.201	-.315	-.592	-.822	-.086	-.123	-.216	-.565	-.587
.350	-.133	-.222	-.286	-.330	-.764					
.450	-.171	-.228	-.265	-.278	-.680					
.550	-.168	-.213	-.224	-.235	-.585					
.650	-.135	-.151	-.153	-.158	-.489	-.102	-.073	-.109	-.339	-.467
.750	-.061	-.072	-.074	-.080	-.384	-.065	-.030	-.072	-.287	-.443
.850	-.007	-.010	-.004	-.010	-.280	-.023	.013	-.042	-.227	-.403
.900	.017	.025	.029	.021	-.230	.003	.037	-.019	-.184	-.384
Right side										
.025	-.769	-.075	.312	.537	.698	-.919	-.180	.166	.336	.473
.075	-.448	-.149	.110	.311	.511	-.427	-.242	-.068	.076	.243
.150	-.331	-.155	.025	.183	.359	-.281	-.198	-.136	-.073	.059
.250	-.298	-.189	.059	.070	.224					
.350	-.287	-.208	-.121	-.011	.119					
.450	-.254	-.217	-.153	-.068	.037					
.550	-.209	-.196	-.150	-.088	-.004					
.650	-.141	-.152	-.112	-.068	-.018	-.089	-.076	-.083	-.094	-.101
.750	-.067	-.052	-.051	-.033	-.024	-.059	-.039	-.068	-.089	-.135
.850	.022	.027	.026	.039	.005	.005	.034	.002	-.028	-.114
.900	.060	.051	.052	.054	-.003	.019	.051	.016	-.027	-.138

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(h) $\alpha = 9.7^\circ$; M = 0.85.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.545	.865	.606	.464	.158	.562	.711	.606	.307	-.136
.025	-.077	-.132	-.102	-.238	-.481	.225	-.127	-.549	-1.401	-1.491
.075	-.003	-.138	-.220	-.283	-.433	.079	-.173	-.395	-.686	-1.308
.150	-.025	-.173	-.274	-.368	-.485	.003	-.173	-.325	-.543	-.958
.250	-.031	-.127	-.200	-.294	-.399	-.052	-.190	-.291	-.443	-.471
.350	-.059	-.141	-.189	-.266	-.334	-.111	-.235	-.298	-.378	-.450
.450	-.091	-.169	-.203	-.258	-.315	-.167	-.265	-.296	-.340	-.409
.550	-.108	-.185	-.203	-.240	-.289	-.159	-.235	-.251	-.284	-.372
.650	-.097	-.165	-.182	-.214	-.260	-.139	-.197	-.200	-.221	-.335
.750	-.089	-.148	-.159	-.187	-.247	-.090	-.127	-.126	-.143	-.267
.850	-.036	-.081	-.078	-.113	-.178	-.008	-.032	-.028	-.043	-.158
.900	.006	-.029	-.025	-.056	-.119	.044	.023	.024	.016	-.085
	Right side									
.025	.009	-.018	-.022	.216	.304	-.538	-.063	.271	.509	.666
.075	-.198	-.134	-.001	.092	.230	-.369	-.136	.092	.282	.449
.150	-.228	-.141	-.011	.057	.199	-.289	-.151	.015	.160	.315
.250	-.175	-.117	-.029	.044	.165	-.259	-.172	-.045	.073	.210
.350	-.184	-.136	-.063	.004	.104	-.262	-.202	-.099	.010	.124
.450	-.193	-.158	-.092	-.034	.053	-.263	-.228	-.143	-.063	.037
.550	-.176	-.153	-.099	-.050	.027	-.231	-.220	-.155	-.088	-.008
.650	-.153	-.136	-.083	-.041	.023	-.170	-.177	-.125	-.077	-.019
.750	-.132	-.132	-.076	-.044	.013	-.086	-.100	-.069	-.040	-.011
.850	-.048	-.053	-.018	-.007	.026	.006	-.006	.011	.016	.016
.900	-.003	-.018	.006	.009	.030	.045	.015	.025	.023	.006
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.438	.663	.474	.067	-.382	.403	.588	.293	-.167	-.531
.025	.255	-.177	-1.038	-1.607	-.916	.099	-.323	-1.115	-.970	-.582
.075	.093	-.202	-.511	-1.438	-.892	-.073	-.332	-.580	-.941	-.592
.150	-.007	-.226	-.429	-.967	-.832	-.155	-.241	-.323	-.854	-.599
.250	-.084	-.241	-.343	-.450	-.783	-.089	-.159	-.236	-.677	-.587
.350	-.146	-.268	-.306	-.417	-.739					
.450	-.187	-.278	-.287	-.331	-.672					
.550	-.186	-.254	-.246	-.251	-.596					
.650	-.142	-.182	-.170	-.170	-.524	-.101	-.095	-.120	-.343	-.481
.750	-.058	-.094	-.088	-.091	-.427	-.055	-.047	-.079	-.286	-.451
.850	-.004	-.022	-.012	-.026	-.322	-.014	.001	-.045	-.223	-.415
.900	.034	.018	.028	.014	-.272	.016	.023	-.023	-.186	-.396
	Right side									
.025	-.852	-.086	.307	.532	.700	-1.048	-.207	.152	.342	.478
.075	-.448	-.166	.103	.310	.510	-.511	-.301	-.096	.069	.251
.150	-.348	-.177	.016	.183	.364	-.269	-.243	-.172	-.117	.059
.250	-.325	-.226	-.076	.064	.226					
.350	-.293	-.255	-.146	-.033	.115					
.450	-.260	-.260	-.183	-.094	.033					
.550	-.211	-.233	-.176	-.117	-.022					
.650	-.136	-.187	-.136	-.096	-.035	-.077	-.088	-.100	-.113	-.109
.750	-.063	-.073	-.069	-.057	-.039	-.055	-.057	-.078	-.104	-.151
.850	.034	.016	.019	.020	-.012	.010	.022	-.001	-.038	-.126
.900	.073	.045	.044	.040	-.022	.033	.036	.014	-.028	-.150

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL. AND HORIZONTAL TAIL - Continued

(i) $\alpha = 9.7°$; $M = 0.90$.

$\dfrac{x}{c_v}$	C_p for $z/b_v = 0.11$					C_p for $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.508	.866	.598	.698		.600	.711	.654	.423	
.025	-.137	-.131	-.044	-.211		.209	-.131	-.767	-1.192	
.075	-.032	-.142	-.201	-.240		.057	-.182	-.418	-.948	
.150	-.052	-.183	-.303	-.370		-.015	-.189	-.363	-.582	
.250	-.060	-.138	-.226	-.331		-.074	-.198	-.323	-.520	
.350	-.086	-.151	-.215	-.317		-.140	-.254	-.350	-.495	
.450	-.122	-.182	-.234	-.322		-.209	-.294	-.362	-.471	
.550	-.145	-.202	-.234	-.299		-.214	-.275	-.266	-.357	
.650	-.133	-.186	-.203	-.247		-.196	-.230	-.222	-.203	
.750	-.130	-.173	-.189	-.209		-.126	-.137	-.138	-.137	
.850	-.069	-.083	-.091	-.124		-.030	-.030	-.026	-.042	
.900	-.022	-.031	-.029	-.065		.022	.026	.033	.009	
Right side										
.025	.061	-.018	-.052	.118		-.572	-.067	.271	.485	
.075	-.184	-.143	-.013	.048		-.436	-.151	.089	.262	
.150	-.258	-.150	-.021	.045		-.327	-.166	.010	.146	
.250	-.216	-.129	-.040	.036		-.305	-.187	-.049	.055	
.350	-.225	-.151	-.076	-.007		-.324	-.222	-.108	-.010	
.450	-.236	-.169	-.107	-.047		-.337	-.258	-.167	-.090	
.550	-.217	-.173	-.115	-.059		-.273	-.254	-.187	-.121	
.650	-.188	-.158	-.100	-.058		-.206	-.210	-.155	-.116	
.750	-.173	-.153	-.095	-.065		-.109	-.110	-.085	-.070	
.850	-.074	-.062	-.036	-.026		-.009	-.007	-.001	-.002	
.900	-.023	-.022	-.009	-.010		.031	.017	.018	.006	

$\dfrac{x}{c_v}$	C_p for $z/b_v = 0.66$					C_p for $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.464	.666	.512	.195		.421	.589	.340	-.045	
.025	.246	-.173	-1.048	-1.378		.098	-.375	-1.089	-1.338	
.075	.082	-.207	-.606	-1.234		-.094	-.335	-1.025	-1.224	
.150	-.022	-.238	-.462	-1.113		-.307	-.294	-.383	-1.090	
.250	-.104	-.259	-.437	-.645		-.140	-.145	-.181	-.940	
.350	-.192	-.315	-.418	-.612						
.450	-.254	-.321	-.276	-.377						
.550	-.253	-.298	-.268	-.179						
.650	-.190	-.191	-.178	-.137		-.118	-.091	-.127	-.222	
.750	-.088	-.091	-.087	-.079		-.073	-.041	-.080	-.199	
.850	-.019	-.011	-.007	-.020		-.022	.005	-.044	-.172	
.900	.015	.026	.033	.016		.003	.034	-.017	-.136	
Right side										
.025	-.951	-.089	.308	.517		-1.096	-.217	.150	.329	
.075	-.459	-.181	.107	.295		-.897	-.366	-.109	.052	
.150	-.405	-.190	.018	.169		-.389	-.287	-.291	-.177	
.250	-.416	-.243	-.084	.040						
.350	-.431	-.297	-.170	-.067						
.450	-.273	-.305	-.225	-.152						
.550	-.249	-.275	-.215	-.183						
.650	-.165	-.211	-.159	-.141		-.094	-.095	-.105	-.121	
.750	-.076	-.073	-.078	-.083		-.070	-.051	-.078	-.109	
.850	.026	.023	.018	.004		.002	.027	-.002	-.031	
.900	.088	.051	.046			.018	.045	.014	-.018	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(j) $\alpha = 9.7°$; M = 0.92.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.526		.621	.721		.602		.654	.441	
.025	-.158		-.015	-.189		.200		-.733	-1.175	
.075	-.038		-.171	-.219		.050		-.439	-.914	
.150	-.055		-.302	-.345		-.021		-.391	-.564	
.250	-.061		-.241	-.313		-.078		-.340	-.498	
.350	-.088		-.230	-.301		-.141		-.358	-.479	
.450	-.124		-.247	-.311		-.220		-.383	-.457	
.550	-.148		-.260	-.289		-.233		-.355	-.369	
.650	-.148		-.239	-.236		-.224		-.240	-.187	
.750	-.146		-.220	-.198		-.136		-.139	-.121	
.850	-.082		-.104	-.112		-.034		-.029	-.027	
.900	-.029		-.042	-.053		.018		.031	.022	
Right side										
.025	.079		-.092	.126		-.556		.248	.490	
.075	-.161		-.037	.059		-.436		.068	.269	
.150	-.248		-.034	.058		-.353		-.008	.158	
.250	-.216		-.050	.047		-.303		-.068	.070	
.350	-.225		-.087	.004		-.327		-.126	.002	
.450	-.235		-.121	-.040		-.349		-.189	-.079	
.550	-.223		-.133	-.053		-.316		-.220	-.112	
.650	-.199		-.129	-.049		-.207		-.206	-.105	
.750	-.183		-.126	-.059		-.107		-.108	-.061	
.850	-.078		-.060	-.019		-.007		-.015	.010	
.900	-.029		-.025	-.003		.034		.008	.021	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.466		.523	.219		.424		.347	-.014	
.025	.242		-1.024	-1.364		.095		-1.130	-1.353	
.075	.079		-.746	-1.220		-.092		-1.011	-1.222	
.150	-.025		-.471	-1.113		-.324		-.585	-1.098	
.250	-.109		-.439	-.625		-.210		-.286	-.972	
.350	-.199		-.459	-.597						
.450	-.286		-.442	-.426						
.550	-.286		-.235	-.158						
.650	-.208		-.169	-.118		-.117		-.123	-.191	
.750	-.088		-.084	-.064		-.073		-.080	-.174	
.850	-.017		-.005	-.006		-.021		-.042	-.154	
.900	.017		.039	.028		.009		-.016	-.122	
Right side										
.025	-.926		.292	.521		-1.076		.145	.341	
.075	-.540		.094	.301		-.908		-.105	.069	
.150	-.397		.005	.180		-.459		-.320	-.162	
.250	-.411		-.096	.057						
.350	-.447		-.192	-.051						
.450	-.358		-.281	-.140						
.550	-.231		-.294	-.177						
.650	-.161		-.195	-.133		-.096		-.102	-.122	
.750	-.075		-.087	-.071		-.073		-.081	-.090	
.850	.029		.010	.024		.005		-.001	-.016	
.900	.067		.040	.045		.022		.017	-.003	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(k) $\alpha = 15.6°$; M = 0.60.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.782		.807	.543	.216	.562		.645	.244	.202
.025	-.220		-.133	.219	.112	.145		-.392	-.817	-.734
.075	-.144		-.174	-.082	-.024	.011		-.314	-.537	-.564
.150	-.135		-.213	-.226	-.154	-.050		-.259	-.409	-.469
.250	-.088		-.158	-.194	-.154	-.095		-.252	-.345	-.426
.350	-.102		-.174	-.212	-.188	-.142		-.263	-.333	-.408
.450	-.131		-.195	-.233	-.217	-.171		-.263	-.308	-.385
.550	-.153		-.211	-.240	-.228	-.173		-.234	-.267	-.337
.650	-.146		-.188	-.217	-.213	-.153		-.202	-.228	-.294
.750	-.142		-.188	-.208	-.194	-.117		-.147	-.158	-.222
.850	-.097		-.126	-.149	-.136	-.052		-.078	-.087	-.163
.900	-.059		-.087	-.101	-.092	-.171		-.183	-.171	-.149
Right side										
.025	-.025		-.115	-.502	-.179	-.447		.151	.395	.232
.075	-.169		-.144	-.164	-.147	-.330		-.003	.148	.055
.150	-.196		-.128	-.066	-.095	-.272		-.060	.046	-.020
.250	-.158		-.099	-.032	-.047	-.250		-.106	-.028	-.061
.350	-.173		-.108	-.048	-.040	-.252		-.138	-.085	-.099
.450	-.196		-.135	-.091	-.049	-.256		-.167	-.121	-.126
.550	-.191		-.144	-.107	-.054	-.232		-.170	-.130	-.140
.650	-.176		-.135	-.117	-.054	-.185		-.144	-.114	-.126
.750	-.153		-.119	-.107	-.056	-.117		-.101	-.075	-.108
.850	-.095		-.071	-.071	-.029	-.043		-.019	-.025	-.072
.900	-.063		-.044	-.048	-.018	-.019		-.028	-.014	-.077

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.286		.412	-.222	-.814	.244		.167	-.422	-1.077
.025	.226		-.607	-1.201	-1.709	.125		-.682	-.909	-.952
.075	.058		-.424	-.954	-1.628	-.048		-.410	-.854	-.959
.150			-.353	-.623	-1.353	-.126		-.291	-.722	-.909
.250	.035		-.291	-.413	-.755	-.088		-.204	-.601	-.832
.350	-.155		-.282	-.363	-.480					
.450	-.187		-.268	-.320	-.385					
.550	-.189		-.236	-.276	-.317					
.650	-.160		-.183	-.206	-.233	-.135		-.133	-.301	-.634
.750	-.120		-.122	-.139	-.170	-.111		-.103	-.263	-.560
.850	-.061		-.069	-.085	-.115	-.075		-.076	-.222	-.469
.900	-.043		-.035	-.048	-.095	-.052		-.064	-.192	-.414
Right side										
.025	-.617		.229	.498	.629	-.642		.135	.333	.484
.075	-.447		.039	.242	.359	-.422		-.085	.068	.234
.150	-.335		-.039	.107	.191	-.297		-.147	-.066	.064
.250	-.303		-.110	-.005	.053					
.350	-.288		-.151	-.096	-.045					
.450	-.263		-.174	-.135	-.092					
.550	-.223		-.172	-.146	-.120					
.650	-.169		-.144	-.133	-.126	-.088		-.124	-.123	-.077
.750	-.111		-.103	-.098	-.115	-.106		-.110	-.117	-.099
.850	-.028		-.014	-.039	-.077	-.052		-.046	-.062	-.072
.900	.006		-.003	-.007	-.067	-.034		-.028	-.043	-.081

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL - Continued

(l) $\alpha = 15.8°$; M = 0.80.

$z/b_v = 0.11$ and $z/b_v = 0.38$

Left side

x/c_v	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.774	.838	.846	.303	.434	.694	.739	.749	.661	.289
.025	-.357	-.197	-.053	.525	.423	.089	-.126	-.374	-.680	-.280
.075	-.196	-.193	-.168	.116	.181	-.040	-.184	-.341	-.493	-.377
.150	-.180	-.229	-.244	-.178	-.030	-.088	-.193	-.291	-.413	-.419
.250	-.125	-.167	-.188	-.194	-.088	-.125	-.205	-.285	-.375	-.452
.350	-.128	-.164	-.194	-.224	-.148	-.170	-.236	-.307	-.360	-.454
.450	-.161	-.193	-.219	-.245	-.187	-.214	-.270	-.315	-.342	-.429
.550	-.185	-.215	-.232	-.252	-.210	-.215	-.258	-.283	-.301	-.394
.650	-.182	-.211	-.229	-.240	-.219	-.199	-.233	-.250	-.260	-.352
.750	-.199	-.229	-.241	-.254	-.248	-.152	-.171	-.179	-.189	-.282
.850	-.146	-.159	-.169	-.181	-.186	-.081	-.087	-.095	-.113	-.215
.900	-.105	-.112	-.113	-.132	-.138	-.162	-.173	-.168	-.139	-.138

Right side

x/c_v	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	.062	-.069	-.232	-.897	-.691	-.406	-.126	.101	.222	-.107
.075	-.162	-.191	-.221	-.562	-.292	-.354	-.188	-.048	.012	-.114
.150	-.218	-.200	-.175	-.151	-.160	-.305	-.191	-.097	-.059	-.113
.250	-.188	-.159	-.135	-.059	-.079	-.276	-.199	-.127	-.089	-.108
.350	-.202	-.170	-.135	-.052	-.055	-.289	-.224	-.163	-.126	-.123
.450	-.214	-.194	-.168	-.065	-.055	-.297	-.252	-.201	-.162	-.138
.550	-.209	-.202	-.180	-.080	-.058	-.273	-.252	-.209	-.171	-.142
.650	-.206	-.197	-.180	-.094	-.064	-.226	-.212	-.182	-.150	-.131
.750	-.205	-.200	-.186	-.107	-.075	-.149	-.147	-.128	-.112	-.114
.850	-.132	-.128	-.122	-.070	-.050	-.058	-.058	-.053	-.056	-.085
.900	-.093	-.090	-.089	-.053	-.040	-.028	-.038	-.037	-.049	-.087

$z/b_v = 0.66$ and $z/b_v = 0.93$

Left side

x/c_v	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.000	.469	.665	.582	.316	-.055	.364	.582	.304	-.175	-.763
.025	.216	-.149	-.782	-1.613	-1.551	.115	-.256	-1.110	-1.077	-.945
.075	.046	-.197	-.473	-1.129	-1.364	-.078	-.283	-.508	-1.002	-.950
.150	-.056	-.226	-.411	-.642	-1.125	-.165	-.239	-.324	-.830	-.910
.250	-.115	-.177	-.306	-.455	-.764	-.091	-.147	-.239	-.672	-.849
.350	-.194	-.270	-.326	-.402	-.566					
.450	-.239	-.288	-.317	-.346	-.461					
.550	-.249	-.276	-.282	-.293	-.376	-.147	-.131	-.154	-.328	-.630
.650	-.205	-.215	-.212	-.221	-.283	-.109	-.085	-.116	-.284	-.549
.750	-.140	-.140	-.135	-.148	-.202	-.069	-.040	-.078	-.237	-.463
.850	-.076	-.067	-.063	-.086	-.148	-.047	-.022	-.057	-.206	-.415
.900	-.044	-.035	-.028	-.055	-.123					

Right side

x/c_v	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
.025	-.798	-.161	.225	.477	.522	-1.076	-.264	.133	.355	.513
.075	-.498	-.223	.023	.213	.256	-.548	-.306	-.113	.066	.248
.150	-.382	-.209	-.048	.078	.119	-.317	-.258	-.180	-.098	.064
.250	-.353	-.244	-.136	-.042	.006					
.350	-.330	-.279	-.204	-.132	-.081					
.450	-.303	-.280	-.235	-.178	-.126					
.550	-.262	-.255	-.232	-.194	-.145					
.650	-.197	-.197	-.189	-.178	-.149	-.108	-.118	-.130	-.124	-.088
.750	-.118	-.125	-.131	-.133	-.140	-.112	-.087	-.119	-.115	-.107
.850	-.023	-.026	-.040	-.064	-.110	-.041	-.011	-.042	-.046	-.075
.900	.010	.004	-.012	-.044	-.094	-.026	.001	-.030	-.047	-.093

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE HIGH WING, FUSELAGE,

VERTICAL TAIL, AND HORIZONTAL TAIL - Concluded

(m) $\alpha = 15.9°$; M = 0.85.

$\frac{x}{c_v}$	C_p for–					C_p for–				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.755	.849	.854	.307		.704	.741	.748	.647	
.025	-.547	-.243	-.006	.511		.062	-.158	-.415	-.731	
.075	-.252	-.238	-.160	.127		-.076	-.225	-.380	-.518	
.150	-.223	-.281	-.282	-.184		-.118	-.223	-.323	-.446	
.250	-.152	-.208	-.214	-.223		-.153	-.233	-.310	-.420	
.350	-.153	-.192	-.221	-.252		-.196	-.262	-.332	-.410	
.450	-.183	-.225	-.241	-.274		-.245	-.300	-.343	-.384	
.550	-.207	-.249	-.252	-.268		-.243	-.294	-.306	-.341	
.650	-.213	-.243	-.252	-.261		-.227	-.272	-.279	-.300	
.750	-.237	-.277	-.278	-.287		-.185	-.205	-.204	-.233	
.850	-.179	-.199	-.196	-.213		-.101	-.111	-.110	-.157	
.900	-.135	-.143	-.137	-.163		-.173	-.185	-.167	-.144	
Right side										
.025	.104	-.092	-.346	-1.011		-.441	-.148	.090	.183	
.075	-.166	-.238	-.271	-.614		-.405	-.222	-.065	-.035	
.150	-.262	-.249	-.198	-.189		-.350	-.225	-.115	-.096	
.250	-.226	-.201	-.149	-.096		-.306	-.230	-.142	-.122	
.350	-.234	-.201	-.152	-.078		-.325	-.256	-.177	-.153	
.450	-.245	-.223	-.174	-.093		-.333	-.289	-.218	-.192	
.550	-.238	-.232	-.187	-.102		-.310	-.290	-.227	-.198	
.650	-.240	-.230	-.196	-.115		-.267	-.255	-.207	-.178	
.750	-.255	-.249	-.213	-.131		-.180	-.177	-.146	-.140	
.850	-.169	-.165	-.149	-.099		-.084	-.085	-.065	-.083	
.900	-.129	-.124	-.115	-.081		-.050	-.059	-.055	-.086	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.499	.669	.599	.368		.390	.578	.341	-.116	
.025	.194	-.175	-.865	-1.537		.087	-.327	-1.142	-1.389	
.075	.022	-.226	-.495	-1.338		-.124	-.364	-.574	-1.173	
.150	-.084	-.255	-.448	-.728		-.226	-.279	-.312	-.973	
.250	-.274	-.272	-.349	-.554		-.121	-.164	-.252	-.751	
.350	-.226	-.310	-.347	-.429						
.450	-.278	-.332	-.347	-.373						
.550	-.293	-.321	-.315	-.315						
.650	-.243	-.250	-.232	-.245		-.172	-.151	-.164	-.340	
.750	-.163	-.155	-.153	-.175		-.129	-.103	-.118	-.295	
.850	-.095	-.079	-.071	-.106		-.086	-.049	-.084	-.248	
.900	-.063	-.046	-.038	-.074		-.057	-.026	-.068	-.222	
Right side										
.025	-.892	-.167	.212	.449		-1.235	-.289	.112	.335	
.075	-.496	-.238	.014	.189		-.673	-.381	-.157	.027	
.150	-.434	-.232	-.062	.051		-.313	-.291	-.224	-.153	
.250	-.412	-.280	-.153	-.076						
.350	-.364	-.320	-.230	-.165						
.450	-.347	-.325	-.268	-.224						
.550	-.301	-.294	-.264	-.235						
.650	-.227	-.230	-.217	-.208		-.148	-.144	-.153	-.181	
.750	-.141	-.145	-.150	-.168		-.132	-.106	-.135	-.144	
.850	-.036	-.043	-.054	-.083		-.053	-.021	-.049	-.064	
.900	-.001	-.007	-.023	-.071		-.037	-.005	-.037	-.073	

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL

(a) $\alpha = 0°$; $M = 0.60$.

$\dfrac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.636	.644	.327	-.435	-1.287	.281	.728	.301	-.433	-1.248
.025	.262	-.113	-.546	-1.133	-2.371	.322	-.094	-.643	-1.082	-1.180
.075	.126	-.119	-.375	-.678	-1.461	.142	-.136	-.442	-.939	-1.143
.150	.031	-.140	-.306	-.484	-.686	.050	-.143	-.325	-.708	-1.094
.250	.017	-.101	-.202	-.308	-.436	-.013	-.152	-.274	-.445	-1.048
.350	-.024	-.113	-.177	-.246	-.354	-.068	-.170	-.255	-.345	-.883
.450	-.057	-.126	-.167	-.211	-.315	-.105	-.182	-.232	-.285	-.672
.550	-.066	-.108	-.133	-.165	-.331	-.096	-.152	-.179	-.218	-.505
.650	-.034	-.073	-.091	-.126	-.266	-.073	-.113	-.131	-.158	-.377
.750	-.015	-.036	-.047	-.096	-.183	-.034	-.055	-.066	-.084	-.264
.850	.020	.010	.003	-.031	-.097	.024	.017	.015	-.006	-.150
.900	.040	.038	.038	.006	-.041	.050	.045	.050	.040	-.092
	Right side									
.025	-.458	-.032	.315	.583	.815	-.634	-.075	.378	.592	.734
.075	-.380	-.115	.128	.338	.530	-.424	-.129	.156	.352	.513
.150	-.294	-.124	.050	.209	.372	-.318	-.133	.063	.221	.363
.250	-.204	-.106	.020	.137	.263	-.260	-.143	-.001	.114	.233
.350	-.179	-.113	-.017	.070	.170	-.234	-.157	-.052	.043	.147
.450	-.165	-.122	-.052	.020	.105	-.216	-.157	-.084	-.001	.082
.550	-.128	-.099	-.052	.010	.080	-.177	-.138	-.080	-.020	.038
.650	-.082	-.059	-.022	.029	.077	-.117	-.092	-.057	-.008	.026
.750	-.034	-.027	.001	.033	.073	-.050	-.032	-.020	.015	.033
.850	.020	.026	.043	.068	.089	.024	.036	.054	.061	.047
.900	.045	.043	.052	.068	.080	.052	.052	.057	.061	.031
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.234	.718	.253	-.403	-.700	.378	.672	.214	-.294	-.482
.025	.334	-.131	-.856	-1.070	-.765	.179	-.177	-.717	-.863	-.556
.075	.151	-.152	-.498	-.969	-.751	.020	-.163	-.389	-.810	-.554
.150	.040	-.163	-.375	-.789	-.742	-.050	-.147	-.290	-.710	-.591
.250	-.031	-.168	-.294	-.549	-.707	-.040	-.110	-.228	-.574	-.589
.350	-.087	-.182	-.260	-.375	-.672					
.450	-.119	-.189	-.234	-.288	-.637					
.550	-.117	-.170	-.195	-.223	-.586					
.650	-.089	-.119	-.131	-.156	-.538	-.080	-.073	-.112	-.391	-.419
.750	-.054	-.055	-.068	-.094	-.463	-.059	-.038	-.087	-.366	-.394
.850	.008	.006	-.008	-.036	-.398	-.038	-.008	-.066	-.320	-.371
.900	.031	.024	.024	-.003	-.354	-.015	.012	-.047	-.269	-.354
	Right side									
.025	-.715	-.085	.391	.588	.687	-.567	-.143	.223	.366	.437
.075	-.477	-.147	.160	.348	.493	-.368	-.166	.013	.105	.196
.150	-.336	-.143	.059	.209	.328	-.271	-.152	-.045	.006	.068
.250	-.285	-.166	-.017	.100	.198					
.350	-.264	-.177	-.077	.017	.096					
.450	-.230	-.177	-.103	-.036	.029					
.550	-.186	-.154	-.103	-.050	-.013					
.650	-.126	-.108	-.075	-.031	-.027	-.091	-.069	-.068	-.070	-.085
.750	-.057	-.052	-.036	-.008	-.036	-.077	-.045	-.075	-.075	-.132
.850	.020	.029	.031	.029	-.039	-.022	.015	-.015	-.034	-.129
.900	.050	.054	.052	.054	-.076	-.015	.026	.001	-.027	-.155

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(b) $\alpha = 0°$; $M = 0.80$.

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.11$					C_p for — $z/b_v = 0.38$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.736	.664	.523	.187	-.235	.466	.760	.555	.048	-.570
.025	.269	-.075	-.534	-1.147	-1.481	.319	-.080	-.929	-1.641	-.974
.075	.140	-.099	-.365	-.863	-1.237	.146	-.136	-.446	-1.459	-.985
.150	.046	-.142	-.323	-.535	-.982	.055	-.146	-.339	-.512	-.935
.250	.022	-.107	-.217	-.303	-.617	-.014	-.163	-.282	-.377	-.846
.350	-.020	-.122	-.184	-.241	-.407	-.078	-.183	-.270	-.342	-.748
.450	-.069	-.137	-.173	-.215	-.342	-.122	-.202	-.246	-.288	-.633
.550	-.063	-.113	-.136	-.166	-.291	-.111	-.164	-.185	-.210	-.522
.650	-.036	-.078	-.089	-.130	-.217	-.086	-.122	-.125	-.139	-.431
.750	-.013	-.048	-.039	-.084	-.175	-.034	-.051	-.054	-.060	-.332
.850	.020	.014	.020	-.021	-.104	.036	.031	.035	.020	-.235
.900	.048	.049	.054	.020	-.050	.075	.074	.075	.064	-.169
Right side										
.025	-.411	-.008	.314	.583	.812	-.934	-.058	.359	.588	.745
.075	-.393	-.107	.130	.343	.552	-.431	-.126	.154	.356	.539
.150	-.301	-.120	.054	.222	.392	-.331	-.140	.060	.225	.385
.250	-.219	-.107	.021	.154	.291	-.273	-.149	-.009	.123	.259
.350	-.189	-.119	-.026	.081	.192	-.248	-.166	-.065	.049	.170
.450	-.166	-.131	-.066	.020	.113	-.226	-.176	-.101	-.010	.085
.550	-.128	-.107	-.060	.008	.082	-.178	-.152	-.104	-.031	.041
.650	-.076	-.063	-.027	.026	.085	-.110	-.102	-.071	-.016	.029
.750	-.033	-.030	-.003	.035	.072	-.030	-.023	-.018	.017	.026
.850	.031	.031	.039	.069	.080	.048	.051	.057	.067	.032
.900	.058	.055	.053	.072	.069	.075	.069	.068	.070	.012

$\frac{x}{c_v}$	C_p for — $z/b_v = 0.66$					C_p for — $z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.434	.748	.495	.013	-.395	.463	.695	.383	-.110	-.296
.025	.334	-.114	-1.086	-1.312	-.707	.155	-.228	-1.101	-1.248	-.492
.075	.154	-.154	-.535	-1.174	-.710	-.004	-.187	-.484	-1.043	-.483
.150	.040	-.178	-.398	-.957	-.707	-.075	-.163	-.280	-.833	-.523
.250	-.042	-.186	-.308	-.743	-.659	-.046	-.117	-.235	-.579	-.502
.350	-.104	-.201	-.268	-.518	-.621					
.450	-.137	-.205	-.240	-.329	-.573					
.550	-.134	-.181	-.191	-.215	-.522					
.650	-.098	-.125	-.122	-.130	-.475	-.067	-.063	-.096	-.326	-.391
.750	-.049	-./7S	-.042	-.057	-.425	-.049	-.023	-.063	-.309	-.366
.850	.034	.025	.026	.005	-.371	-.014	.025	-.036	-.267	-.336
.900	.049	./7/	.057	.029	-.341	.011	.045	-.015	-.221	-.321
Right side										
.025	-1.249	-.069	.376	.580	.707	-1.089	-.161	.198	.365	.475
.075	-.502	-.145	.154	.351	.514	-.430	-.186	-.026	.096	.225
.150	-.348	-.146	.057	.216	.361	-.270	-.158	-.069	-.016	.092
.250	-.301	-.176	-.024	.102	.225					
.350	-.272	-.195	-.098	.008	.116					
.450	-.231	-.193	-.125	-.042	.044					
.550	-.181	-.166	-.122	-.062	-.003					
.650	-.108	-.108	-.084	-.042	-.027	-.072	-.051	-.056	-.063	-.072
.750	-.036	-.031	-.027	-.010	-.039	-.054	-.020	-.053	-.063	-.127
.850	.052	.054	.051	.045	-.036	.004	.043	.015	-.013	-.115
.900	.083	.083	.075	.061	-.068	.014	.058	.027	-.015	-.142

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(c) $\alpha = 0°$; M = 0.85.

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.767	.670	.568	.298	-.085	.527	.771	.604	.187	-.409
.025	.269	-.069	-.523	-1.069	-1.288	.317	-.082	-1.038	-1.475	-.922
.075	.139	-.097	-.379	-.831	-1.123	.139	-.140	-.422	-1.333	-.934
.150	.039	-.143	-.342	-.681	-.958	.052	-.154	-.352	-1.183	-.925
.250	.011	-.111	-.229	-.364	-.761	-.026	-.170	-.300	-.334	-.850
.350	-.033	-.123	-.193	-.255	-.595	-.089	-.196	-.283	-.284	-.774
.450	-.083	-.147	-.186	-.206	-.483	-.140	-.221	-.261	-.264	-.676
.550	-.082	-.124	-.142	-.159	-.391	-.131	-.182	-.195	-.203	-.581
.650	-.058	-.083	-.097	-.129	-.304	-.102	-.134	-.129	-.131	-.500
.750	-.033	-.049	-.046	-.081	-.259	-.045	-.058	-.057	-.056	-.408
.850	.018	.012	.013	-.013	-.171	.029	.026	.031	.036	-.318
.900	.048	.049	.055	.025	-.114	.072	.071	.074	.073	-.254
Right side										
.025	-.382	.000	.318	.588	.811	-.949	-.075	.356	.592	.739
.075	-.402	-.106	.133	.352	.549	-.418	-.133	.149	.359	.526
.150	-.326	-.123	.057	.228	.385	-.349	-.147	.055	.227	.369
.250	-.234	-.111	.018	.159	.274	-.288	-.162	-.019	.124	.242
.350	-.202	-.128	-.033	.079	.171	-.268	-.182	-.077	.045	.147
.450	-.182	-.143	-.077	.015	.086	-.240	-.194	-.118	-.020	.055
.550	-.137	-.117	-.071	.004	.055	-.188	-.165	-.121	-.042	.004
.650	-.082	-.072	-.038	.028	.055	-.117	-.114	-.087	-.026	-.014
.750	-.035	-.039	-.013	.035	.037	-.033	-.035	-.027	.009	-.019
.850	.034	.029	.037	.067	.039	.049	.050	.055	.067	-.014
.900	.061	.053	.054	.073	.023	.079	.070	.067	.073	-.045
$z/b_v = 0.66$						$z/b_v = 0.93$				
Left side										
.000	.492	.753	.544	.141	-.321	.494	.697	.420	.011	-.297
.025	.326	-.120	-1.158	-1.568	-.817	.135	-.235	-1.037	-1.254	-.553
.075	.148	-.164	-.625	-1.398	-.824	-.038	-.215	-.676	-1.090	-.550
.150	.028	-.188	-.405	-1.134	-.839	-.105	-.178	-.254	-.873	-.599
.250	-.057	-.202	-.314	-.711	-.713	-.061	-.127	-.243	-.620	-.578
.350	-.120	-.225	-.277	-.408	-.683					
.450	-.165	-.230	-.257	-.279	-.647					
.550	-.152	-.195	-.203	-.182	-.602					
.650	-.112	-.133	-.131	-.107	-.550	-.075	-.070	-.104	-.376	-.461
.750	-.058	-.051	-.050	-.040	-.486	-.051	-.032	-.070	-.337	-.434
.850	.028	.023	.021	.018	-.416	-.009	.020	-.041	-.286	-.406
.900	.051	.051	.054	.047	-.380	.013	.047	-.021	-.244	-.389
Right side										
.025	-1.152	-.082	.366	.583	.702	-1.140	-.188	.180	.361	.464
.075	-.518	-.160	.146	.352	.500	-.672	-.225	-.058	.080	.210
.150	-.370	-.162	.050	.217	.344	-.205	-.178	-.100	-.056	.052
.250	-.315	-.196	-.041	.098	.203					
.350	-.281	-.219	-.118	-.006	.089					
.450	-.241	-.216	-.152	-.063	.000					
.550	-.192	-.181	-.145	-.077	-.052					
.650	-.114	-.114	-.100	-.056	-.079	-.085	-.056	-.067	-.070	-.130
.750	-.035	-.039	-.041	-.019	-.091	-.061	-.028	-.053	-.069	-.180
.850	.058	.054	.051	.053	-.081	.006	.044	.018	-.012	-.161
.900	.093	.085	.081	.070	-.115	.017	.057	.028	-.022	-.197

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(d) $\alpha = 0°$; $M = 0.90$.

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.799	.681	.613	.394	.068	.585	.785	.657	.318	-.223
.025	.278	-.052	-.484	-.960	-1.156	.316	-.068	-.937	-1.287	-.825
.075	.147	-.092	-.388	-.762	-.987	.140	-.140	-.706	-1.176	-.839
.150	.041	-.153	-.383	-.678	-.845	.047	-.162	-.411	-1.068	-.820
.250	.013	-.125	-.258	-.521	-.687	-.027	-.181	-.323	-.648	-.764
.350	-.034	-.138	-.217	-.344	-.581	-.097	-.218	-.344	-.470	-.708
.450	-.091	-.173	-.216	-.279	-.504	-.160	-.255	-.289	-.305	-.633
.550	-.090	-.146	-.157	-.183	-.419	-.152	-.210	-.204	-.155	-.551
.650	-.061	-.101	-.098	-.124	-.345	-.117	-.149	-.132	-.099	-.484
.750	-.034	-.062	-.050	-.075	-.296	-.053	-.056	-.053	-.036	-.408
.850	.012	.012	.018	-.012	-.204	.035	.033	.041	.044	-.337
.900	.041	.051	.055	.030	-.135	.079	.083	.086	.082	-.266
Right side										
.025	-.334	.005	.325	.580	.821	-.842	-.072	.352	.580	.754
.075	-.407	-.108	.139	.347	.565	-.605	-.146	.149	.347	.541
.150	-.346	-.133	.054	.221	.402	-.377	-.157	.050	.213	.388
.250	-.246	-.126	.013	.145	.294	-.301	-.178	-.026	.107	.263
.350	-.220	-.148	-.042	.060	.191	-.299	-.210	-.092	.019	.164
.450	-.195	-.167	-.090	-.009	.102	-.247	-.229	-.142	-.051	.064
.550	-.136	-.134	-.092	-.025	.063	-.185	-.193	-.150	-.079	.010
.650	-.077	-.080	-.050	.005	.063	-.109	-.128	-.105	-.059	-.010
.750	-.026	-.040	-.027	.013	.043	-.023	-.027	-.036	-.010	-.017
.850	.041	.028	.027	.049	.032	.064	.056	.050	.052	-.018
.900	.071	.059	.043	.053	.013	.091	.080	.067	.063	-.048

$\dfrac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.544	.769	.604	.271	-.166	.532	.702	.469	.144	-.199
.025	.327	-.105	-1.052	-1.378	-.775	.085	-.271	-.985	-1.318	-.559
.075	.146	-.162	-.887	-1.248	-.775	-.069	-.277	-.969	-1.285	-.513
.150	.022	-.198	-.485	-1.151	-.798	-.173	-.214	-.500	-1.114	-.556
.250	-.070	-.229	-.400	-1.093	-.655	-.074	-.133	-.162	-.964	-.550
.350	-.149	-.266	-.273	-.662	-.629					
.450	-.192	-.269	-.241	-.275	-.601					
.550	-.176	-.223	-.210	-.101	-.562					
.650	-.124	-.141	-.125	-.053	-.527	-.073	-.064	-.104	-.241	-.443
.750	-.063	-.037	-.042	-.014	-.477	-.042	-.021	-.069	-.278	-.423
.850	.032	.037	.033	.030	-.422	.004	.036	-.042	-.258	-.395
.900	.059	.067	.067	.053	-.380	.025	.060	-.018	-.216	-.376
Right side										
.025	-.992	-.085	.361	.568	.716	-1.037	-.215	.167	.344	.488
.075	-.801	-.171	.143	.340	.515	-.892	-.287	-.097	.049	.228
.150	-.350	-.167	.045	.206	.364	-.293	-.211	-.189	-.108	.068
.250	-.381	-.222	-.062	.075	.224					
.350	-.243	-.261	-.152	-.041	.102					
.450	-.246	-.253	-.192	-.113	.009					
.550	-.192	-.205	-.176	-.132	-.053					
.650	-.106	-.120	-.116	-.094	-.069	-.071	-.053	-.066	-.079	-.121
.750	-.024	-.031	-.045	-.039	-.085	-.051	-.024	-.050	-.072	-.173
.850	.072	.065	.057	.044	-.068	.021	.057	.027	-.010	-.145
.900	.103	.101	.086	.067	-.099	.029	.075	.035	-.012	-.181

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(e) $\alpha = 0°$; $M = 0.92$.

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.811	.684	.627	.442	.111	.593	.791	.681	.381	-.179
.025	.267	-.060	-.466	-.885	-1.154	.309	-.075	-.927	-1.174	-.816
.075	.133	-.100	-.389	-.714	-.973	.126	-.151	-.719	-1.070	-.825
.150	.019	-.166	-.396	-.650	-.832	.030	-.174	-.434	-.972	-.792
.250	-.011	-.139	-.295	-.570	-.671	-.050	-.195	-.341	-.713	-.763
.350	-.060	-.155	-.238	-.406	-.578	-.126	-.244	-.360	-.576	-.710
.450	-.124	-.196	-.238	-.365	-.531	-.203	-.295	-.365	-.494	-.643
.550	-.129	-.178	-.179	-.273	-.446	-.199	-.242	-.190	-.288	-.571
.650	-.094	-.120	-.104	-.170	-.379	-.149	-.167	-.125	-.084	-.506
.750	-.069	-.075	-.052	-.088	-.343	-.078	-.068	-.048	-.016	-.434
.850	-.017	-.007	.019	-.020	-.251	.010	.031	.044	.052	-.376
.900	.017	.043	.059	.027	-.186	.055	.077	.090	.094	-.312
Right side										
.025	-.354	-.002	.324	.577	.824	-.903	-.094	.353	.564	.750
.075	-.438	-.125	.139	.342	.567	-.677	-.166	.147	.339	.538
.150	-.403	-.151	.050	.213	.399	-.450	-.174	.048	.208	.387
.250	-.306	-.143	.009	.138	.292	-.342	-.198	-.031	.100	.261
.350	-.259	-.163	-.055	.048	.189	-.350	-.232	-.101	.008	.161
.450	-.236	-.186	-.106	-.026	.092	-.327	-.264	-.163	-.072	.055
.550	-.168	-.161	-.108	-.046	.056	-.200	-.220	-.171	-.105	-.002
.650	-.100	-.096	-.063	-.017	.051	-.126	-.142	-.125	-.091	-.025
.750	-.049	-.052	-.038	-.009	.024	-.041	-.039	-.042	-.037	-.035
.850	.022	.018	.023	.032	.011	.051	.050	.048	.040	-.039
.900	.050	.051	.039	.043	-.020	.077	.072	.065	.060	-.077
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.550	.769	.631	.334	-.127	.538	.706	.497	.205	-.179
.025	.315	-.114	-1.030	-1.260	-.758	.124	-.258	-1.051	-1.218	-.527
.075	.132	-.174	-.886	-1.146	-.762	-.092	-.299	-.968	-1.173	-.512
.150	.007	-.212	-.711	-1.054	-.787	-.248	-.258	-.785	-1.038	-.551
.250	-.092	-.249	-.402	-1.008	-.667	-.133	-.150	-.262	-.874	-.554
.350	-.184	-.318	-.439	-.968	-.637					
.450	-.251	-.299	-.198	-.545	-.606					
.550	-.225	-.252	-.186	-.133	-.575					
.650	-.161	-.139	-.118	-.021	-.550	-.090	-.069	-.104	-.437	-.466
.750	-.061	-.046	-.035	.022	-.509	-.060	-.024	-.072	-.349	-.445
.850	.015	.032	.042	.060	-.456	-.014	.027	.050	-.299	-.424
.900	.041	.064	.077	.078	-.424	.010	.057	-.028	-.236	-.406
Right side										
.025	-1.021	-.102	.359	.559	.712	-1.054	-.234	.168	.342	.490
.075	-.858	-.198	.143	.330	.511	-.939	-.352	-.102	.049	.225
.150	-.573	-.190	.046	.204	.362	-.657	-.260	-.224	-.119	.062
.250	-.410	-.246	-.068	.068	.221					
.350	-.425	-.319	-.172	-.061	.092					
.450	-.219	-.277	-.225	-.144	-.010					
.550	-.196	-.227	-.199	-.174	-.075					
.650	-.126	-.126	-.124	-.128	-.098	-.093	-.060	-.063	-.098	-.145
.750	-.041	-.038	-.044	-.059	-.111	-.072	-.028	-.047	-.075	-.202
.850	.059	.061	.059	.041	-.091	.001	.050	.031	-.005	-.162
.900	.090	.093	.088	.073	-.125	.002	.067	.040	-.001	-.204

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(f) $\alpha = 9.4°$; $M = 0.60$.

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	C_p for $\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
		$z/b_v = 0.11$					$z/b_v = 0.38$			
					Left side					
.000	.451	.820	.562	.091	-1.139	.032	.629	-.041	-1.040	-1.779
.025		-.127	-.041	-.463	-1.002	.323	-.080	-.747	-1.359	-1.647
.075	.006	-.130	-.169	-.417	-.702	.155	-.107	-.416	-1.452	-2.034
.150	.013	-.148	-.215	-.375	-.524	.071	-.112	-.295	-.428	-1.665
.250	.011	-.127	-.158	-.271	-.364	.004	-.127	-.251	-.320	-.433
.350	-.023	-.123	-.164	-.234	-.301	-.058	-.155	-.249	-.320	-.389
.450	-.065	-.155	-.176	-.228	-.287	-.097	-.173	-.235	-.290	-.371
.550	-.095	-.173	-.180	-.216	-.273	-.099	-.155	-.199	-.241	-.324
.650	-.099	-.159	-.160	-.188	-.241	-.097	-.132	-.160	-.200	-.280
.750	-.104	-.150	-.144	-.156	-.206	-.067	-.089	-.110	-.142	-.224
.850	-.065	-.100	-.089	-.091	-.120	-.012	-.016	-.034	-.059	-.143
.900	-.028	-.064	-.048	-.040	-.062	.022	.021	.002	-.015	-.085
					Right side					
.025	.029	-.089	-.055	.195	.662	-.604	-.018	.386	.532	.609
.075	-.168	-.143	.014	.174	.506	-.372	-.073	.183	.350	.479
.150	-.202	-.132	.036	.160	.390	-.283	-.089	.091	.223	.356
.250	-.161	-.114	.030	.156	.305	-.234	-.107	.023	.128	.242
.350	-.168	-.123	-.009	.112	.216	-.225	-.127	-.023	.059	.168
.450	-.182	-.150	-.055	.061	.138	-.218	-.141	-.075	.001	.091
.550	-.175	-.159	-.073	.020	.100	-.191	-.139	-.089	-.031	.049
.650	-.159	-.150	-.080	-.001	.068	-.145	-.118	-.073	-.031	.033
.750	-.136	-.137	-.082	-.020	.042	-.090	-.073	-.043	-.010	.029
.850	-.069	-.080	-.037	.015	.063	-.017	.009	.014	.031	.045
.900	-.042	-.055	-.014	.024	.059	.011	.018	.025	.031	.038
		$z/b_v = 0.66$					$z/b_v = 0.93$			
					Left side					
.000	-.005	.563	-.098	-.786	-1.255	.151	.506	-.096	-.431	-.802
.025	.293	-.137	-.945	-1.107	-1.009	.135	-.221	-.804	-.646	-.619
.075	.132	-.150	-.482	-1.077	-1.002	-.028	-.196	-.427	-.613	-.614
.150	.034	-.157	-.356	-1.040	-1.004	-.113	-.166	-.267	-.576	-.605
.250	-.037	-.159	-.281	-.807	-1.018	-.072	-.114	-.187	-.488	-.570
.350	-.085	-.168	-.258	-.419	-.927					
.450	-.122	-.182	-.231	-.232	-.765					
.550	-.127	-.166	-.201	-.202	-.607					
.650	-.097	-.125	-.146	-.156	-.440	-.097	-.066	-.107	-.331	-.477
.750	-.058	-.071	-.085	-.098	-.287	-.062	-.025	-.073	-.292	-.461
.850	-.014	-.016	-.025	-.040	-.176	-.028	.007	-.043	-.248	-.433
.900	.004	.007	.002	-.010	-.129	-.003	.027	-.028	-.211	-.408
					Right side					
.025	-.730	-.043	.347	.505	.558	-.590	-.134	.185	.276	.332
.075	-.434	-.112	.157	.327	.442	-.370	-.173	-.023	.068	.156
.150	-.306	-.118	.062	.195	.316	-.244	-.159	-.107	-.068	-.006
.250	-.271	-.146	-.016	.091	.193					
.350	-.248	-.162	-.069	.015	.103					
.450	-.223	-.168	-.101	-.031	.040					
.550	-.186	-.153	-.110	-.057	.001					
.650	-.134	-.123	-.078	-.047	-.013	-.053	-.059	-.085	-.105	-.120
.750	-.076	-.073	-.048	-.027	-.009	-.069	-.034	-.073	-.105	-.157
.850	.004	.025	.023	.024	.012	-.014	.023	-.009	-.061	-.141
.900	.029	.039	.036	.038	.015	.006	.039	-.002	-.061	-.157

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(g) $\alpha = 9.6^\circ$; M = 0.80.

$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.451	.871	.577	.270	-.398	.262	.667	.236	-.384	-.962
.025		-.129	.018	-.372	-.575	.324	-.068	-.857	-1.336	-1.226
.075	.008	-.120	-.152	-.406	-.524	.164	-.098	-.441	-1.307	-1.222
.150	.023	-.144	-.236	-.416	-.451	.081	-.103	-.294	-.717	-1.088
.250	.022	-.120	-.170	-.296	-.236	.014	-.121	-.259	-.316	-.684
.350	-.013	-.118	-.174	-.244	-.188	-.056	-.164	-.262	-.320	-.379
.450	-.060	-.153	-.190	-.232	-.174	-.107	-.194	-.256	-.301	-.293
.550	-.100	-.184	-.201	-.224	-.155	-.121	-.181	-.222	-.264	-.249
.650	-.119	-.179	-.183	-.205	-.141	-.122	-.159	-.184	-.226	-.207
.750	-.127	-.185	-.178	-.185	-.120	-.086	-.107	-.122	-.157	-.149
.850	-.084	-.121	-.106	-.105	-.045	-.019	-.023	-.033	-.066	-.047
.900	-.053	-.075	-.060	-.047	.018	.017	.021	.016	-.011	.013
Right side										
.025	.102	-.065	-.076	.195	.626	-.717	-.011	.386	.539	.691
.075	-.148	-.141	.018	.187	.555	-.380	-.068	.193	.353	.566
.150	-.209	-.130	.042	.152	.487	-.279	-.080	.100	.236	.450
.250	-.163	-.109	.034	.148	.411	-.239	-.103	.033	.148	.350
.350	-.172	-.120	-.005	.110	.321	-.233	-.132	-.024	.071	.274
.450	-.183	-.147	-.056	.062	.247	-.232	-.164	-.082	-.003	.195
.550	-.180	-.165	-.082	.021	.198	-.206	-.164	-.105	-.041	.148
.650	-.166	-.164	-.094	-.011	.165	-.157	-.141	-.097	-.052	.122
.750	-.162	-.168	-.105	-.037	.129	-.091	-.084	-.060	-.028	.118
.850	-.084	-.098	-.056	-.006	.141	-.009	.007	.010	.021	.136
.900	-.053	-.066	-.028	.007	.143	.020	.019	.024	.026	.123

$\frac{x}{c_v}$	$z/b_v = 0.66$					$z/b_v = 0.93$				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
Left side										
.000	.193	.592	.154	-.392	-.751	.242	.520	.096	-.285	-.614
.025	.293	-.126	-.987	-1.031	-.701	.130	-.237	-.729	-.595	-.438
.075	.143	-.146	-.622	-1.012	-.726	-.043	-.237	-.583	-.580	-.437
.150	.034	-.161	-.369	-1.015	-.698	-.142	-.184	-.381	-.541	-.430
.250	-.037	-.173	-.302	-.909	-.669	-.071	-.117	-.221	-.471	-.405
.350	-.094	-.194	-.271	-.645	-.633					
.450	-.145	-.205	-.245	-.265	-.564					
.550	-.157	-.194	-.212	-.185	-.477					
.650	-.121	-.142	-.148	-.157	-.370	-.086	-.055	-.099	-.339	-.301
.750	-.068	-.069	-.076	-.095	-.264	-.043	-.008	-.060	-.316	-.282
.850	-.009	-.002	-.008	-.035	-.168	-.010	.027	-.019	-.272	-.254
.900	.016	.024	.025	-.002	-.121	.014	.050	.007	-.246	-.246
Right side										
.025	-.907	-.042	.356	.510	.642	-.819	-.147	.172	.305	.453
.075	-.476	-.110	.163	.328	.526	-.456	-.213	-.050	.079	.281
.150	-.312	-.120	.068	.206	.414	-.242	-.182	-.146	-.083	.111
.250	-.273	-.159	-.021	.096	.303					
.350	-.260	-.182	-.088	.009	.209					
.450	-.235	-.193	-.125	-.055	.139					
.550	-.195	-.175	-.132	-.083	.094					
.650	-.135	-.132	-.099	-.067	.076	-.034	-.051	-.074	-.107	-.020
.750	-.063	-.078	-.059	-.040	.071	-.045	-.017	-.059	-.115	-.052
.850	.028	.041	.024	.023	.086	.022	.048	.007	-.067	-.040
.900	.054	.059	.050	.045	.074	.037	.062	.025	-.066	-.062

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(h) $\alpha = 9.7°$; M = 0.85.

$\frac{x}{c_v}$	C_p for $\beta=-3.9°$	C_p for $\beta=0°$	C_p for $\beta=3.9°$	C_p for $\beta=7.9°$	C_p for $\beta=12.7°$	C_p for $\beta=-3.9°$	C_p for $\beta=0°$	C_p for $\beta=3.9°$	C_p for $\beta=7.9°$	C_p for $\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	Left side									
.000	.461	.895	.586	.304	-.482	.318	.679	.316	-.231	-.910
.025		-.123	.047	-.331	-.634	.325	-.072	-.847	-1.300	-1.426
.075	.006	-.123	-.141	-.391	-.633	.167	-.102	-.462	-1.154	-1.293
.150	.026	-.143	-.250	-.447	-.644	.080	-.106	-.298	-.756	-1.145
.250	.024	-.119	-.181	-.320	-.422	.013	-.123	-.267	-.397	-.892
.350	-.016	-.117	-.178	-.261	-.319	-.060	-.167	-.274	-.338	-.598
.450	-.060	-.153	-.195	-.243	-.325	-.116	-.207	-.274	-.317	-.490
.550	-.103	-.186	-.210	-.231	-.308	-.137	-.199	-.241	-.291	-.432
.650	-.126	-.189	-.192	-.217	-.291	-.143	-.180	-.201	-.253	-.398
.750	-.145	-.204	-.201	-.206	-.292	-.104	-.123	-.135	-.177	-.339
.850	-.094	-.137	-.124	-.123	-.217	-.026	-.029	-.042	-.077	-.223
.900	-.053	-.086	-.069	-.060	-.135	.017	.022	.015	-.017	-.135
	Right side									
.025	.123	-.059	-.095	.184	.504	-.790	-.001	.381	.540	.649
.075	-.142	-.140	.010	.183	.470	-.386	-.063	.190	.354	.504
.150	-.212	-.125	.041	.147	.414	-.282	-.080	.101	.239	.381
.250	-.166	-.105	.034	.140	.337	-.243	-.106	.030	.152	.271
.350	-.175	-.116	-.006	.104	.240	-.240	-.136	-.030	.069	.182
.450	-.189	-.146	-.053	.059	.152	-.246	-.174	-.095	-.016	.084
.550	-.186	-.170	-.088	.016	.088	-.222	-.182	-.126	-.060	.023
.650	-.172	-.167	-.109	-.023	.045	-.175	-.159	-.116	-.074	-.013
.750	-.172	-.183	-.129	-.071	-.012	-.099	-.099	-.076	-.048	-.022
.850	-.092	-.110	-.072	-.030	-.001	-.010	.002	.000	.004	-.004
.900	-.052	-.070	-.043	-.013	.006	.024	.021	.017	.014	-.006
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	Left side									
.000	.236	.598	.219	-.307	-.797	.262	.524	.121	-.286	-.721
.025	.292	-.130	-1.019	-1.166	-.848	.102	-.273	-.778	-.660	-.585
.075	.139	-.152	-.661	-1.137	-.849	-.064	-.278	-.641	-.640	-.589
.150	.030	-.167	-.384	-1.086	-.821	-.166	-.197	-.412	-.608	-.588
.250	-.049	-.183	-.321	-.888	-.783	-.109	-.130	-.237	-.534	-.565
.350	-.115	-.211	-.290	-.530	-.756					
.450	-.162	-.229	-.267	-.248	-.699					
.550	-.179	-.216	-.230	-.233	-.624					
.650	-.139	-.156	-.159	-.183	-.541	-.087	-.062	-.102	-.366	-.465
.750	-.083	-.075	-.085	-.106	-.446	-.043	-.012	-.059	-.321	-.441
.850	-.010	-.001	.013	-.036	-.364	-.004	.034	-.020	-.273	-.405
.900	.017	.032	.023	.002	-.306	.024	.058	.008	-.237	-.383
	Right side									
.025	-.913	-.035	.345	.503	.592	-.881	-.159	.164	.304	.388
.075	-.499	-.106	.156	.323	.460	-.524	-.243	-.066	.074	.194
.150	-.316	-.119	.064	.207	.336	-.255	-.202	-.177	-.111	-.005
.250	-.295	-.166	-.029	.090	.213					
.350	-.273	-.194	-.102	-.006	.105					
.450	-.243	-.210	-.151	-.078	.021					
.550	-.203	-.193	-.161	-.111	-.037					
.650	-.137	-.146	-.125	-.100	-.059	-.069	-.053	-.086	-.121	-.151
.750	-.066	-.090	-.072	-.064	-.074	-.043	-.018	-.066	-.127	-.203
.850	.027	.045	.017	.010	-.054	.026	.053	.004	-.073	-.186
.900	.059	.066	.044	.032	-.064	.044	.069	.025	-.067	-.197

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(i) $\alpha = 9.7^{\circ}$; M = 0.90.

$\frac{x}{c_v}$	C_p for—					C_p for—				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.478	.915	.599	.343	-.367	.386	.684	.391	-.116	-.729
.025		-.131	.072	-.276	-.515	.339	-.081	-1.029	-1.414	-1.321
.075	.017	-.136	-.121	-.350	-.558	.180	-.118	-.453	-.957	-1.211
.150	.039	-.166	-.267	-.449	-.617	.101	-.123	-.321	-.680	-1.080
.250	.036	-.139	-.190	-.335	-.504	.034	-.139	-.276	-.503	-.900
.350	.003	-.133	-.187	-.268	-.328	-.041	-.187	-.291	-.361	-.591
.450	-.047	-.173	-.204	-.238	-.305	-.105	-.238	-.295	-.327	-.488
.550	-.090	-.211	-.219	-.225	-.295	-.144	-.250	-.255	-.305	-.434
.650	-.128	-.228	-.206	-.217	-.278	-.170	-.292	-.222	-.256	-.414
.750	-.172	-.309	-.230	-.214	-.307	-.127	-.214	-.144	-.170	-.382
.850	-.111	-.186	-.135	-.113	-.266	-.027	-.050	-.040	-.063	-.272
.900	-.062	-.109	-.075	-.047	-.186	.022	.019	.015	-.002	-.189
Right side										
.025	.169	-.068	-.096	.191	.413	-.856	-.007	.387	.554	.657
.075	-.111	-.158	.008	.191	.441	-.377	-.077	.197	.366	.508
.150	-.218	-.144	.046	.158	.413	-.288	-.097	.110	.251	.385
.250	-.164	-.123	.039	.152	.347	-.236	-.118	.038	.157	.275
.350	-.176	-.135	-.005	.116	.248	-.243	-.157	-.024	.082	.189
.450	-.181	-.167	-.049	.066	.164	-.257	-.207	-.099	-.006	.087
.550	-.184	-.199	-.088	.024	.096	-.230	-.243	-.141	-.056	.015
.650	-.173	-.214	-.119	-.022	.040	-.178	-.263	-.149	-.083	-.030
.750	-.185	-.275	-.156	-.082	-.042	-.093	-.183	-.100	-.062	-.050
.850	-.090	-.166	-.091	-.037	-.041	.001	-.021	-.010	.003	-.036
.900	-.055	-.106	-.056	-.017	-.040	.038	.010	.008	.018	-.048
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.295	.601	.286	-.204	-.695	.300	.524	.155	-.323	-.660
.025	.306	-.136	-1.104	-1.482	-.821	.131	-.314	-1.099	-.877	-.571
.075	.155	-.162	-.644	-1.370	-.813	-.043	-.290	-.886	-.793	-.563
.150	.053	-.184	-.403	-.888	-.772	-.250	-.327	-.383	-.718	-.566
.250	-.033	-.209	-.366	-.694	-.743	-.122	-.167	-.147	-.631	-.556
.350	-.111	-.264	-.326	-.401	-.711					
.450	-.188	-.323	-.271	-.320	-.668					
.550	-.221	-.309	-.244	-.257	-.609					
.650	-.161	-.241	-.167	-.177	-.550	-.077	-.063	-.105	-.327	-.478
.750	-.078	-.081	-.080	-.086	-.487	-.026	-.014	-.056	-.256	-.456
.850	.001	.003	-.005	-.017	-.415	.019	.035	-.009	-.189	-.421
.900	.035	.039	.034	.020	-.381	.042	.067	.020	-.154	-.406
Right side										
.025	-1.040	-.042	.352	.512	.601	-1.085	-.178	.174	.324	.401
.075	-.454	-.118	.162	.332	.464	-.645	-.303	-.064	.089	.210
.150	-.319	-.133	.071	.217	.342	-.277	-.320	-.275	-.122	-.009
.250	-.326	-.192	-.028	.101	.218					
.350	-.313	-.248	-.117	.001	.103					
.450	-.260	-.302	-.180	-.086	.007					
.550	-.210	-.286	-.196	-.124	-.062					
.650	-.134	-.228	-.139	-.108	-.096	-.064	-.050	-.085	-.124	-.181
.750	-.049	-.084	-.077	-.059	-.109	-.023	-.020	-.059	-.111	-.233
.850	.048	.043	.018	.016	-.081	.042	.056	.019	-.041	-.207
.900	.080	.068	.048	.039	-.100	.061	.077	.038	-.024	-.226

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(j) $\alpha = 9.7°$; M = 0.92.
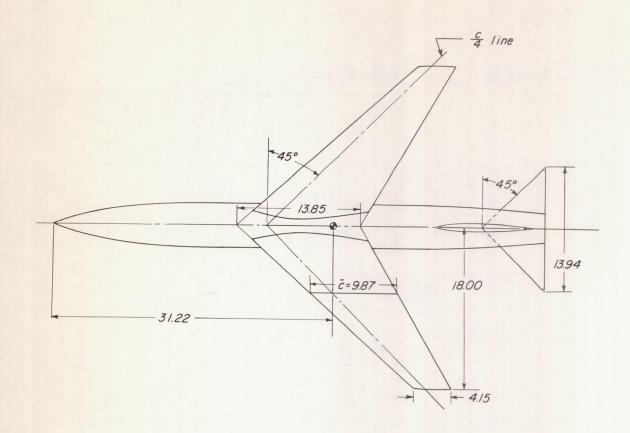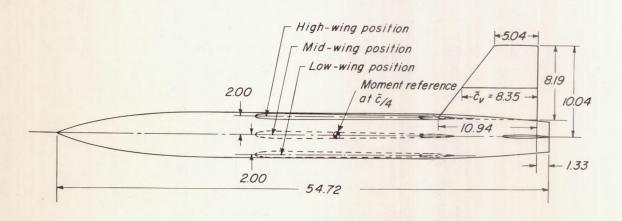
$\frac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.479	.924	.610	.358	-.327	.399	.690	.420	-.034	-.669
.025		-.115	.089	-.243	-.478	.327	-.071	-1.024	-1.348	-1.269
.075	.000	-.122	-.112	-.324	-.535	.174	-.106	-.471	-.934	-1.169
.150	.023	-.153	-.271	-.442	-.601	.087	-.112	-.329	-.695	-1.050
.250	.021	-.126	-.197	-.377	-.544	.019	-.126	-.277	-.542	-.901
.350	-.014	-.116	-.194	-.310	-.369	-.058	-.171	-.295	-.419	-.650
.450	-.062	-.160	-.214	-.270	-.323	-.126	-.226	-.312	-.336	-.511
.550	-.106	-.195	-.236	-.237	-.303	-.167	-.242	-.290	-.333	-.463
.650	-.147	-.209	-.216	-.235	-.290	-.209	-.267	-.237	-.328	-.441
.750	-.198	-.280	-.262	-.264	-.319	-.153	-.170	-.156	-.203	-.418
.850	-.136	-.157	-.148	-.144	-.315	-.047	-.042	-.045	-.081	-.316
.900	-.082	-.098	-.081	-.066	-.240	.009	.024	.019	-.011	-.217
Right side										
.025	.159	-.052	-.103	.174	.379	-.822	.005	.386	.549	.663
.075	-.123	-.144	.006	.181	.433	-.396	-.065	.199	.362	.512
.150	-.235	-.128	.047	.152	.414	-.306	-.081	.107	.248	.391
.250	-.182	-.110	.041	.142	.349	-.251	-.107	.037	.154	.278
.350	-.194	-.118	.002	.106	.248	-.262	-.140	-.023	.077	.190
.450	-.204	-.149	-.051	.056	.162	-.281	-.191	-.101	-.014	.087
.550	-.205	-.178	-.090	.014	.092	-.259	-.224	-.155	-.080	.014
.650	-.192	-.194	-.131	-.035	.033	-.205	-.234	-.187	-.124	-.044
.750	-.213	-.249	-.177	-.119	-.058	-.109	-.145	-.120	-.106	-.076
.850	-.109	-.143	-.107	-.082	-.075	-.014	-.002	-.020	-.026	-.066
.900	-.066	-.094	-.068	-.051	-.079	.026	.020	.006	-.001	-.076
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	.302	.610	.311	-.128	-.658	.300	.525	.182	-.256	-.656
.025	.298	-.126	-1.131	-1.390	-.834	.080	-.306	-1.153	-.926	-.593
.075	.143	-.149	-.716	-1.220	-.821	-.058	-.280	-1.045	-.825	-.577
.150	.034	-.175	-.409	-1.069	-.779	-.301	-.314	-.457	-.755	-.577
.250	-.051	-.196	-.370	-.747	-.742	-.153	-.126	-.153	-.685	-.572
.350	-.135	-.254	-.388	-.524	-.712					
.450	-.222	-.313	-.295	-.420	-.668					
.550	-.260	-.300	-.245	-.327	-.617					
.650	-.188	-.196	-.169	-.202	-.563	-.093	-.059	-.103	-.357	-.493
.750	-.094	-.073	-.080	-.102	-.507	-.045	-.006	.057	-.279	-.474
.850	-.012	.005	.001	-.022	-.443	.006	.046	-.006	-.206	-.439
.900	.023	.043	.041	.020	-.406	.034	.068	.023	-.169	-.420
Right side										
.025	-1.008	-.029	.349	.505	.605	-1.065	-.165	.173	.322	.408
.075	-.448	-.107	.162	.326	.466	-.727	-.298	-.064	.085	.213
.150	-.332	-.120	.073	.213	.346	-.362	-.304	-.290	-.145	-.015
.250	-.345	-.178	-.027	.103	.221					
.350	-.368	-.238	-.122	-.013	.104					
.450	-.266	-.288	-.211	-.107	.003					
.550	-.235	-.264	-.245	-.181	-.080					
.650	-.147	-.204	-.157	-.156	-.117	-.089	-.056	-.088	-.127	-.210
.750	-.066	-.069	-.085	-.098	-.138	-.040	-.013	-.061	-.127	-.244
.850	.035	.052	.018	-.003	-.108	.031	.060	.018	-.061	-.236
.900	.071	.072	.052	.026	-.124	.052	.081	.036	-.049	-.241

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(k) $\alpha = 15.6°$; M = 0.60.

$\frac{x}{c_v}$	\multicolumn{5}{c}{C_p for - }	\multicolumn{5}{c}{C_p for - }
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	\multicolumn{5}{c}{$z/b_v = 0.11$}	\multicolumn{5}{c}{$z/b_v = 0.38$}
Left side
.000	.387	.843	.702	.613	.321	-.292	.756	-.157	-1.566	-2.173
.025	-.512	-.248	.148	-.040	-.449	.362	-.014	-.642	-2.111	-2.173
.075	-.198	-.221	-.068	-.175	-.458	.213	-.066	-.408	-.601	-2.442
.150	-.083	-.216	-.230	-.270	-.465	.132	-.075	-.307	-.504	-.628
.250	-.021	-.180	-.225	-.215	-.343	.055	-.100	-.269	-.442	-.557
.350	-.035	-.162	-.235	-.219	-.284	-.007	-.137	-.269	-.426	-.552
.450	-.085	-.196	-.269	-.242	-.270	-.062	-.175	-.273	-.410	-.529
.550	-.129	-.223	-.278	-.251	-.251	-.079	-.164	-.246	-.394	-.490
.650	-.168	-.239	-.278	-.258	-.233	-.088	-.155	-.221	-.373	-.444
.750	-.205	-.246	-.273	-.261	-.212	-.069	-.116	-.162	-.311	-.373
.850	-.198	-.207	-.219	-.201	-.139	-.033	-.066	-.098	-.221	-.297
.900	-.173	-.175	-.180	-.155	-.097	-.003	-.023	-.055	-.164	-.238
Right side
.025	.256	-.155	-.403	-.024	.156	-.806	-.043	.353	.454	.493
.075	-.076	-.239	-.228	.006	.183	-.441	-.084	.194	.323	.441
.150	-.223	-.209	-.098	.048	.202	-.329	-.093	.107	.225	.344
.250	-.228	-.175	-.030	.071	.144	-.278	-.109	.052	.146	.250
.350	-.237	-.164	-.052	.034	.133	-.269	-.139	-.009	.084	.172
.450	-.271	-.207	-.098	-.017	.123	-.271	-.168	-.057	.027	.098
.550	-.273	-.223	-.125	-.065	.075	-.255	-.173	-.084	-.008	.054
.650	-.269	-.225	-.150	-.097	.029	-.221	-.157	-.080	-.024	.031
.750	-.255	-.223	-.164	-.120	-.001	-.159	-.114	-.055	-.015	.029
.850	-.198	-.182	-.150	-.097	.015	-.092	-.045	-.014	.013	.041
.900	-.166	-.155	-.137	-.081	.011	-.053	-.034	-.004	.020	.031
	\multicolumn{5}{c}{$z/b_v = 0.66$}	\multicolumn{5}{c}{$z/b_v = 0.93$}
Left side
.000	-.324	.485	-.276	-1.120	-1.180	-.079	.415	-.296	-.640	-.729
.025	.330	-.064	-.902	-1.157	-1.127	.137	-.153	-.802	-.621	-.709
.075	.181	-.091	-.635	-1.198	-1.215	-.033	-.175	-.510	-.596	-.723
.150	.077	-.107	-.373	-1.240	-1.477	-.115	-.162	-.255	-.564	-.700
.250	.002	-.123	-.260	-.628	-1.288	-.122	-.112	-.173	-.525	-.661
.350	-.049	-.141	-.237	-.203	-.798					
.450	-.095	-.157	-.228	-.226	-.403					
.550	-.118	-.148	-.203	-.224	-.293					
.650	-.088	-.112	-.148	-.171	-.224	-.097	-.052	-.112	-.318	-.619
.750	-.060	-.071	-.096	-.123	-.169	-.062	-.020	-.073	-.254	-.550
.850	-.019	-.016	-.036	-.054	-.093	-.026	.009	-.030	-.192	-.456
.900	-.003	.005	-.007	-.033	-.067	.002	.032	-.007	-.152	-.394
Right side
.025	-.870	-.071	.344	.431	.459	-.627	-.164	.160	.241	.261
.075	-.670	-.107	.169	.300	.404	-.462	-.191	-.059	.038	.107
.150		-.109	.075	.192	.305	-.333	-.168	-.143	-.084	-.026
.250	-.260	-.127	-.004	.093	.192					
.350	-.251	-.146	-.059	.006	.105					
.450	-.221	-.155	-.091	-.033	.045					
.550	-.193	-.141	-.098	-.054	.006					
.650	-.147	-.109	-.082	-.061	-.017	-.072	-.050	-.089	-.125	-.155
.750	-.095	-.061	-.059	-.040	-.021	-.060	-.027	-.064	-.120	-.162
.850	-.012	.018	.009	.006	.002	-.001	.027	-.009	-.072	-.129
.900	.018	.037	.030	.025	.006	.013	.043	.011	-.054	-.120

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Continued

(*l*) $\alpha = 15.8°$; M = 0.80.

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$	$\beta=-3.9°$	$\beta=0°$	$\beta=3.9°$	$\beta=7.9°$	$\beta=12.7°$
	$z/b_v=0.11$					$z/b_v=0.38$				
	Left side									
.000	.565	.910	.801	.730	.405	.114	.814	.229	-.726	-1.497
.025	-.663	-.228	.207	.083	-.259	.362	.015	-.566	-1.511	-1.752
.075	-.200	-.213	-.015	-.096	-.349	.224	-.050	-.397	-.683	-1.868
.150	-.070	-.205	-.222	-.278	-.489	.151	-.062	-.302	-.541	-.777
.250	-.004	-.155	-.218	-.236	-.384	.079	-.082	-.262	-.459	-.615
.350	-.019	-.128	-.222	-.215	-.301	.004	-.128	-.265	-.429	-.518
.450	-.067	-.173	-.253	-.236	-.276	-.053	-.181	-.274	-.423	-.594
.550	-.114	-.198	-.262	-.235	-.252	-.082	-.186	-.265	-.411	-.557
.650	-.157	-.224	-.270	-.258	-.258	-.106	-.189	-.248	-.403	-.507
.750	-.207	-.271	-.299	-.294	-.266	-.097	-.155	-.191	-.320	-.423
.850	-.215	-.244	-.259	-.245	-.202	-.056	-.096	-.121	-.219	-.332
.900	-.200	-.210	-.221	-.200	-.152	-.021	-.050	-.069	-.151	-.264
	Right side									
.025	.317	-.116	-.452	-.063	.127	-.698	-.044	.348	.463	.534
.075	-.027	-.237	-.225	-.005	.157	-.438	-.088	.204	.334	.452
.150	-.222	-.202	-.075	.050	.186	-.330	-.087	.134	.248	.358
.250	-.230	-.155	-.006	.083	.164	-.276	-.096	.072	.170	.268
.350	-.230	-.137	-.027	.054	.098	-.268	-.137	.013	.103	.187
.450	-.261	-.183	-.066	.004	.080	-.279	-.180	-.049	.030	.104
.550	-.271	-.208	-.107	-.047	.044	-.273	-.195	-.083	-.013	.044
.650	-.268	-.224	-.149	-.099	-.013	-.239	-.189	-.095	-.039	.013
.750	-.291	-.253	-.181	-.146	-.061	-.184	-.149	-.073	-.028	.006
.850	-.238	-.216	-.164	-.128	-.050	-.106	-.079	-.026	.002	.021
.900	-.212	-.189	-.152	-.118	-.051	-.073	-.065	-.015	.008	.010
	$z/b_v=0.66$					$z/b_v=0.93$				
	Left side									
.000	-.109	.521	-.043	-.788	-1.102	.002	.431	-.165	-.559	-.793
.025	.339	-.038	-.845	-1.192	-1.021	.151	-.173	-.862	-.629	-.757
.075	.193	-.073	-.695	-1.204	-1.033	-.037	-.207	-.774	-.596	-.763
.150	.086	-.099	-.406	-1.189	-1.222	-.149	-.181	-.288	-.563	-.763
.250	.008	-.123	-.271	-.814	-1.164	-.132	-.114	-.145	-.527	-.729
.350	-.059	-.154	-.250	-.155	-.888					
.450	-.117	-.175	-.239	-.213	-.621					
.550	-.143	-.169	-.221	-.230	-.472					
.650	-.112	-.131	-.158	-.181	-.341	-.102	-.044	-.103	-.348	-.580
.750	-.074	-.084	-.101	-.128	-.247	-.059	-.011	-.053	-.259	-.523
.850	-.024	-.001	-.029	-.057	-.141	-.013	.035	-.006	-.186	-.447
.900	-.002	.014	-.001	-.027	-.099	.015	.053	.019	-.141	-.400
	Right side									
.025	-.898	-.053	.358	.447	.486	-.813	-.175	.175	.265	.281
.075	-.724	-.097	.187	.313	.412	-.530	-.228	-.058	.057	.130
.150	-.391	-.097	.098	.213	.311	-.365	-.178	-.149	-.082	-.028
.250	-.273	-.122	.011	.109	.200					
.350	-.261	-.161	-.060	.018	.103					
.450	-.238	-.173	-.101	-.042	.023					
.550	-.207	-.161	-.121	-.074	-.025					
.650	-.157	-.122	-.101	-.077	-.044	-.071	-.043	-.066	-.141	-.198
.750	-.096	-.064	-.061	-.053	-.050	-.053	-.011	-.052	-.132	-.196
.850	-.010	.023	.009	.001	-.019	.013	.053	.013	-.065	-.164
.900	.031	.050	.037	.018	-.019	.033	.067	.028	-.047	-.159

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL

TAIL, AND HORIZONTAL TAIL - Concluded

(m) $\alpha = 15.9^\circ$; M = 0.85.

$\dfrac{x}{c_v}$	C_p for –					C_p for –				
	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$	$\beta=-3.9^\circ$	$\beta=0^\circ$	$\beta=3.9^\circ$	$\beta=7.9^\circ$	$\beta=12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
Left side										
.000	.621	.926	.836	.759	.412	.213	.836	.314	-.569	-1.246
.025	-.701	-.215	.226	.114	-.195	.352	.012	-.543	-1.373	-1.606
.075	-.210	-.218	.001	-.076	-.292	.216	-.056	-.411	-.720	-1.641
.150	-.074	-.207	-.226	-.285	-.473	.150	-.066	-.319	-.585	-.820
.250	-.002	-.157	-.226	-.265	-.437	.080	-.081	-.272	-.499	-.686
.350	-.024	-.123	-.226	-.246	-.322	.005	-.127	-.273	-.463	-.444
.450	-.065	-.174	-.262	-.258	-.276	-.062	-.188	-.288	-.447	-.594
.550	-.108	-.192	-.260	-.241	-.252	-.098	-.197	-.279	-.437	-.616
.650	-.161	-.228	-.272	-.264	-.276	-.127	-.211	-.269	-.432	-.549
.750	-.223	-.286	-.321	-.331	-.309	-.120	-.184	-.216	-.362	-.473
.850	-.226	-.262	-.279	-.287	-.249	-.072	-.114	-.140	-.244	-.365
.900	-.212	-.227	-.240	-.238	-.185	-.036	-.066	-.085	-.167	-.280
Right side										
.025	.338	-.103	-.506	-.095	.112	-.675	-.044	.337	.455	.548
.075	-.011	-.248	-.239	-.030	.141	-.450	-.091	.205	.330	.458
.150	-.229	-.204	-.080	.039	.182	-.335	-.089	.136	.243	.366
.250	-.234	-.151	-.009	.073	.169	-.282	-.100	.075	.167	.277
.350	-.232	-.131	-.029	.045	.098	-.275	-.140	.015	.095	.196
.450	-.262	-.178	-.068	-.007	.065	-.289	-.194	-.054	.020	.107
.550	-.272	-.204	-.108	-.061	.028	-.289	-.215	-.098	-.036	.042
.650	-.272	-.227	-.153	-.118	-.033	-.260	-.211	-.111	-.068	.000
.750	-.311	-.271	-.200	-.173	-.093	-.203	-.168	-.097	-.062	-.016
.850	-.259	-.237	-.186	-.160	-.086	-.121	-.097	-.048	-.026	-.004
.900	-.230	-.205	-.173	-.154	-.089	-.087	-.081	-.035	-.017	-.010
	$z/b_v = 0.66$					$z/b_v = 0.93$				
Left side										
.000	-.051	.534	.015	-.683	-1.187	.002	.424	-.138	-.566	-.770
.025	.341	-.040	-.886	-1.232	-1.067	.144	-.184	-.949	-.659	-.745
.075	.194	-.074	-.724	-1.232	-1.049	-.051	-.239	-.838	-.633	-.758
.150	.088	-.103	-.428	-1.223	-1.147	-.181	-.205	-.341	-.601	-.754
.250	.005	-.130	-.289	-.856	-1.085	-.156	-.130	-.150	-.566	-.728
.350	-.072	-.168	-.269	-.189	-.870					
.450	-.133	-.197	-.260	-.246	-.680					
.550	-.167	-.194	-.236	-.275	-.549					
.650	-.137	-.154	-.180	-.219	-.410	-.108	-.063	-.114	-.369	-.578
.750	-.092	-.100	-.114	-.153	-.309	-.064	-.020	-.067	-.278	-.529
.850	-.035	-.005	-.042	-.075	-.191	-.015	.029	-.014	-.200	-.454
.900	-.008	.014	-.008	-.040	-.145	.014	.052	.018	-.150	-.406
Right side										
.025	-.952	-.050	.359	.445	.494	-1.054	-.182	.170	.262	.292
.075	-.781	-.099	.186	.305	.418	-.577	-.254	-.071	.046	.138
.150	-.404	-.099	.098	.206	.323	-.318	-.202	-.174	-.108	-.030
.250	-.283	-.140	.008	.102	.205					
.350	-.276	-.177	-.071	.004	.101					
.450	-.260	-.198	-.125	-.072	.019					
.550	-.236	-.191	-.146	-.111	-.041					
.650	-.177	-.148	-.127	-.111	-.073	-.095	-.053	-.084	-.161	-.221
.750	-.113	-.090	-.078	-.086	-.079	-.059	-.017	-.065	-.159	-.229
.850	-.016	.017	-.001	-.027	-.044	.007	.052	.007	-.091	-.098
.900	.022	.044	.028	-.004	-.044	.032	.068	.024	-.061	-.190
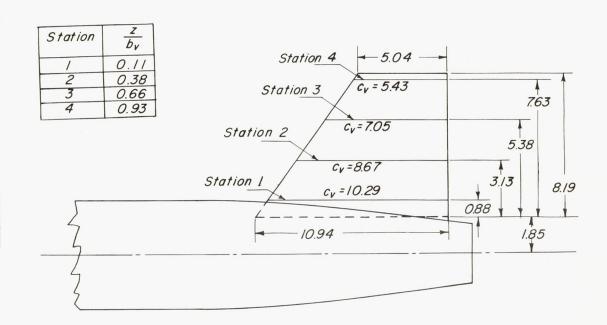
(a) Complete model.
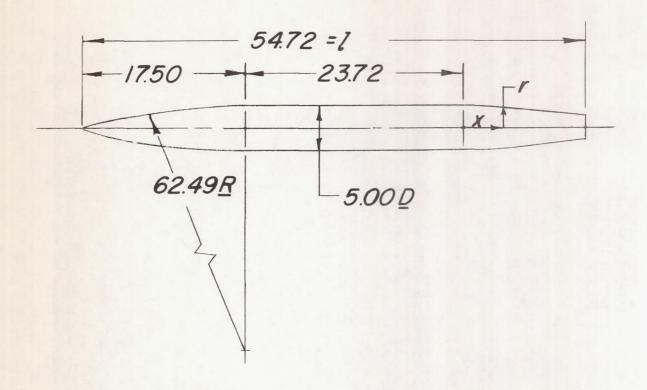
Figure 1.- Details of test model. All linear dimensions are in inches.

Upper and lower surface chordwise locations of orifices for all spanwise stations, x/c_v
.000
.025
.075
.150
.250
.350
.450
.550
.650
.750
.850
.900

Station	$\dfrac{z}{b_v}$
1	0.11
2	0.38
3	0.66
4	0.93

Station 4 — 5.04 —

$c_v = 5.43$

Station 3

7.63

$c_v = 7.05$

Station 2

5.38

$c_v = 8.67$

3.13

8.19

Station 1

$c_v = 10.29$

0.88

10.94

1.85

(b) Details of vertical-tail pressure-orifice locations.

Figure 1.- Continued.

Afterbody Coordinates

x/l	r/l
0	.0456
.0320	.0445
.0639	.0427
.1187	.0390
Straight-line taper	
.2460	.0301

(c) Fuselage dimensions; fineness ratio, 10.94.

Figure 1.- Concluded.

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Figure 2.- Photograph of model mounted on sting-support system in Langley
high-speed 7- by 10-foot tunnel.
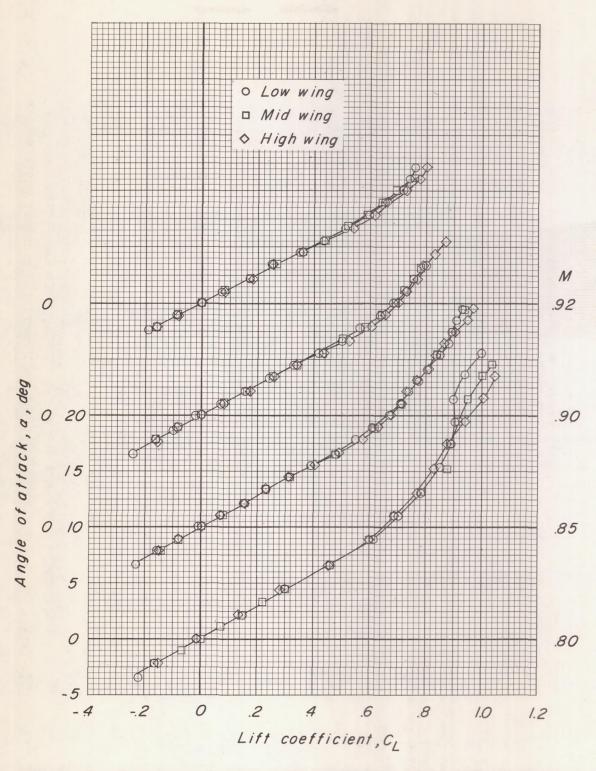
Figure 3.- Variation of model lift coefficient with angle of attack. Horizontal tail off.

(a) $\beta = 0^{\circ}$.
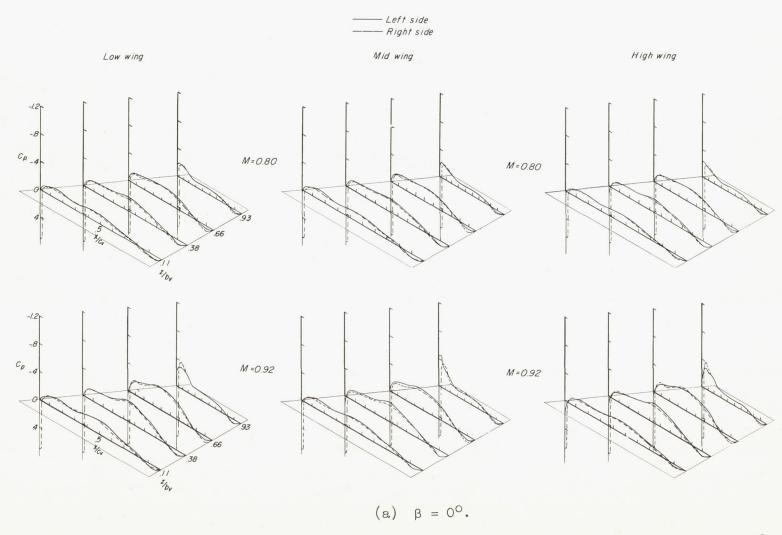
Figure 4.- Effect of wing height on vertical-tail chordwise pressure distribution. $i_t = 0^{\circ}$; $\alpha = 0^{\circ}$.

(b) β = 3.9°.

Figure 4.- Continued.

(c) $\beta = 7.8^\circ$.

Figure 4.- Continued.

Left side
Right side

Low wing

Mid wing

High wing

(d) β = 12.7°.

Figure 4.- Concluded.

—— Left side
———— Right side

Low wing   Mid wing   High wing

(a) β = 0°.
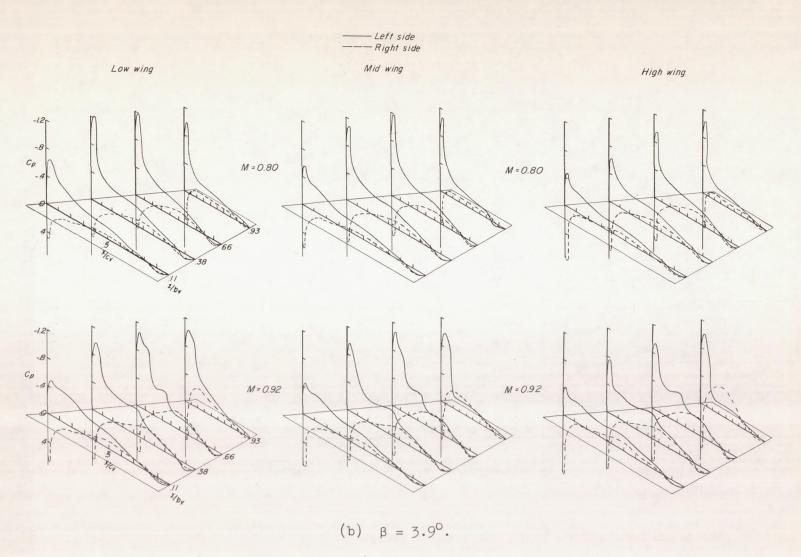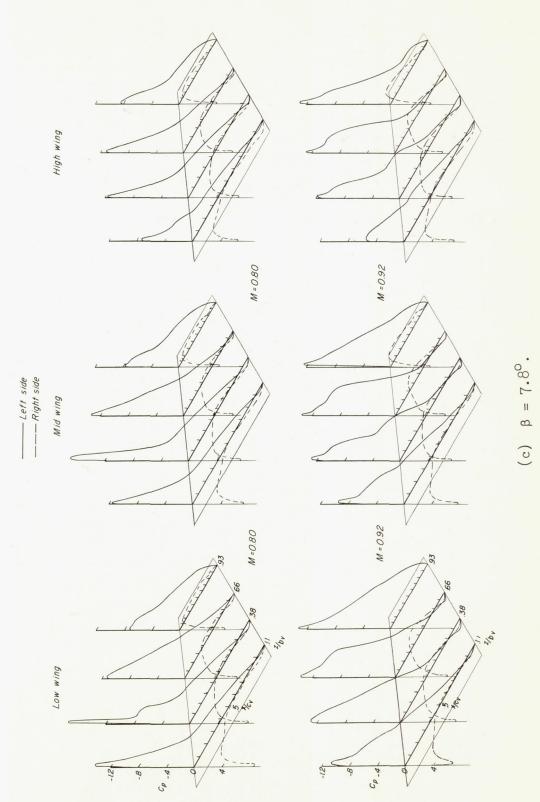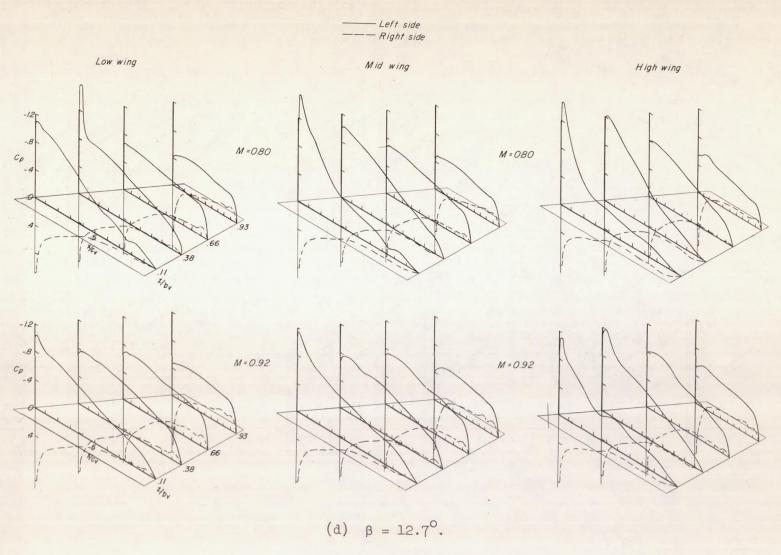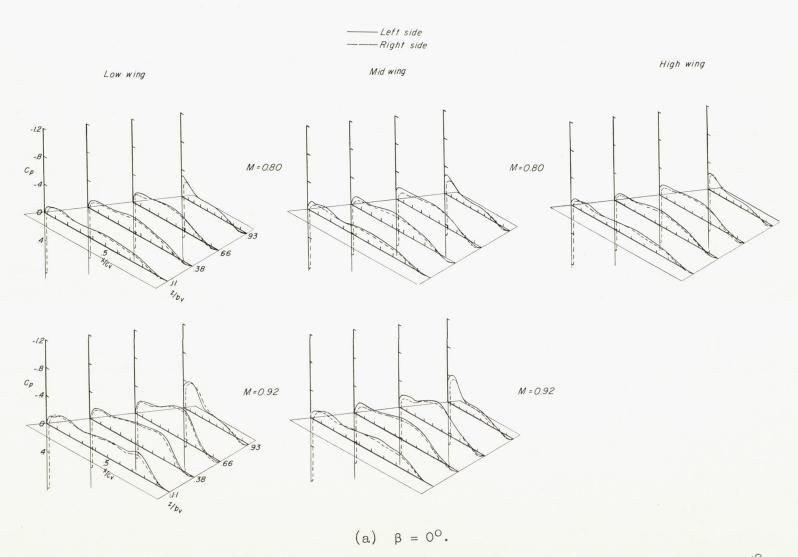
Figure 5.- Effect of wing height on vertical-tail pressure distribution. $i_t = 0°$; $\alpha = 9.6°$.

(b) β = 3.9°.

Figure 5.- Continued.
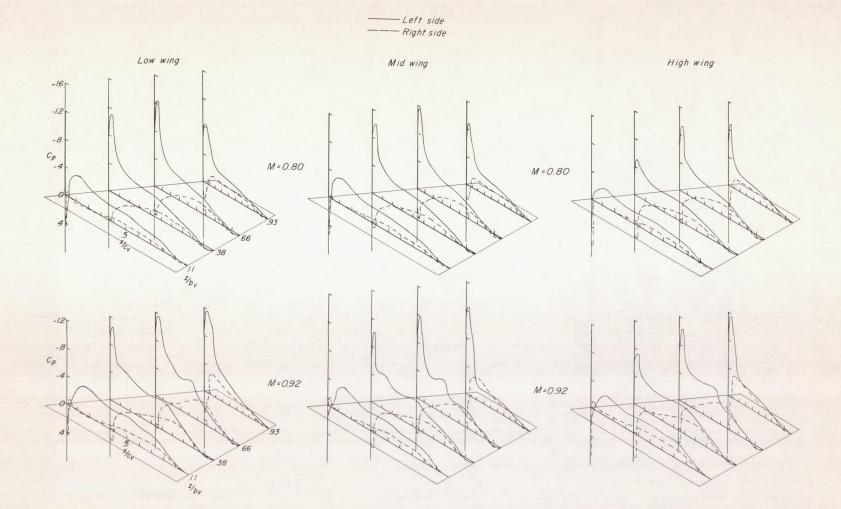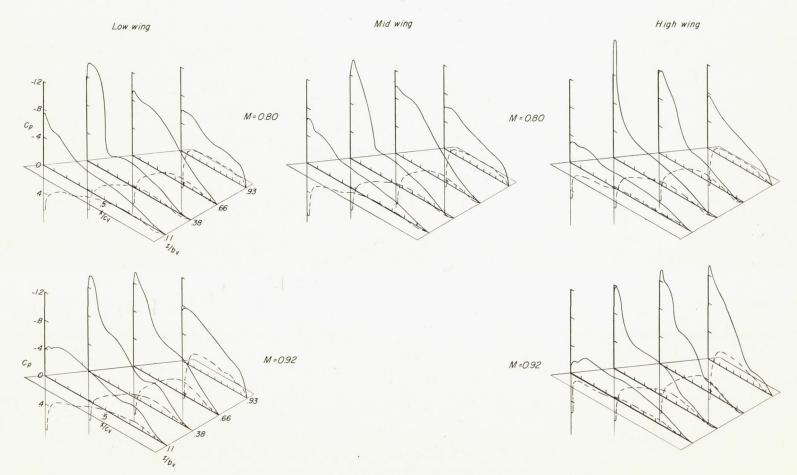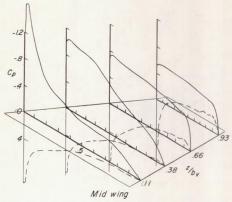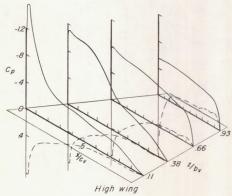
(c) β = 7.8°.

Figure 5.- Continued.
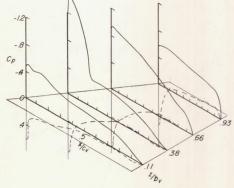
(d) β = 12.7°.

Figure 5.- Concluded.
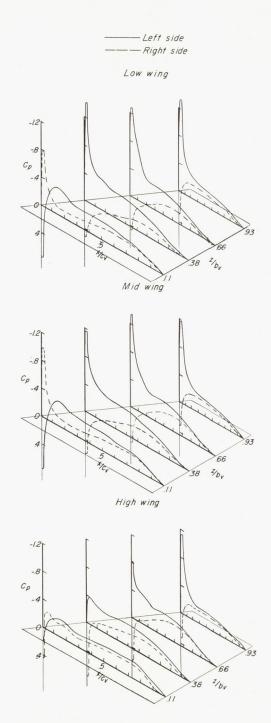
Figure 6.- Effect of wing height on the vertical-tail chordwise pressure
distribution. $i_t = 0^o$; $\alpha = 15.8^o$; $\beta = 3.9^o$; $M = 0.80$.

(a) M = 0.80.
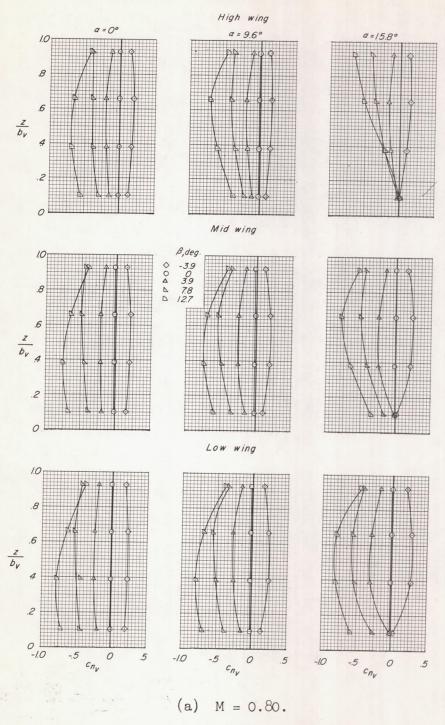
Figure 7.- Effect of angle of sideslip on the spanwise variation of
vertical-tail section normal-force coefficients. $i_t = 0°$.

(b) M = 0.92.
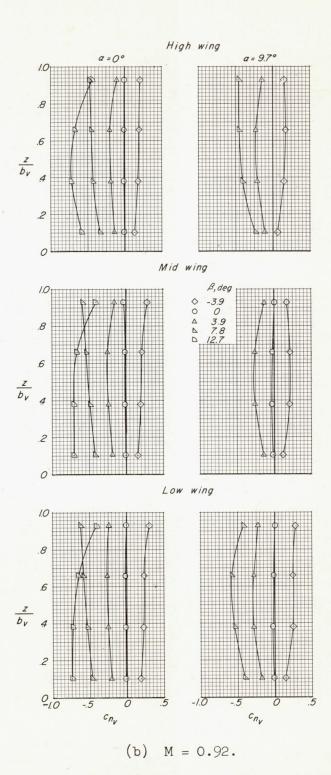
Figure 7.- Concluded.
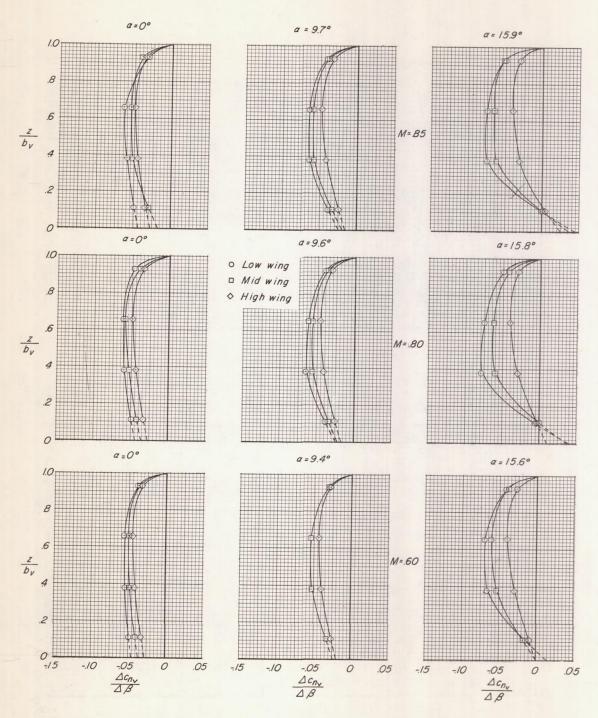
Figure 8.- Effect of wing height on the spanwise variation of vertical-tail section normal-force coefficient per degree of sideslip. $\Delta\beta \approx 8°$; $i_t = 0°$.
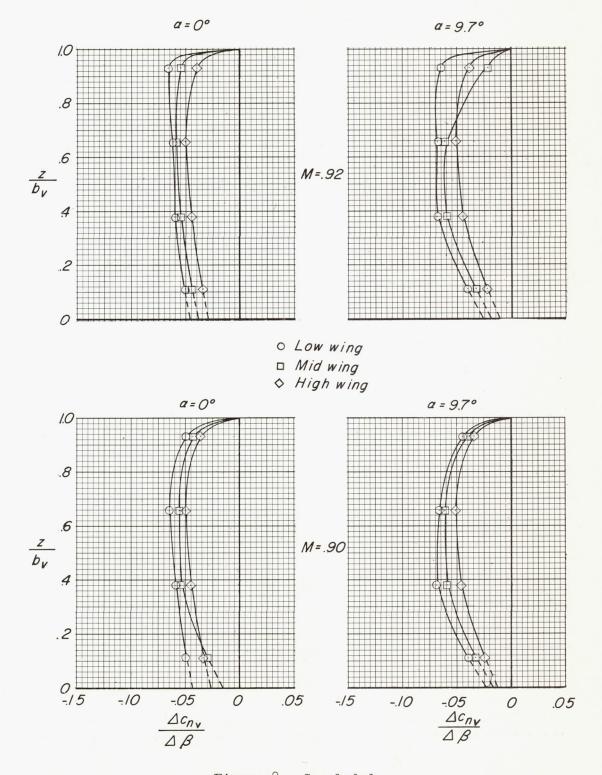
Figure 8.- Concluded.

(a) $\alpha = 0°$.
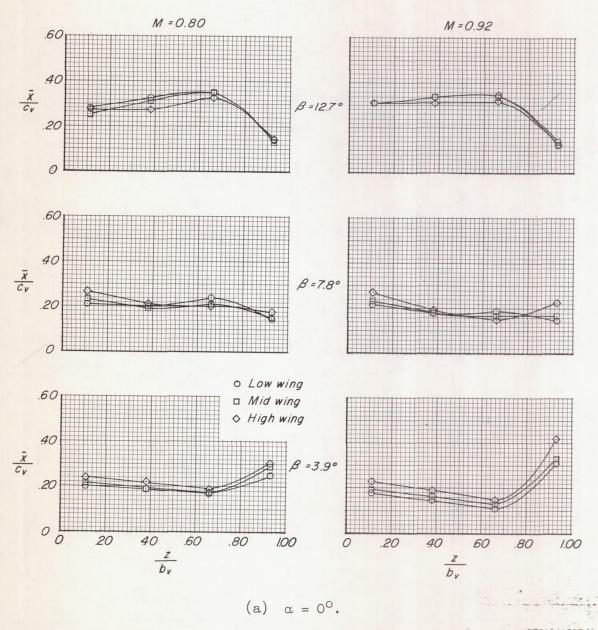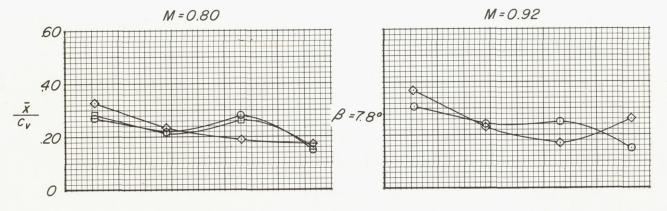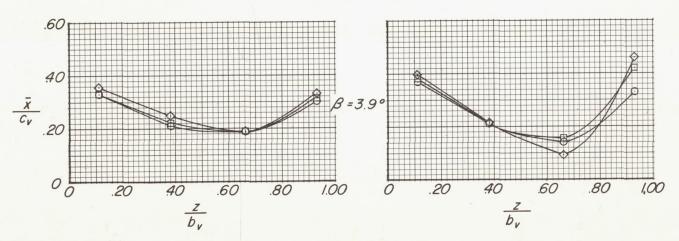
Figure 9.- Effect of wing height on the chordwise location of vertical-
tail local centers of pressure. $i_t = 0°$.

(b) α = 9.6°.

Figure 9.- Continued.

(c) α = 15.8°.
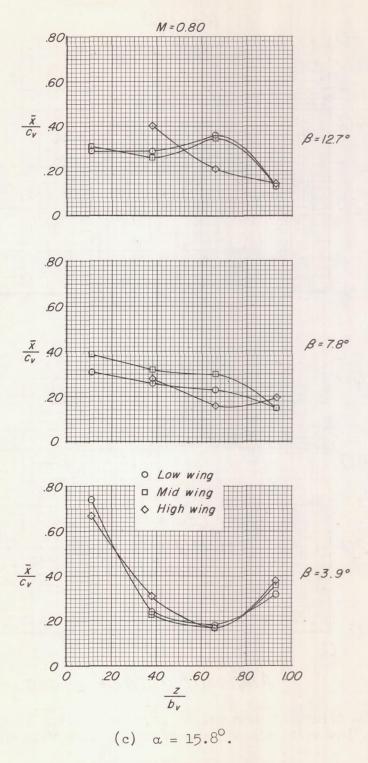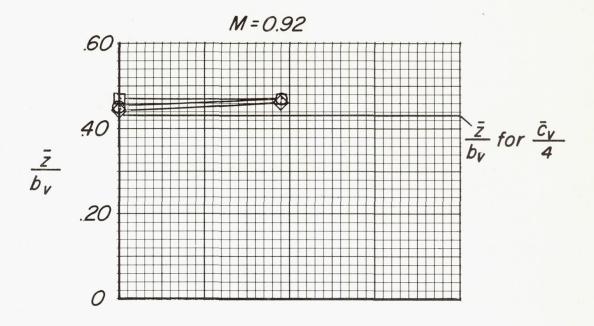
Figure 9.- Concluded.
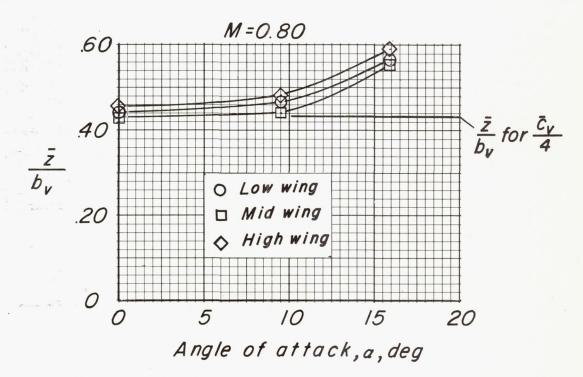
Figure 10.- Effect of wing height on the variation of the spanwise location of the vertical-tail center of pressure with angle of attack. $\Delta\beta \approx 8°$; $i_t = 0°$.
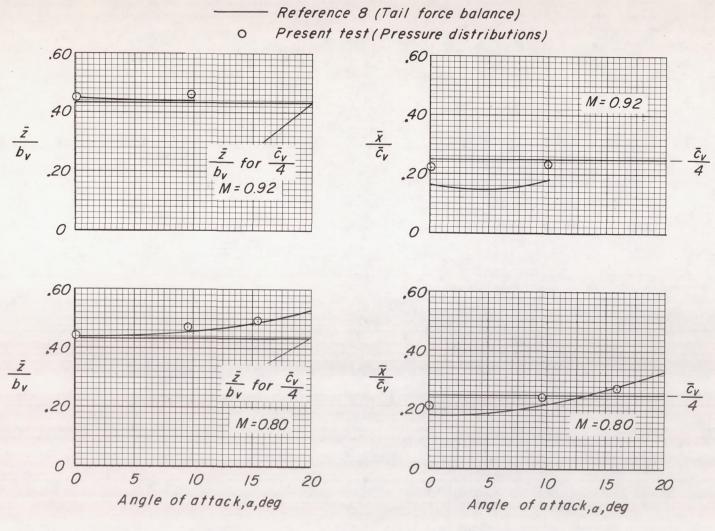
Figure 11.- Comparison of the location of the vertical-tail spanwise centers of pressure as obtained from force measurements and pressure measurements. Wing off; $\Delta\beta \approx 8°$; $i_t = 0°$.